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NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

TECHNICAL NOTE

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DESIGN CHARTS FOR FLAT COMPRESSION PANELS HAVING
LONGITUDINAL EXTRUDED Y-SECTION STIFFENERS
AND COMPARISON WITH PANELS HAVING
FORMED Z-SECTION STIFFENERS

By Norris F. Dow and William A. Hickman

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SUMMARY

Design charts are presented for 24S-T (bare sheet) and 75S-T (Alclad sheet) aluminum-alloy flat compression panels with longitudinal extruded Y-section stiffeners. In addition, comparisons are made among panels designed from these charts and 24S-T aluminum-alloy panels having formed Z-section stiffeners designed from available design charts. The comparisons indicate that, if the ratio of intensity of loading to sheet thickness is relatively high, the charts presented may be used to design a Y-stiffened panel in either 24S-T or 75S-T material which is lighter in weight than a 24S-T Z-stiffened panel designed from the available charts to meet the same conditions. The amount of weight saving depends upon the specific design conditions and is greatest for the 75S-T Y-stiffened panels. The comparisons also indicate that the 24S-T Y-stiffener will have a height somewhat greater than the comparable 24S-T Z-stiffener or 75S-T Y-stiffener; the height of the 24S-T Z-stiffener generally is the smallest. In addition, the comparisons indicate that the average spacing of rivet lines is generally somewhat less for the 24S-T Y-stiffened panels than for the 24S-T Z-stiffened panels or for the 75S-T Y-stiffened panels; the average spacing generally is greatest for the 24S-T Z-stiffened panels. If the ratio of intensity of loading to sheet thickness is relatively low, however, the comparative designs indicate that a 24S-T Z-stiffened panel designed from the available charts will be slightly lighter in weight than a Y-stiffened panel of either 24S-T or 75S-T material. If the present design charts are extended to lower values of the ratio of stiffener thickness to sheet thickness to cover the region of heavy sheet thickness more thoroughly (where the ratio of intensity of loading to sheet thickness is relatively low), a Y-stiffened-panel design in this region will probably compare more favorably with a Z-stiffened-panel design than the charts presented indicate. If no sheet thickness is specified so that the design may have optimum proportions, it is concluded that both the 24S-T and 75S-T Y-stiffened panels will be lighter than the 24S-T Z-stiffened panels throughout the range of design conditions investigated.

INTRODUCTION

The problem of the design of wing compression panels of minimum weight is one that has confronted aircraft structural engineers since the advent of stressed-skin construction. Although the final solution of this problem has not yet been achieved, progress has been made toward its solution as the cumulative result of numerous theoretical and experimental studies to determine "optimum proportions" and "efficient" stiffener shapes.

Recently two such studies (references 1 and 2) have established a type of plot which appears particularly useful in connection with the design of wing compression panels of minimum weight. Reference 1 presented a theoretical comparison of the efficiencies of various stiffener shapes by plotting the average stress at failure - an inverse measure of the weight - against a parameter containing the main design conditions, the load per chordwise inch of panel, and the effective length of panel. Reference 2 used the same type of plot to provide design charts for Z-stiffened panels based on extensive test data from which the optimum proportions can be determined for a particular design.

Study of references 1 and 2 reveals that if panels with longitudinal stiffeners are to have high structural efficiency a stiffener shape is required which has both high-column strength and local-buckling strength. Because a stiffener in the shape of a Y appeared more nearly to meet this requirement than the Z-section or hat-section stiffeners of references 2 and 3, an investigation was made in the Langley structures research laboratory of the National Advisory Committee for Aeronautics to determine the compressive strength of panels having Y-section stiffeners. Both 24S-T (bare sheet) and 75S-T (Alclad sheet) aluminum-alloy panels were tested in this investigation. The results of these tests are presented herein in the form of design charts similar to the design charts for panels with Z-section stiffeners of reference 2.

In order to show the relative structural efficiencies of Y-section and Z-section stiffeners, comparisons are also presented of panels of both types designed to have the minimum weight required to meet a large range of loading conditions.

SYMBOLS

The symbols used to represent the various dimensions of the panels are shown in figure 1. In addition, the following symbols are used:

$\bar{\sigma}_f$	average stress at failing load, ksi
σ_{cr}	stress for local buckling of the sheet, ksi
σ_{cy}	compressive yield stress, ksi
P_i	compressive load per inch of panel width, kips per inch
c	coefficient of end fixity as used in Euler column formula
A_i	cross-sectional area per inch of panel width, or equivalent thickness of panel, inches
$\bar{\epsilon}_f$	shortening per unit length at failing load
p	radius of gyration, inches
I_i	moment of inertia per inch of panel width, cubic inches

TEST SPECIMENS AND PROCEDURE

The test specimens were constructed with six stiffeners and five bays as shown in figure 1. Three sizes of stiffeners were used that corresponded to values of b_w/t_w of 20, 25, and 30 with the nominal value of t_w held constant at 0.064 inch (see fig. 2) and various values of t_w/t_s were obtained by varying the sheet thickness. The stiffeners were riveted to the sheets with Al7S-T flat-head rivets (AN442AD) on all panels.

Values of the with-grain compressive yield stress for the material used for the sheets (bare sheets were used for the 24S-T panels and Alclad sheets for the 75S-T panels) and for the extrusions are given in table 1. The values of compressive yield stress for the extrusions represent the average values for specimens cut from the three webs and the outstanding flange of the Y-section extrusions at the locations shown in figure 3. Values of the compressive yield stress for the material used to construct the Z-stiffened panels of reference 2 are also given in table 1 for comparison.

The test procedure was essentially the same as that used in other panel tests in the Langley structures research laboratory. (See references 2 and 3.) The panels were tested flat-ended without side support (because their transverse stiffness was small) in a hydraulic testing machine having an accuracy of one-half of 1 percent of the load (See fig. 4.) The stress for local buckling of the sheet was determined by the "strain-reversal method." A discussion of this and other methods of experimentally determining the stress for local buckling is given in reference 4. For panels having a greater width of sheet under the Y's than between the Y's, strain gages were mounted inside the stiffeners, as indicated in figure 5. The ends of the panels were ground flat and parallel, and the method of alinement in the testing machine was such as to insure uniform bearing on the ends of the specimens. An end fixity coefficient of 3.75 has been indicated for such panel tests in this machine, and this value was therefore used in reducing the test data.

Proportions of the specimens and test data - including values of the ratios of rivet diameter to sheet thickness d/t_S and pitch to sheet thickness p/t_S , average stress at failing load $\bar{\sigma}_f$, and unit shortening at failing load $\bar{\epsilon}_f$ - are given in tables 2 to 4. The unit shortening was measured as the average of the strains indicated by four, $\frac{1}{2}$ -inch gage length, resistance-type wire strain gages mounted on the quarter points of the second and fifth stiffeners, as may be seen in figure 4.

Figure 6 shows a 24S-T aluminum-alloy Y-stiffened panel and its 75S-T counterpart after failure. There tended to be a greater shattering of the 75S-T panels than of the 24S-T panels.

DESIGN CHARTS

Design charts for panels with extruded Y-section stiffeners are presented in figures 7 to 11 for 24S-T and in figures 12 to 16 for 75S-T aluminum alloy. These charts were prepared from the test data of tables 2 to 4 in a manner similar to that described in the appendix of reference 2 for Z-stiffened panels. The use of design charts of this type is described fully in reference 2, and a procedure similar to that given in reference 2 for designing a panel for maximum structural efficiency (minimum weight) by use of the charts is included in appendix A of the present paper. This design procedure makes it possible to achieve the balance for given values of P_f , L/\sqrt{C} , and t_S between the proportions which

will produce the highest average stress at failure and the proportions which will make the area such that the failing stress is just reached at the design load.

A comparison of the curves of the design charts and the test data from which the curves were derived indicates that the stresses given by the curves of the design charts for both the Y-stiffened panels of the present paper and the Z-stiffened panels of reference 2 are on the whole very slightly less than the stresses given by the test data. For both stiffener types, however, there are regions on the design charts in which the curves are interpolated or extrapolated far from the test data, and in these regions the accuracy of the charts is probably less than that indicated by the comparisons of curves and data. The region of the Y-stiffened-panel charts for which there

is the least test data is that for $\frac{t_w}{t_s} = 0.40$ at wide stiffener

spacings. (See figs. 7 and 12.) Slightly greater caution should be exercised in the use of the charts in this region than elsewhere in the design charts. The region of the Z-stiffened-panel charts for

which there is the least test data is that for $\frac{t_w}{t_s} = 1.00$ at close

stiffener spacings. In this region, additional unpublished test data have indicated that the curves may be as much as 5 or 6 percent too high.

GENERALIZED COMPARISON OF Y-STIFFENED PANELS

AND Z-STIFFENED PANELS

Without restrictions on the sheet thickness. - If there are no restrictions on the sheet thickness that may be used, Y-stiffened and Z-stiffened panels may be compared by envelope curves faired over the curves of their design charts. Such a comparison of envelope curves is shown in figure 17. Because the average stresses at failing load for the envelope curves for the Y-stiffened panels of both 24S-T and 75S-T material are above those for the Z-stiffened panels, the Y-stiffened panels of optimum proportions are evidently lighter in weight than the Z-stiffened panels of optimum proportions throughout the range covered by the present design charts.

With restrictions on sheet thickness. - The sheet thickness needed to achieve the stresses of the envelope curves of figure 17 are fixed for any given intensity of loading by the proportions required by the

envelope curves. In the design of wing compression panels, however, the sheet thickness is often fixed by other considerations such as torsional stiffness of the wing. Accordingly, curves which show the effect of a variation in sheet thickness should provide a more useful evaluation of the relative structural efficiencies of Y-stiffened and Z-stiffened panels than do the envelopes of figure 17; therefore, figures 18 to 20 were prepared. In these figures, the average stresses at failure $\bar{\sigma}_f$ carried by Y-stiffened and Z-stiffened-panel designs, selected for minimum weight according to the procedure given in appendix A, are plotted against the parameter $\frac{P_1}{t_S}$ for a series of values of $\frac{P_1}{L/\sqrt{C}}$. A discussion of this type of plot is given in appendix B.

The chief importance of figures 18 to 20 is that the figures indicate directly the average stress at failure $\bar{\sigma}_f$ carried by the minimum-weight designs of Y-stiffened or Z-stiffened panels which can be achieved within the large range of proportions covered by the design charts for given values of P_1 , L/\sqrt{C} , and t_S . The effect of a change in any one of the variables $\bar{\sigma}_f$, P_1 , L/\sqrt{C} , and t_S on any of the others, therefore, may be studied from these figures. For example, consider the effect of a change in t_S on the value of $\bar{\sigma}_f$. The relative flatness of the curves at the higher values of P_1/t_S indicates that the sheet thickness can be varied over a rather large range with very little change in the value of $\bar{\sigma}_f$ which can be achieved.

A comparison of figures 18 to 20 brings out the following facts:

(1) Minimum-weight designs of both 24S-T and 75S-T Y-stiffened panels are lighter in weight (carry higher stresses) than minimum-weight designs of 24S-T Z-stiffened panels in the region of high values of P_1/t_S (thin sheet); but the Z-stiffened designs are of slightly lighter weight in the region of low values of P_1/t_S (thick sheet). No sharply defined boundary exists between these two regions. Instead, there is a range of values of P_1/t_S , which varies with $\frac{P_1}{L/\sqrt{C}}$, for which the curves of figures 18 to 20 coincide.

(2) The actual amount by which the Y-stiffened-panel design is lighter than the Z-stiffened panel (or vice versa) varies somewhat

erratically as the design conditions, $\frac{P_i}{L/\sqrt{c}}$ and $\frac{P_i}{t_S}$ are varied because of the cusped nature of the curves.

(3) The value of $\frac{t_W}{t_S}$, which produces the minimum-weight design for given values of $\frac{P_i}{L/\sqrt{c}}$ and $\frac{P_i}{t_S}$, is smallest for the 75S-T Y-stiffened panels and largest for the 24S-T Z-stiffened panels.

COMPARISON OF MINIMUM-WEIGHT DESIGNS OF Y-STIFFENED PANELS AND Z-STIFFENED PANELS

Although figures 18 to 20 show in a general way the relative structural efficiencies of Y- and Z-stiffened panels, probably the best way to evaluate two types of panel construction is to compare panels of each type designed to meet the same conditions. A comparison of this nature permits consideration of each of the many factors which influence the choice of the most desirable construction for a given situation, such as the number of rivet lines, the space required for the stiffeners, and the distance from the outside surface of the sheet to the axis of the center of gravity of the panel. A series of comparative designs of Y- and Z-stiffened panels, therefore, was made in a manner similar to that used in making the designs from which figures 18 to 20 were prepared. Four values of P_i , namely, 2.0, 3.0, 5.0, and 8.0 kips per inch and also four values of L , namely, 10, 20, 30, and 40 inches were used for the comparative designs. The end fixity coefficient c was assumed equal to 1 in all cases.

In making the comparative designs, obtainment of extruded Y-stiffeners in the thicknesses required by the designs was assumed possible. A minimum thickness in which these shapes can be successfully extruded exists, however, and this minimum thickness is probably above the thickness required for many of the designs for which P_i is equal to or less than 3.0 kips per inch. The reasons for retaining these designs are (1) they may be scaled up for higher intensities of loading for which the minimum thickness that can be extruded is no longer a limitation, and (2) to emphasize the fact that if the intensity of loading is low, the Y-stiffener will not be satisfactory simply because it cannot be obtained.

Numerical values of the properties of the comparative designs for all values of t_W/t_S covered by the design charts are given in tables 5 to 8. The values for the particular ratio of t_W/t_S for

which minimum weight is achieved are enclosed in parentheses. In order to show graphically the general variation of the proportions of these designs as the panel length and the sheet thickness are varied, figures 21 and 22 have been prepared. These figures present cross-sectional views, drawn to scale, of some of the minimum-weight designs of Y-stiffened and Z-stiffened panels for $P_i = 5.0$ kips per inch (table 7).

The comparative designs were made according to the procedure given in appendix A except that all values of t_w/t_s given by the design charts were investigated for each design. Because the design charts cover only a limited range of proportions, the comparisons between the designs are in some cases affected by the limited range of proportions covered by the charts. With this qualification, comparisons of the minimum-weight designs of tables 5 to 8 and figures 21 and 22 show that:

(1) At relatively high values of P_i/t_s , which are associated with thin sheets, the average stresses at failure $\bar{\sigma}_f$ for both the 24S-T and the 75S-T Y-stiffened panels are greater than those for the 24S-T Z-stiffened panels, and these stresses indicate that less weight is required for the Y-stiffened than for the Z-stiffened panels, the least weight being required for the 75S-T Y-stiffened panels. On the other hand, at relatively low values of P_i/t_s , which are associated with thick sheets, the average stresses at failure $\bar{\sigma}_f$ for both the 24S-T and 75S-T Y-stiffened panels are slightly less than those for the 24S-T Z-stiffened panels, and these stresses indicate that the Z-stiffened panel is slightly lighter in weight. The magnitude of the difference in weight between the two types of panel varies with the values of P_i/t_s and $\frac{P_i}{L/\sqrt{c}}$.

(2) The height of the stiffeners H is generally somewhat greater and, hence, consumes more space inside the wing for the 24S-T Y-stiffened panels than for the 24S-T Z-stiffened panels or for the 75S-T Y-stiffened panels; the height of the 24S-T Z-stiffened panel generally is the smallest.

(3) The average spacing of rivet lines S is generally somewhat less and, hence, requires more rivets for the 24S-T Y-stiffened panels than for the 24S-T Z-stiffened panels or for the 75S-T Y-stiffened panels; the average spacing generally is greatest for the 24S-T Z-stiffened panels.

(4) Only if the values of both P_i/t_S and $\frac{P_i}{L/\sqrt{c}}$ are relatively high does the value of the stress for local buckling of the sheet σ_{cr} tend to be higher for the 24S-T or 75S-T Y-stiffened panels than for the 24S-T Z-stiffened panels.

(5) The distance from the outside surface of the sheet to the axis of the center of gravity of the panel \bar{h} , which tends to reduce the effectiveness of the panel to resist bending of the wing, is generally greater for the 24S-T Y-stiffened panels than for the 24S-T Z-stiffened panels or for the 75S-T Y-stiffened panels; the distance \bar{h} generally tends to be smallest for the 24S-T Z-stiffened panels at low values of P_i/t_S (thick sheet) and smallest for the 75S-T Y-stiffened panels at high values of P_i/t_S (thin sheet). (The magnitude of the reduction in effectiveness of the panel to resist bending of the wing depends on the thickness of the wing. The thinner the wing, the greater the reduction.)

(6) The value of the radius of gyration ρ is generally greater (and also the value of $\rho^2 A_i = I_i$ is generally greater) for the 24S-T Y-stiffened panel than for the 24S-T Z-stiffened panel or for the 75S-T Y-stiffened panel; generally, ρ tends to be smallest for the 24S-T Z-stiffened panels at low values of P_i/t_S (thick sheet) and smallest for the 75S-T Y-stiffened panels at high values of P_i/t_S (thin sheet). (The greater the value of $\rho^2 A_i$, the greater the effectiveness of the panel to resist local air loads.)

EFFECT OF SMALL DIFFERENCES IN TEST SPECIMENS ON THE COMPARISONS OF 24S-T Y-STIFFENED AND Z-STIFFENED PANELS

Only small differences occurred between the test specimens for the 24S-T Y-stiffened and Z-stiffened panels. Differences occurred in material properties, diameter and pitch of rivets, and range of proportions of the elements of the panels actually tested and hence the proportions covered by the resulting design charts.

The effect of these differences on the comparisons of 24S-T Y-stiffened and Z-stiffened panels are discussed in the following sections.

Effect of material properties. - If the material properties of the Y-stiffened panels and the Z-stiffened panels had been identical, would the comparisons have been more or less favorable to the Y-stiffened panels? Table 1 indicates that the average compressive

yield stress of the material used for the Z-stiffened panels and that of the material used for the sheets of the 24S-T Y-stiffened panels were identical but that the average compressive yield stress of the extruded Y-stiffeners as measured was between 3 and 4 percent less than that of the Z-stiffeners before forming. Because forming tends to raise the compressive yield stress (see reference 5), the average properties of the formed Z-stiffeners were probably more than 3 percent above those for the extruded Y-stiffeners. Accordingly, it may be inferred that if the Y-stiffeners and the Z-stiffeners had had identical properties - as might have been obtained if extruded Z-stiffeners had been used, for example - the 24S-T Y-stiffened panels tested would have increased in strength relative to the Z-stiffened panels, and the comparisons would have been more favorable to the Y-stiffened panels.

Effect of riveting. - If the riveting of the Y-stiffened panels and the Z-stiffened panels had been identical, would the comparisons have been more or less favorable to the Y-stiffened panels? A comparison of rivet proportions listed in tables 2 to 4 with those of reference 2 indicates that the Y-stiffened panels were more strongly riveted than the Z-stiffened panels. Reference 6 shows that the strength of short panels having close stiffener spacing increased with an increase in the diameter of the rivets and also increased with a decrease in the pitch of the rivets. Subsequent tests have indicated that as the length of the panel is increased the size and pitch of rivets have progressively less effect on the strength of the panel until the panel strength may actually decrease with an increase in the strength of riveting. If the Y-stiffened panels and Z-stiffened panels had had identical riveting, therefore, the comparative designs would probably have come out less favorable to the Y-stiffened panel

in the case of the short panels ($\frac{P_i}{L/Vc}$) and possibly very slightly more favorable to the Y-stiffened panel in the case of the long panels ($\frac{P_i}{L/Vc}$).

Effect of panel proportions. - If proportions of Y-stiffened panels or Z-stiffened panels different from those tested and, hence, those covered by the resulting design charts had been considered, would the comparisons have been more or less favorable to the Y-stiffened panels? It can be seen by inspection of tables 5 to 8 that:

- (1) The lightest weight 24S-T Y-stiffened-panel design for a given set of design conditions often requires a stiffener which is 2 sheet gages thinner than that for the comparable Z-stiffened-

panel design. (This agrees with the fact that the value of t_w/t_s for minimum weight is smaller for the Y-stiffened panel than for the Z-stiffened panel. See figs. 18 to 20.) Also, the present charts do not cover a large enough range of proportions to permit a Y-stiffener more than 1 gage thinner than a Z-stiffener in the region of heavy sheet thickness. If the design charts were extended to cover lower values of the ratio t_w/t_s so that a Y-stiffened-panel design could always be made which had a stiffener 2 gages thinner than the stiffener for the best Z-stiffened-panel design, then the Y-stiffened-panel design would probably be less inferior to the Z-stiffened-panel design in the region of heavy sheet thickness. Similarly, if the charts were extended in the other direction so that in all cases a Z-stiffened-panel design with a Z-stiffener 2 sheet gages thicker than the comparable Y-stiffened-panel design could be made, possibly the Y-stiffened panel would be less superior to the Z-stiffened panel in the region of very light sheet thickness.

(2) The lightest weight Y-stiffened-panel designs - in far more cases than for the Z-stiffened-panel designs - are obtained at the maximum or minimum values of b_w/t_w given by the design charts. Extending the range of proportions covered to higher and lower values of b_w/t_w would be likely, therefore, to result in lighter weight designs of Y-stiffened panels in more cases than in lighter designs of Z-stiffened panels.

Because a very extensive test program was run to establish optimum proportions for the Z-stiffener ($\frac{b_F}{b_w} = 0.3 \text{ to } 0.5$), and no such program has been run to establish optimum proportions for the Y-stiffener, the proportions of the Y-stiffener possibly could be improved and, hence, the comparative designs made more favorable to the Y-stiffened panel for all sheet thicknesses. Among the changes in proportions of the Y-stiffened panels which might result in overall improvements in their structural efficiencies are: (1) a change in the angle included between the legs of the Y-stiffeners in order to effect a better balance between the width of sheet under the Y-stiffeners and between adjacent Y-stiffeners, (2) a change in relative proportions of the outstanding "T" part of the Y-stiffeners, and (3) a reduction in the width of attachment flanges of the Y-stiffeners, particularly for $\frac{t_w}{t_s} = 1.00$.

GENERAL TRENDS INDICATED BY MINIMUM-WEIGHT DESIGNS

In addition to the comparisons of Y-stiffened and Z-stiffened panels afforded by the designs of tables 5 to 8 and figures 21 and 22, there are several general trends indicated by the designs and by figures 18 to 20 which apply to both types of construction. These general trends are in some cases affected by the limited range of proportions covered by the present design charts. These trends as well as the comparisons between the two types of construction, are also strictly for minimum-weight designs. With the foregoing qualifications, the comparative designs show that:

For given values of P_i and L/\sqrt{c}

- (1) The weight of panel generally increases ($\bar{\sigma}_f$ decreases) with an increase in sheet thickness, but the lightest panel is often obtained not at the thinnest sheet gage at which a design can be achieved but with the sheet 1 or 2 gages thicker than the minimum.
- (2) The stress for local buckling of the sheet σ_{cr} and also the ratio $\sigma_{cr}/\bar{\sigma}_f$ generally decreases with an increase in sheet thickness, but the maximum value of the stress for local buckling of the sheet is often obtained not at the thinnest sheet gage at which a design can be achieved but with the sheet 1 or 2 gages thicker than the minimum.
- (3) The average spacing of rivet lines S increases (requiring fewer rivets) with an increase in sheet thickness.
- (4) The distance from the outside surface of the sheet to the axis of the center of gravity of the panel \bar{h} , which tends to decrease the effectiveness of the panel to resist bending of the wing, generally decreases with an increase in sheet thickness.

And for given values of P_i and t_S

- (1) The weight of panel increases ($\bar{\sigma}_f$ decreases) with an increase in the value of L/\sqrt{c} .
- (2) The stress for local buckling of the sheet σ_{cr} , but not necessarily the ratio $\sigma_{cr}/\bar{\sigma}_f$, generally decreases with an increase in the value of L/\sqrt{c} , except at the heavy sheet thicknesses.
- (3) The height of the stiffeners H increases with an increase in the value of L/\sqrt{c} .

(4) The average spacing of rivet lines S generally increases (again requiring fewer rivets) with an increase in the value of L/\sqrt{c} , except at the heavy sheet thicknesses.

(5) The distance from the outside surface of the sheet to the axis of the center of gravity of the panel \bar{h} , which tends to decrease the effectiveness of the panel to resist bending of the wing, generally increases with an increase in the value of L/\sqrt{c} .

(6) The radius of gyration ρ increases (not necessarily increasing the effectiveness of the panel to resist local air loads) with an increase in the value of L/\sqrt{c} .

CONCLUDING REMARKS

In this paper, charts have been presented from which 24S-T (bare sheet) and 75S-T (Alclad sheet) aluminum-alloy flat compression panels having longitudinal extruded Y-section stiffeners may be designed to have the minimum weight required to carry a given intensity of loading at a given effective length of panel with a given sheet thickness. Comparisons have been made of panels designed from these charts and similar designs of Z-stiffened panels, in order to bring out the differences in characteristics of 24S-T and 75S-T and of Y- and Z-stiffened-panel designs. In the case of actual wing compression panels, however, there are often additional factors to be considered which have been neglected for the comparisons, such as the effects of local air loads, the distance from the neutral axis of the wing to the center of gravity of the cross section of the panel, the sheet curvature, the edge support, and the shear combined with the compression, or the effects on the design procedure of specifying stiffener height or spacing in addition to sheet thickness. The labor involved in the introduction of so many additional variables into the comparisons, however, is obviously prohibitive. In fact some of the variables cannot be introduced because the necessary research has not been done. Because in any particular design some such additional factor may be important, the choice of a type of construction in most cases is best made by evaluating the characteristics of panels of several types designed to meet all the requirements of the actual application. The design charts of the present paper (figs. 7 to 17) together with the tables of section properties (tables 9 to 13) may be used as an aid in such an evaluation of the characteristics of a 24S-T or 75S-T Y-stiffened panel.

APPENDIX A

METHOD OF DESIGNING A Y-STIFFENED PANEL FOR MINIMUM WEIGHT

The following procedure, which is similar to that given in reference 2 for Z-stiffened panels, permits the selection of the minimum-weight Y-stiffened panel for given values of the design conditions P_i , L/\sqrt{c} , and t_s . In this procedure, the conditions P_i , L/\sqrt{c} , and t_s are first combined to determine the values of the parameters $\frac{P_i}{L/\sqrt{c}}$ and $\frac{P_i}{t_s}$. Next, from figures 18 or 20 the value of t_w/t_s is found for which the minimum-weight design will be achieved. Then a study is made of all the curves of the design chart for that ratio of t_w/t_s at the given value of $\frac{P_i}{L/\sqrt{c}}$. From this study, a plot is made of the variation of the stress at failure with stiffener spacing for panels having all the proportions covered by the chart. Because the chart gives $\bar{\sigma}_f$ in terms of relative proportions (dimension ratios), the absolute size is established for each set of panel proportions by computing the sheet thickness required to make the design load, P_i , divided by the area A_i equal to the failing stress $\bar{\sigma}_f$. The variation of these sheet thicknesses, calculated as $\frac{P_i}{\bar{\sigma}_f A_i}$ is then plotted against stiffener spacing. This second plot makes the establishment of stiffener spacings associated with the design value of the sheet thickness for each of the panel proportions possible. Reference to the first plot permits the determination of the stresses corresponding to these proportions and the selection of the proportions (usually by interpolation) which give the highest stress (minimum weight) at the given sheet thickness.

As an example of this procedure, the values and quantities for the 24S-T design shown in figure 23 for $P_i = 5.0$ kips per inch, $L = 20$ inches, $c = 1$, and $t_s = 0.102$ inch are given in table 14 and are employed in the following steps:

- (1) Compute $\frac{P_i}{L/\sqrt{c}}$ and $\frac{P_i}{t_s}$.

(2) From figures 18 or 20 (in the example, fig. 18 for 24S-T is used) determine the value or values of t_w/t_s which should be investigated to find the minimum-weight design at the values of $\frac{P_i}{L/\sqrt{c}}$ and $\frac{P_i}{t_s}$ determined in step (1) (in the example, $\frac{t_w}{t_s} = 0.40$).

(3) From the curves for the particular value of t_w/t_s determined in step (2) (in the example, fig. 7), pick off for each value of b_w/t_w and b_s/t_s the value of $\bar{\sigma}_f$ corresponding to the value of $\frac{P_i}{L/\sqrt{c}}$ given by step (1).

(4) Pick from tables 9 to 13 (in the example, table 9) the values of A_1/t_s corresponding to the ratios used in step (3).

(5) Compute the sheet thickness that would be required to make the design load P_i divided by the area A_1 equal to the failing stress $\bar{\sigma}_f$ in each case, thus $t_s = \frac{P_i}{\bar{\sigma}_f A_1}$.

(6) Plot the values of $\frac{P_i}{A_1}$ and $\bar{\sigma}_f$ against b_s/t_s for each value of b_w/t_w and mark the values of $\bar{\sigma}_f$ at the value of b_s/t_s for which $\frac{P_i}{\bar{\sigma}_f t_s}$ equals the design value of t_s (in the example, 0.102 in.). The plots of this step for the example under consideration are given as the two lower plots in figure 23. For ease in interpolating to find the value of b_w/t_w for the design, a curve of b_w/t_w against b_s/t_s is also conveniently established by plotting the consecutive values of b_w/t_w (18, 21, 24, and so forth) at the values of b_s/t_s for which $\frac{P_i}{\bar{\sigma}_f t_s}$ equals the design value of t_s (the upper plot in fig. 23).

(7) After step (6) has been completed for all the values of b_w/t_w , draw curves of stress and of b_w/t_w against b_s/t_s through the points determined in step (5) (heavy curves in fig. 23).

(8) Each of the curves drawn in step (7) represents a series of designs, all of which have the required value of t_S (in the example, 0.102 in.). The maximum point on the curve of $\bar{\sigma}_f$ against b_S/t_S indicates the design for minimum weight. Note this maximum value of $\bar{\sigma}_f$, the value of b_S/t_S at which it is reached, and the corresponding value of b_W/t_W which can be picked from the curve of b_W/t_W against b_S/t_S .

(9) Check computations by picking from tables 9 to 13 the value of A_1/t_S corresponding to the ratios selected for minimum weight in step (8). If computations and plots are correct,

$$P_i = \frac{A_1}{\bar{\sigma}_f t_S} t_S$$

(10) Compute the following panel dimensions from the proportions determined by this design procedure with the aid of tables 9 to 13:

$$t_W = \frac{t_W}{t_S} t_S$$

$$t_L = \frac{t_L}{t_W} t_W$$

$$b_S = \frac{b_S}{t_S} t_S$$

$$b_L = \frac{b_L}{b_W} b_W$$

$$b_W = \frac{b_W}{t_W} t_W$$

$$t_F = \frac{t_F}{t_W} t_W$$

$$b_A = \frac{b_A}{t_W} t_W$$

$$b_F = \frac{b_F}{b_W} b_W$$

$$b_Y = \frac{b_Y}{b_W} b_W$$

$$r = \frac{r}{t_W} t_W$$

$$H = \left(1.79 \frac{b_W}{t_W} + 1.6 \right) t_W$$

$$S = 0.5 \frac{b_S}{t_S} t_S + \left(0.52 \frac{b_W}{t_W} + 2.3 \right) t_W$$

$$\bar{h} = \frac{\bar{h}}{t_S} t_S$$

$$p = \frac{p}{t_S} t_S$$

(11) Compute the diameter and pitch of rivets from the proportions listed in tables 9 to 13, as

$$d = \frac{d}{t_S} t_S$$

$$p = \frac{p}{t_S} t_S$$

(12) Find σ_{cr} by interpolation between the short horizontal lines in figures 7 to 16.

If the values of $\frac{P_1}{L/\sqrt{c}}$ and $\frac{P_1}{t_S}$ computed in step (1) are such that the point on figure 18 or 20 corresponding to these values is near a boundary between two values of t_w/t_S , it is advisable to follow the design procedure of steps (1) to (12) for both values of t_w/t_S .

APPENDIX B

DESIGN CHART FOR DETERMINING THE STRUCTURAL EFFICIENCY

If a chart is to be drawn which will provide a direct measure of the structural efficiency of a wing compression panel, that chart must contain in its parameters all the design conditions which apply to the panel. In references 1 and 2 the parameter $\frac{P_i}{L/Vc}$, which contains the design conditions of compressive load and effective length of panel, was used for charts that measure directly the structural efficiency when those are the design conditions.

The trend toward higher speeds and thinner wings and the accompanying requirement of high torsional stiffness, however, tends to establish a minimum acceptable sheet thickness for the panel. It therefore appears desirable to include the sheet thickness t_S within the parameters used for preparing charts indicative of the structural efficiency of panels.

A suitable parameter incorporating the sheet thickness appears to be P_i/t_S . This parameter, which represents the load divided by the area of sheet alone, denotes the upper limit of stress that can be carried by a panel for a given sheet thickness because any stiffeners added to the sheet must increase the panel area and reduce the stress below that determined as P_i/t_S . This upper limit is shown in figures 18 to 20 as the dashed line. Besides indicating the upper limit of stress, this line also represents the stress that would be carried by a panel having a value of $\frac{t_W}{t_S} = 0$ (pure shell construction), but only if such a panel could actually carry the indicated stress without failing.

As the value of $\frac{t_W}{t_S}$ for the panel is increased from zero, the stress carried will decrease from that equal to the value of P_i/t_S . The actual magnitude of the highest stress that can be achieved for each value of $\frac{t_W}{t_S}$ given by the design charts can be determined by assuming values of $\frac{P_i}{L/Vc}$ and $\frac{P_i}{t_S}$ and by examining all the

individual curves of the design charts in a manner similar to the minimum-weight-design procedure at the assumed values of t_w/t_s , $\frac{P_i}{L/\sqrt{c}}$, and $\frac{P_i}{t_s}$. (Values of $\bar{\sigma}_f \frac{A_i}{t_s}$, which are equal to $\frac{P_i}{t_s}$, instead of

$\frac{P_i}{\bar{\sigma}_f \frac{A_i}{t_s}}$ are computed in step (5) and plotted in step (6) of the

procedure. See appendix A. Also designs are made for a series of values of P_i/t_s , corresponding to a series of design values of t_s , from each plot of step (6).)

The foregoing procedure was used to establish the curves given in figures 18 to 20, which indicate the stresses attainable by minimum-weight designs as P_i/t_s is varied for chosen values of $\frac{P_i}{L/\sqrt{c}}$. The stress for any point on one of these curves is therefore a direct measure of the structural efficiency of the best design that can be made to meet the given design conditions P_i , L/\sqrt{c} , and t_s .

Because the design charts are drawn for definite values of t_w/t_s , the curves of figures 18 to 20 contain cusps which correspond to the intersection of the curves resulting from the use of the design charts for consecutive values of t_w/t_s . Light lines have been drawn in figures 18 to 20 connecting these cusps, thus dividing the figures into regions in which the indicated values of t_w/t_s produce the minimum-weight designs. As previously noted, the region for $\frac{t_w}{t_s} = 0$ is the dashed line, for which $\bar{\sigma}_f = \frac{P_i}{t_s}$.

For given values of P_i , L/\sqrt{c} , and t_s , the value of t_w/t_s that will produce the lightest weight Y-stiffened or Z-stiffened panel may be determined directly from figures 18 to 20. Since very slight variations in $\bar{\sigma}_f$ near the cusps of the curves could cause an appreciable shift in the location of the cusps in many cases, the light lines should be considered as only approximate boundaries. If the point corresponding to a particular design being considered lies near a boundary between two values of t_w/t_s , it might be wise to investigate both values of t_w/t_s in making that design.

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TABLE 1

VALUES OF THE COMPRESSIVE YIELD STRESS FOR THE MATERIALS
USED FOR CONSTRUCTING THE Z-STIFFENED PANELS
AND THE Y-STIFFENED PANELS

	σ_{cy} (ksi)	Sheet (bare)	Stiffeners (bare sheet before forming)
24S-T Z-stiffened panels (from reference 2)	Maximum	46.5	46.5
	Average	44.0	44.0
	Minimum	41.0	41.0
24S-T Y-stiffened panels	σ_{cy} (ksi)	Sheet (bare)	Stiffeners (extrusions)
	Maximum	47.3	48.0
	Average	44.0	42.3
75S-T Y-stiffened panels	σ_{cy} (ksi)	Sheet (Alclad)	Stiffeners (extrusions)
	Maximum	69.7	86.5
	Average	67.3	78.2
	Minimum	64.7	67.6

NATIONAL ADVISORY
COMMITTEE FOR AERONAUTICS

TABLE 2

TEST DATA AND PROPORTIONS OF SPECIMENS HAVING $\frac{t_w}{t_s} = 0.40$

[Nominal proportions are given in parentheses]

(a) 24S-T SHEET AND STIFFENERS

Proportions of test specimens														Test data			
t_w (in.)	$\frac{t_w}{t_s}$	$\frac{b_s}{t_s}$	$\frac{b_w}{t_w}$	$\frac{b_A}{t_w}$	$\frac{b_w}{b_Y}$	$\frac{t_w}{t_L}$	$\frac{b_w}{b_L}$	$\frac{t_w}{t_F}$	$\frac{b_w}{b_F}$	$\frac{r}{t_w}$	$\frac{d}{t_s}$	$\frac{p}{t_s}$	$\frac{L}{b_w}$ (a)	σ_{or} (ksi)	$\bar{\sigma}_f$ (ksi)	$\frac{F_1}{L/\sqrt{c}}$ (kip/in.)	$\bar{\epsilon}_f$
(0.064) 0.064 .066 .067 .065	(0.140) 0.399 .403 .406 .397	(25) 25.0 25.3 24.3 24.6	(20) 20.0 19.7 19.1 19.7	(9.3) 9.30 9.05 8.85 9.14	(0.96) 0.96 0.94 0.94 0.93	(0.94) 0.921 0.908 0.946 0.931	(1.07) 1.07 1.05 1.06 1.05	(0.47) 0.487 0.486 0.503 0.488	(1.14) 1.14 1.12 1.11 1.10	(1) 1 1 1 1	(1.54) 1.54 1.53 1.53 1.52	(46) 46 46 46 46	12.5 24.2 24.2 43.8 74.6	40.7	42.7 42.3 39.0 39.0 20.8	1.222 .592 .592 .292 .101	672×10^{-5} 592 592 362 218
.066 .068 .065 .065	.396 .416 .390 .407	23.9 24.6 24.1 25.1	24.1 25.8 24.8 24.5	8.96 8.79 8.79 9.11	.936 .947 .930 .906	.936 .947 .930 .906	.488 .497 .461 .486						12.9 25.8 45.0 77.3	40.7 39.2 45.9 24.3	43.0 41.9 43.9 0.095	1.028 .496 .359 .095	584 575 328 232
.066 .066 .066 .064	.409 .407 .405 .394	25.0 24.8 24.5 24.5	29.1 29.1 29.1 30.0	9.01 9.00 9.03 9.30	.953 .971 .971 .945	.953 .971 .971 .945	.471 .472 .469 .457						13.2 26.3 46.1 79.1	41.8	43.3 41.1 34.7 23.8	.855 .107 .199 .078	541 507 567 225
.067 .066 .066 .066	.416 .411 .416 .409	35.1 35.0 35.2 34.4	19.1 19.3 19.3 19.2	8.85 8.96 8.99 8.93	.936 .994 .947 .937	.936 .994 .947 .937	.501 .491 .496 .501						11.7 23.5 41.2 70.7	33.3 34.0 35.4 35.4	36.9 35.6 34.7 20.7	1.055 .504 .279 .099	727 391 346 196
.066 .067 .066 .067	.401 .406 .403 .410	33.9 34.2 34.4 35.0	24.2 24.0 24.0 24.0	8.99 8.89 9.05 8.92	.923 .915 .904 .939	.923 .915 .904 .939	.490 .492 .484 .490						12.3 24.6 42.9 73.6	32.7 33.8 32.9 32.9	36.0 35.7 34.8 21.4	.888 .410 .230 .082	552 521 331 201
.063 .062 .063 .064	.378 .372 .378 .393	34.1 34.1 33.7 34.6	30.6 30.2 30.5 30.0	9.49 9.57 9.48 9.30	.953 .947 .949 .973	.953 .947 .949 .973	.455 .454 .460 .468						12.6 25.2 44.1 75.7	30.3 34.4 33.9 33.9	36.2 36.3 34.5 21.1	.696 .354 .192 .068	516 394 325 204
.067 .067 .066 .067	.410 .409 .400 .407	49.3 49.2 48.9 49.7	19.1 19.2 19.3 19.6	8.87 8.91 8.97 9.04	.932 1.010 .946 .940	.932 1.010 .946 .940	.497 .493 .497 .490						10.8 21.8 37.3 65.5	19.1 19.9 20.1 21.2	32.0 29.2 29.0 22.6	.933 .124 .212 .109	1148 591 211 211
.067 .067 .066 .067	.408 .410 .403 .421	49.2 49.2 49.7 50.8	24.0 23.9 24.4 23.9	8.91 8.87 9.05 8.88	.912 .938 .925 .970	.912 .938 .925 .970	.491 .490 .487 .496						11.4 23.0 40.1 68.7	19.5 19.8 20.2 20.0	31.4 31.2 29.2 20.5	.726 .359 .190 .077	713 627 325 192
.062 .061 .064 .065	.393 .379 .405 .409	51.0 50.5 50.7 50.5	30.7 31.6 29.9 29.4	9.52 9.79 9.25 9.11	.934 .905 .976 .985	.934 .905 .976 .985	.457 .440 .469 .475						11.9 23.8 41.6 71.3	16.7 19.1 20.5 19.8	30.6 30.4 29.5 22.5	.567 .286 .158 .070	565 426 320 214

^a Lengths are for the actual test specimens for which $c = 3.75$ approximately.

NATIONAL ADVISORY
COMMITTEE FOR AERONAUTICS

TABLE 2.— Concluded

TEST DATA AND PROPORTIONS OF SPECIMENS HAVING $\frac{t_w}{t_s} = 0.40$ — Concluded

(b) ALCLAD 75S-T SHEET AND 75S-T STIFFENERS

Proportions of test specimens															Test data			
t_w	$\frac{t_w}{t_s}$	$\frac{b_s}{t_s}$	$\frac{b_w}{t_w}$	$\frac{b_A}{t_w}$	$\frac{b_w}{b_A}$	$\frac{t_w}{t_L}$	$\frac{b_w}{b_L}$	$\frac{t_w}{t_F}$	$\frac{b_w}{b_F}$	r	$\frac{d}{t_s}$	$\frac{p}{t_s}$	$\frac{L}{b_w}$ (a)	σ_{cr} (ksi)	$\bar{\sigma}_f$ (ksi)	$\frac{P_1}{E/\sqrt{c}}$ (kip/in.)	\bar{e}_f	
(.064)	(.10)	(25)	(20)	(9.3)	.99	(0.96)	(1.07)	(0.47)	(1.14)	(1)	(1.54)	(4.6)						
.060	.380	25.6	21.4	9.96		.984		.442					12.4	49.8	57.6	1.630	640×10^5	
.067	.422	25.3	19.2	8.92		.967		.493					24.9	51.5	58.1	.834	602	
.064	.411	25.8	20.0	9.27		.955		.481					43.7	-----	55.0	.443	526	
.060	.370	24.7	21.3	9.90		.932		.431					75.0	-----	25.9	.125	261	
			(25)															
.064	0.402	25.2	25.1	9.31		.988		.464					12.8	51.9	57.9	1.351	589	
.062	.395	25.2	25.3	9.40		.966		.470					25.6	55.0	58.2	.675	565	
.066	.418	25.4	24.1	8.96		.938		.486					45.0	-----	52.9	.352	535	
			(35)	(20)														
.062	0.404	37.1	20.6	9.58		.887		.440					11.7	30.7	52.3	1.468	535	
.062	.389	35.5	20.7	9.63		.870		.439					23.5	30.5	50.9	.720	697	
.064	.402	35.3	20.0	9.30		.955		.478					41.2	33.9	45.8	.369	582	
.064	.402	35.4	20.1	9.34		.951		.479					70.8	-----	24.7	.115	243	
			(25)															
.064	.398	35.2	25.1	9.31		.957		.461					12.2	28.3	48.4	1.102	615	
.066	.411	35.2	24.3	9.03		.929		.488					24.5	33.4	47.7	.543	640	
.064	.404	35.2	25.0	9.28		.896		.464					42.8	35.7	49.1	.319	620	
.066	.416	35.8	24.4	9.04		.937		.468					73.6	-----	27.1	.102	265	
			(30)															
.066	.422	35.9	28.9	8.95		.921		.477					12.6	30.8	47.4	.906	567	
.063	.394	35.4	30.5	9.46		.937		.453					25.2	33.0	47.1	.449	520	
.067	.429	36.0	28.8	8.91		.929		.471					44.2	34.5	44.5			
.066	.422	35.7	28.9	8.96		.931		.477					75.8					
			(50)	(20)														
.065	.418	52.2	19.7	9.17		.940		.489					10.1	17.7	43.9	1.320	690	
.064	.410	51.6	19.9	9.25		.925		.485					21.8	16.8	42.3	.594	740	
.066	.421	51.6	19.4	9.03		.939		.491					38.1	17.1	40.2	.324	569	
.061	.393	51.9	21.0	9.74		.860		.441					65.4	17.5	23.8	.111	258	
			(25)															
.065	.411	51.1	24.7	9.17		.967		.481					11.4	15.8	42.0	.944	539	
.066	.417	51.4	24.4	9.07		.919		.482					22.8	16.6	43.6	.491	590	
.065	.416	51.2	24.5	9.08		.949		.469					40.0	17.2	41.6	.266	583	
.065	.415	51.3	24.5	9.08		.941		.485					68.7	17.9	22.9	.085	264	

^aLengths are for the actual test specimens for which c = 3.75 approximately.

TABLE 3
TEST DATA AND PROPORTIONS OF SPECIMENS HAVING $\frac{t_w}{t_s} = 0.63$

[Nominal proportions are given in parentheses.]

(a) 24S-T SHEET AND SUPPORTERS

Lengths are for the actual test specimens for which $c = 3.75$ approximately.

NATIONAL ADVISORY
COMMITTEE FOR AERONAUTICS

TABLE 3.- Concluded

TEST DATA AND PROPORTIONS OF SPECIMENS HAVING $\frac{t_w}{t_s} = 0.63$ - Concluded

(b) ALCLAD 75S-T SHEET AND 75S-T STIFFPENERS

Proportions of test specimens												Test data					
t_w	$\frac{t_w}{t_s}$	$\frac{b_s}{t_s}$	$\frac{b_w}{t_w}$	$\frac{b_A}{t_w}$	$\frac{b_w}{t_y}$	$\frac{t_w}{t_L}$	$\frac{b_w}{b_L}$	$\frac{t_w}{t_F}$	$\frac{b_w}{b_F}$	$\frac{r}{t_w}$	$\frac{d}{t_s}$	$\frac{p}{t_s}$	$\frac{L}{b_w}$ (a.)	σ_{cr} (ksi)	$\bar{\sigma}_f$ (ksi)	$\frac{P_f}{L/\sqrt{E}}$ (kips/in. in.)	ϵ_f
(0.064)	(0.63)	(25)	(20)	(9.3)	(0.86)	(0.94)	(1.07)	(0.17)	(1.14)	(1)	(1.8)	(61)	14.0	58.2	63.2	1.458	619×10^{-5}
0.066	0.633	25.9	19.2	8.93	0.962	0.974	1.07	0.194	0.512	18.9	60.5	63.2	27.9	413	726	620	
0.067	0.643	23.8	19.1	8.85	0.937	0.948	1.06	0.190	0.490	18.9	57.9	57.9	19.1	376	562	562	
0.068	0.641	21.4	19.4	9.03	0.881	0.916	1.06	0.166	0.446	18.9	57.9	57.9	19.0	097	256	256	
0.062	0.605	21.2	20.5	9.52													
			(25)														
0.064	0.621	24.5	25.1	9.32		0.981		0.475					14.2	57.9	1.104	563	
0.066	0.657	25.2	24.1	8.96		0.935		0.493					28.4	56.3	58.9	553	573
0.064	0.615	24.3	25.2	9.35		0.971		0.461					49.8	54.6	54.6	294	524
			(25)														
0.062	0.596	33.8	20.5	9.53		0.866		0.456					13.6	35.9	54.7	1.197	669
0.064	0.616	33.9	20.0	9.26		0.917		0.481					27.3	43.7	57.5	607	616
0.068	0.656	34.2	18.9	8.79		0.960		0.486					47.6	37.5	52.2	315	555
0.064	0.611	33.7	19.9	9.22		0.928		0.488					81.8	—	23.6	084	255
			(25)														
0.064	0.623	34.6	21.8	9.21		0.976		0.479					13.2	38.1	55.1	953	590
0.065	0.645	35.0	21.7	9.15		0.997		0.471					27.8	36.3	53.5	462	602
0.065	0.640	34.9	24.5	9.10		0.934		0.487					48.8	38.9	49.5	244	519
			(25)														
0.065	0.662	35.7	(50)	9.08		0.905		0.472					14.1	32.3	48.6	736	573
0.066	0.654	34.6	29.3	8.97		0.904		0.473					28.2	35.1	49.9	378	548
0.064	0.632	34.6	29.8	9.24		0.903		0.465					49.3	35.2	47.1	204	489
			(25)														
0.065	0.616	48.0	(50)	19.7	9.14		0.938		0.454				13.0	20.4	51.0	1.019	648
0.063	0.593	47.5	20.3	9.43		0.867		0.446					26.2	20.9	50.9	509	689
0.063	0.603	48.8	20.5	9.50		0.868		0.444					45.9	21.9	46.3	260	587
0.061	0.583	48.2	21.0	9.74		0.836		0.437					78.7	22.7	23.7	078	237
			(25)														
0.065	0.623	48.8	21.7	9.18		0.939		0.481					13.5	19.9	50.0	802	652
0.067	0.632	47.7	21.0	8.91		0.933		0.478					27.0	20.8	51.2	423	617
0.066	0.628	48.2	21.2	8.98		0.938		0.476					47.3	21.6	46.6	219	572
0.064	0.610	48.1	25.0	9.28		0.938		0.464					81.0	21.9	23.7	065	234
			(25)														
0.066	0.638	48.6	(30)	21.0	8.99		0.938		0.476				13.8	19.5	45.6	611	566
0.067	0.666	50.4	28.8	8.91		0.937		0.490					27.5	19.5	46.6	325	561
0.066	0.651	49.9	28.9	8.96		0.925		0.479					48.2	19.7	44.3	176	531
0.063	0.621	49.9	30.3	9.40		0.898		0.463					82.6	21.9	23.6	055	256
			(25)														
0.061	0.579	72.4	(20)	21.0	9.77		0.909		0.435				12.2	10.0	43.5	824	739
0.066	0.629	72.9	19.4	9.01		0.956		0.494					24.6	8.7	43.6	413	688
0.065	0.619	72.7	19.7	9.17		0.938		0.483					43.1	9.9	40.2	217	576
0.063	0.602	72.6	20.3	9.41		0.868		0.450					74.0	9.9	22.2	070	279
			(25)														
0.066	0.635	74.0	(25)	21.4	9.04		0.922		0.487				12.8	7.7	43.8	665	634
0.065	0.650	76.2	21.6	9.12		0.904		0.472					25.6	7.9	42.2	313	592
0.065	0.641	75.7	21.8	9.20		0.946		0.478					44.8	9.5	40.5	172	585
0.066	0.642	73.7	21.1	8.95		0.924		0.485					76.8	8.4	22.6	058	276
			(25)														
0.067	0.663	75.4	(20)	28.7	8.89		0.929		0.480				13.2	7.4	39.7	509	560
0.064	0.637	75.8	30.0	9.30		0.921		0.459					26.4	8.9	39.6	252	532
0.066	0.656	75.6	29.1	9.01		0.906		0.471					46.0	8.7	39.5	164	559
0.066	0.659	75.7	29.0	8.97		0.927		0.476					79.0	10.6	22.1	047	280

^aLengths are for the actual test specimens for which $c = 3.75$ approximately.

TABLE 4
TEST DATA AND PROPORTIONS OF SPECIMENS HAVING $\frac{t_w}{t_s} = 1.00$

[Nominal proportions are given in parentheses]

(a) 24S-T SHEET AND STIFFENERS

Proportions of test specimens													Test data				
t_w (in.)	$\frac{t_w}{t_s}$	$\frac{b_s}{t_s}$	$\frac{b_w}{t_w}$	$\frac{b_A}{t_w}$	$\frac{b_w}{b_Y}$	$\frac{t_w}{t_L}$	$\frac{b_w}{b_L}$	$\frac{t_w}{t_F}$	$\frac{b_w}{b_F}$	$\frac{r}{t_w}$	$\frac{d}{t_g}$	$\frac{P}{t_g}$	$\frac{L}{b_w}$ (a)	σ_{cr} (ksi)	$\bar{\sigma}_f$ (ksi)	$\frac{P_1}{L/\sqrt{G}}$ (psi/in.) in.	$\bar{\epsilon}_f$
(0.064)	(1.00)	(25)	(20)	(9.3)	0.96	(0.94)	(1.07)	(0.47)	(1.44)	(1)	(2.44)	(7.4)					
.067	1.035	23.4	19.0	8.53		0.939		.500					13.9	42.4	.966	567×10^{-5}	
.064	.981	27.4	20.2	9.37		.916		.482					29.5	41.2	.433	497	
.068	1.080	24.8	18.9	9.49		.944		.504					50.9	33.9	.206	312	
.069	1.099	25.1	18.6	8.63		1.006		.508					57.6	21.2	.073	209	
		(25)															
.068	1.045	22.9	23.7	8.79		.942		.496					14.7	35.6	.713	590	
.067	1.028	23.0	23.9	8.67		.978		.494					29.3	38.8	.352	490	
.066	1.059	22.9	24.1	8.95		.935		.490					51.3	35.4	.183	336	
.067	1.065	23.4	24.0	8.91		.930		.499					57.9	23.4	.070	227	
		(35)	(20)														
.067	1.039	32.3	19.1	8.89		.949		.499					13.5	38.5	41.1	.843	
.068	1.027	32.3	18.9	8.77		.942		.506					28.9	39.9	.384	455	
.063	.946	34.9	20.2	9.37		.902		.474					50.7	35.1	.189	355	
.067	1.075	32.9	19.0	8.84		1.006		.497					57.1	21.1	.065	245	
		(25)															
.067	1.026	33.1	24.0	8.89		.941		.492					14.6	37.9	39.1	.639	
.067	1.047	32.6	23.7	8.68		.953		.494					29.2	38.5	.312	447	
.067	1.059	33.5	24.0	8.59		.940		.488					51.1	34.7	.158	319	
.066	1.064	33.7	24.1	8.95		.925		.496					57.5	24.3	.064	235	
		(30)															
.064	0.971	35.2	29.9	9.25		.958		.467					14.7	33.3	37.6	.536	
.066	1.003	35.1	29.2	9.05		.988		.477					29.3	32.8	.257	385	
.063	.972	35.6	30.3	9.37		.935		.461					51.3	33.2	.339	357	
.064	1.021	35.6	30.0	9.28		.920		.459					55.0	22.7	.053	218	
		(50)	(20)														
.067	1.052	46.8	19.1	8.85		.940		.501					14.4	25.7	38.7	.646	
.063	.960	49.8	20.3	9.44		.897		.473					28.6	28.1	.371	521	
.065	1.030	46.7	19.8	9.18		.934		.450					50.1	27.6	.151	359	
.067	1.032	47.0	19.1	8.87		.938		.499					56.0	21.3	.059	215	
		(25)															
.067	1.032	47.0	24.0	8.89		.933		.487					14.5	25.5	37.0	.532	
.063	1.023	46.7	23.7	8.79		.931		.498					28.9	27.6	.36.3	649	
.065	1.020	46.6	24.3	9.01		.939		.483					50.7	27.2	.32.3	312	
.066	1.025	47.9	24.1	9.05		.985		.486					56.9	22.3	.052	227	
		(30)															
.068	1.061	46.3	28.2	8.74		.952		.488					14.5	23.2	35.3	.118	
.067	1.039	46.6	28.6	8.55		.974		.484					29.2	24.4	.34.7	.221	
.065	1.008	46.2	29.3	9.08		.960		.475					50.7	27.8	.31.9	.115	
.063	1.002	50.0	30.4	9.43		.957		.463					57.6	21.9	.044	205	
		(75)	(20)														
.068	1.023	65.6	18.9	8.77		.958		.505					13.8	10.0	34.2	.513	
.067	1.026	71.1	19.1	8.55		.941		.503					26.5	12.9	34.5	.268	
.064	1.002	73.9	20.0	9.28		.914		.478					48.6	12.5	29.2	.121	
.066	1.023	72.0	19.2	8.93		.927		.503					54.0	13.5	19.5	.048	
		(25)															
.068	1.010	70.5	23.7	8.80		.932		.494					14.3	11.5	34.2	.430	
.067	1.020	70.7	23.9	8.88		.984		.492					28.1	11.9	33.0	.206	
.067	1.040	72.0	23.9	8.88		.956		.490					49.6	12.2	29.6	.104	
.067	1.042	71.6	24.0	8.89		.990		.488					55.5	12.5	20.0	.041	
		(30)															
.063	.972	74.8	30.3	9.40		.921		.462					14.4	10.5	31.7	.340	
.064	.980	73.8	29.7	9.21		.979		.470					26.6	13.1	32.6	.178	
.066	1.002	72.1	29.2	9.04		.963		.477					50.5	12.6	29.5	.092	
.062	.956	76.6	30.8	9.53		.947		.454					26.5	11.8	20.5	.036	
		(30)															

* Lengths are for the actual test specimens for which $c = 3.75$ approximately.

NATIONAL ADVISORY
COMMITTEE FOR AERONAUTICS

TABLE 4.— Concluded
 TEST DATA AND PROPORTIONS OF SPECIMENS HAVING $\frac{t_w}{t_s} = 1.00$ — Concluded

(b) ALCLAD 75S-T SHEET AND 75S-T STIFFENERS

t_w	$\frac{t_w}{t_s}$	Proportions of test specimens												Test data					
		b_s	$\frac{b_s}{t_s}$	b_w	$\frac{b_w}{t_w}$	b_A	$\frac{b_A}{t_w}$	t_w	$\frac{t_w}{t_p}$	b_w	$\frac{b_w}{b_p}$	x	$\frac{x}{t_w}$	d	$\frac{d}{t_s}$	$\frac{L}{t_w}$	σ_{cr} (ksi)	$\bar{\sigma}_f$ (ksi)	$\frac{E_1}{L/\sqrt{G}}$ (psi/in.)
(0.064)	(1.00)	(25)	(20)	(9.3)	(0.96)	(0.94)	(1.07)	0.47	1.44	(1)	(2.14)	(7.8)							
.064	1.003	24.5	20.0	9.31				0.876	0.452					14.6	60.8	64.0	1.350	615×10^5	
.063	1.002	24.8	20.3	9.43				.933	0.475					29.2	61.1	61.6	.638	593	
.064	1.019	24.9	19.9	9.24				.927	0.484					50.9	56.2	58.0	.345	551	
.065	1.027	24.2	19.7	9.14				.929	0.491					67.6	---	25.3	.087	237	
.065	.949	24.0	24.6	9.14				.902	0.479					14.7	48.9	57.4	1.048	583	
.064	.973	23.6	24.9	9.21				.921	0.471					29.4	42.4	58.7	.533	600	
.065	.991	24.2	24.7	9.18				.907	0.477					51.3	---	52.9	.269	524	
.062	.955	(35)	(20)					.833	0.467					14.5	47.7	58.3	1.095	650	
.064	.997	34.3	20.8	9.66				.896	0.455					29.0	48.7	56.6	.534	547	
.065	1.006	34.5	20.8	9.30				.945	0.461					30.7	45.6	52.3	.278	497	
.064	.998	34.1	20.1	9.35				.946	0.461					87.1	---	25.9	.081	219	
.061	1.013	34.8	25.0	9.27				.912	0.478					14.6	42.4	54.8	.877	592	
.062	.989	33.7	24.8	9.21				.926	0.477					29.2	44.7	53.9	.429	512	
.066	1.027	35.1	24.4	9.07				.928	0.483					51.1	47.4	49.4	.223	522	
.065	1.008	33.6	(30)	9.08				.901	0.471					14.7	33.3	48.3	.690	560	
.065	1.011	34.4	29.4	9.10				.903	0.468					29.3	27.5	48.3	.343	520	
.065	1.019	34.5	29.6	9.15				.906	0.468					51.3	26.2	44.7	.179	457	
.061	.914	(50)	(20)	9.74				.852	0.429					14.3	23.3	55.6	.925	704	
.062	.967	48.8	21.0	9.74				.858	0.452					28.6	25.2	27.0	.475	541	
.062	.963	49.1	20.6	9.58				.927	0.466					50.2	25.7	50.1	.235	524	
.062	.946	48.9	20.5	9.52				.858	0.452					86.0	21.6	24.8	.069	255	
.065	.951	46.9	21.5	9.08				.910	0.483					14.5	25.7	53.1	.769	666	
.065	.986	46.3	24.7	9.18				.956	0.485					29.2	24.5	50.6	.356	551	
.063	.952	47.4	25.2	9.37				.922	0.470					50.7	24.8	48.5	.197	511	
.064	.949	47.5	25.2	9.35				.968	0.461					87.0	23.3	24.8	.058	250	
.065	1.014	50.2	29.7	9.21				.896	0.482					14.6	23.8	47.1	.586	517	
.064	1.022	50.3	29.9	9.25				.908	0.462					29.1	21.9	47.9	.288	505	
.065	1.002	49.1	29.7	9.21				.901	0.463					50.6	21.9	44.3	.161	465	
.063	.989	49.8	30.5	9.46				.904	0.454					87.4	23.3	24.9	.051	260	
.062	.960	74.2	20.7	9.63				.848	0.437					14.0	9.2	49.3	.717	638	
.061	.953	74.6	21.1	9.82				.853	0.427					28.0	11.9	49.6	.360	593	
.067	1.059	75.2	19.1	8.88				.964	0.499					48.9	11.1	46.9	.194	581	
.062	.967	74.6	20.8	9.64				.866	0.436					84.1	13.4	22.9	.052	236	
.064	.933	70.4	25.1	9.32				.974	0.471					14.3	11.7	47.7	.520	839	
.064	1.002	74.5	25.0	9.28				.910	0.472					28.5	10.4	47.2	.289	558	
.064	.961	71.6	25.1	9.32				.967	0.460					49.8	11.7	43.8	.155	543	
.064	.973	71.7	24.9	9.25				.946	0.461					84.2	12.0	23.2	.049	260	
.067	1.026	73.3	28.8	8.93				.921	0.479					14.4	11.1	43.5	.477	545	
.064	1.005	73.9	29.8	9.22				.887	0.462					28.8	10.7	42.6	.232	539	
.066	1.022	74.3	29.2	9.04				.907	0.472					50.4	11.8	43.4	.133	486	
.060	.926	73.0	31.8	9.83				.878	0.442					86.5	11.6	22.9	.041	258	

^aLengths are for the actual test specimens for which $c = 3.75$ approximately.

NATIONAL ADVISORY
COMMITTEE FOR AERONAUTICS

TABLE 5.- COMPARATIVE DESIGNS OF Y- AND Z-STIFFENED PANELS ($P_c = 2.0$ kips/inch; $\epsilon = 1$; minimum-weight designs in parentheses).

TABLE 5. - COMPARATIVE DESIGNS OF Y- AND Z-SUPPORTED PANELS - Concluded

NATIONAL ADVISORY
COMMITTEE FOR AERONAUTICS

t_w	t_b	Panel	$L = 10$ in.	$L = 20$ in.
0.020	0.025	(0.025) A-36	0.025	0.025

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COMMITTEE FOR AERONAUTICS

TABLE 6.- COMPARATIVE DESIGNS OF Y- AND Z-STIFFENED PANELS - Concluded

		L = 30 in.										L = 40 in.												
		t _g , in.										t _g , in.												
		Panel										Panel												
σ_x (ksi)	$\frac{t_y}{t_g}$.40	$\frac{Y}{Z}$	755-1								.025	.032	.040	.051	.064	.081	.102	.125	.156	.188			
		.51	$\frac{Y}{Z}$	755-1									.025	.032	.040	.051	.064	.081	.102	.125	.156	.188		
		.63	$\frac{Y}{Z}$	755-1										.025	.032	.040	.051	.064	.081	.102	.125	.156	.188	
		.79	$\frac{Y}{Z}$	755-1										.025	.032	.040	.051	.064	.081	.102	.125	.156	.188	
		1.00	$\frac{Y}{Z}$	755-1										.025	.032	.040	.051	.064	.081	.102	.125	.156	.188	
σ_{xy} (ksi)	$\frac{t_y}{t_g}$.40	$\frac{Y}{Z}$	755-1																				
		.51	$\frac{Y}{Z}$	755-1																				
		.63	$\frac{Y}{Z}$	755-1																				
		.79	$\frac{Y}{Z}$	755-1																				
		1.00	$\frac{Y}{Z}$	755-1																				
κ (in.)	$\frac{t_y}{t_g}$.40	$\frac{Y}{Z}$	755-1																				
		.51	$\frac{Y}{Z}$	755-1																				
		.63	$\frac{Y}{Z}$	755-1																				
		.79	$\frac{Y}{Z}$	755-1																				
		1.00	$\frac{Y}{Z}$	755-1																				
s (in.)	$\frac{t_y}{t_g}$.40	$\frac{Y}{Z}$	755-1																				
		.51	$\frac{Y}{Z}$	755-1																				
		.63	$\frac{Y}{Z}$	755-1																				
		.79	$\frac{Y}{Z}$	755-1																				
		1.00	$\frac{Y}{Z}$	755-1																				
κ (in.)	$\frac{t_y}{t_g}$.40	$\frac{Y}{Z}$	755-1																				
		.51	$\frac{Y}{Z}$	755-1																				
		.63	$\frac{Y}{Z}$	755-1																				
		.79	$\frac{Y}{Z}$	755-1																				
		1.00	$\frac{Y}{Z}$	755-1																				
E (in.)	$\frac{t_y}{t_g}$.40	$\frac{Y}{Z}$	755-1																				
		.51	$\frac{Y}{Z}$	755-1																				
		.63	$\frac{Y}{Z}$	755-1																				
		.79	$\frac{Y}{Z}$	755-1																				
		1.00	$\frac{Y}{Z}$	755-1																				
P (in.)	$\frac{t_y}{t_g}$.40	$\frac{Y}{Z}$	755-1																				
		.51	$\frac{Y}{Z}$	755-1																				
		.63	$\frac{Y}{Z}$	755-1																				
		.79	$\frac{Y}{Z}$	755-1																				
		1.00	$\frac{Y}{Z}$	755-1																				
$\frac{t_y}{t_g}$	$\frac{t_z}{t_g}$.40	$\frac{Y}{Z}$	755-1																				
		.51	$\frac{Y}{Z}$	755-1																				
		.63	$\frac{Y}{Z}$	755-1																				
		.79	$\frac{Y}{Z}$	755-1																				
		1.00	$\frac{Y}{Z}$	755-1																				
$\frac{t_y}{t_g}$	$\frac{t_z}{t_g}$.40	$\frac{Y}{Z}$	755-1																				
		.51	$\frac{Y}{Z}$	755-1																				
		.63	$\frac{Y}{Z}$	755-1																				
		.79	$\frac{Y}{Z}$	755-1																				
		1.00	$\frac{Y}{Z}$	755-1																				
$\frac{t_y}{t_g}$	$\frac{t_z}{t_g}$.40	$\frac{Y}{Z}$	755-1																				
		.51	$\frac{Y}{Z}$	755-1																				
		.63	$\frac{Y}{Z}$	755-1																				
		.79	$\frac{Y}{Z}$	755-1																				
		1.00	$\frac{Y}{Z}$	755-1																				
$\frac{t_y}{t_g}$	$\frac{t_z}{t_g}$.40	$\frac{Y}{Z}$	755-1																				
		.51	$\frac{Y}{Z}$	755-1																				
		.63	$\frac{Y}{Z}$	755-1																				
		.79	$\frac{Y}{Z}$	755-1																				
		1.00	$\frac{Y}{Z}$	755-1																				
$\frac{t_y}{t_g}$	$\frac{t_z}{t_g}$.40	$\frac{Y}{Z}$	755-1																				
		.51	$\frac{Y}{Z}$	755-1																				
		.63	$\frac{Y}{Z}$	755-1																				
		.79	$\frac{Y}{Z}$	755-1																				
		1.00	$\frac{Y}{Z}$	755-1																				
$\frac{t_y}{t_g}$	$\frac{t_z}{t_g}$.40	$\frac{Y}{Z}$	755-1																				
		.51	$\frac{Y}{Z}$	755-1																				
		.63	$\frac{Y}{Z}$	755-1																				
		.79	$\frac{Y}{Z}$	755-1																				
		1.00	$\frac{Y}{Z}$	755-1																				
$\frac{t_y}{t_g}$	$\frac{t_z}{t_g}$.40	$\frac{Y}{Z}$	755-1																				
		.51	$\frac{Y}{Z}$	755-1																				
		.63	$\frac{Y}{Z}$	755-1																				
		.79	$\frac{Y}{Z}$	755-1																				
		1.00	$\frac{Y}{Z}$	755-1																				

TABLE 7.- COMPARATIVE DESIGNS OF T- AND Z-STIFFERED PANELS ($P_1 = 5.0$ kips/inch; $c = 1$; minimum-weight designs in parentheses)

$\frac{t_w}{t_s}$		Panel	L = 10 in.								L = 20 in.											
			t_{eq} , in.								t_{eq} , in.											
0.052		0.040	0.051	0.061	0.082	0.102	0.125	0.156	0.188	0.204	0.040		0.051	0.061	0.081	0.102	0.125	0.156	0.188	0.204		
$\frac{t_w}{t_s}$ (cont.)	0.50	$\frac{75}{21}$			(54.1)	110.8	135.9	152.1	172.0	22.7		(68.0)	142.5	172.3	211.9	257.6	222.5					
	.51	$\frac{75}{21}$			(57.3)	51.5	44.5	37.7				(57.5)	52.4	50.4	23.1							
	.63	$\frac{75}{21}$			(60.2)	142.5	130.5	131.6	126.7	125.2	(21.5)		16.1	11.1	32.0	21.1						
	.79	$\frac{75}{21}$			(60.2)	55.3	49.6	42.5	35.9	29.8	25.2		(149.8)	36.1	32.1	38.0	21.8					
	1.00	$\frac{75}{21}$			(58.5)	52.4	46.4	40.8	36.9	32.5	28.6	24.7	(51.2)	48.4	45.0	31.9	28.2	21.7				
	.40	$\frac{75}{21}$			(58.5)	54.4	49.2	43.5	39.7	34.8	30.5	27.1		(50.9)	35.9	35.5	35.6	30.4	27.0			
	.51	$\frac{75}{21}$			(55.5)	31.0	25.2	18.2	13.0	10.6	10.6	(9.5)		(38.1)	37.0	35.1	34.0	30.0	27.7			
$\frac{t_w}{t_s}$ (cont.)	.63	$\frac{75}{21}$			(58.6)	38.2	17.5	9.7	34.1	22.8	19.1			(41.1)	11.9	12.2	17.7	9.5	8.1			
	.79	$\frac{75}{21}$			(58.5)	20.0	10.4	10.6	16.6	24.1	16.2	9.5		(51.2)	24.0	13.1	35.6	26.4	17.4	9.0		
	1.00	$\frac{75}{21}$			(58.5)	21.0	10.5	13.7	25.9	17.8	11.0			(26.2)	13.1	21.3	8.5	10.8				
	.40	$\frac{75}{21}$			(58.5)	31.5	16.7							(1.36)	1.49	1.49	1.72	2.16	2.76			
	.51	$\frac{75}{21}$			(.84)	1.08	1.35	1.72						(1.86)	1.80	1.90	1.75	2.16	2.76			
	.63	$\frac{75}{21}$			(.88)	1.08	1.35	1.72						(1.85)	1.87	1.75	2.16	2.76				
	.79	$\frac{75}{21}$			(1.17)	1.72	2.16	2.74						(1.43)	1.90	2.11	2.16	2.74	2.63			
t (in.)	1.00	$\frac{75}{21}$			(1.17)	1.72	2.16	2.74						(1.52)	1.53	2.16	2.74					
	.40	$\frac{75}{21}$			(1.17)	1.72	2.16	2.74						(1.58)	2.05	2.16	2.74					
	.51	$\frac{75}{21}$			(1.17)	1.72	2.16	2.74						(1.58)	2.05	2.16	2.74					
	.63	$\frac{75}{21}$			(1.17)	1.72	2.16	2.74						(1.59)	2.28	2.62	2.74					
	.79	$\frac{75}{21}$			(1.17)	1.72	2.16	2.74						(1.50)	1.96	2.16	2.74					
	1.00	$\frac{75}{21}$			(1.17)	1.72	2.16	2.74						(1.23)	2.01	2.80	3.98	5.96	6.41			
	.40	$\frac{75}{21}$			(1.17)	1.72	2.16	2.74						(1.23)	2.01	2.80	3.98	5.96	6.41			
t (in.)	.51	$\frac{75}{21}$			(.93)	2.75	4.27	4.27						(1.24)	2.28	2.62	2.74					
	.63	$\frac{75}{21}$			(.76)	1.13	2.21	2.69	3.93	6.18	9.10				(1.33)	2.15	3.28	3.62	5.78	13.6		
	.79	$\frac{75}{21}$			(1.08)	1.35	1.72	2.16	2.74						(1.33)	2.15	3.28	3.62	5.78			
	1.00	$\frac{75}{21}$			(1.08)	1.72	2.16	2.74						(1.09)	1.72	2.16	2.74		11.3			
	.40	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.51	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.63	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
t (in.)	.79	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	1.00	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.40	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.51	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.63	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.79	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	1.00	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
t (in.)	.40	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.51	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.63	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.79	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	1.00	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.40	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.51	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
t (in.)	.63	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.79	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	1.00	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.40	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.51	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.63	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.79	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
t (in.)	1.00	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.40	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.51	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.63	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.79	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	1.00	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.40	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
t (in.)	.51	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.63	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.79	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	1.00	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.40	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.51	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07						(1.33)	2.15	3.28	3.62	5.78			
	.63	$\frac{75}{21}$			(1.15)	1.55	3.22	4.73	7.07													

TABLE 7.-COMPARATIVE DESIGNS OF X- AND X-STIFFENED PANELS - Concluded

				L = 40 in.																	
				t _x , in.						t _x , in.											
		Panel		0.040	0.051	0.064	0.081	0.102	0.125	0.156	0.188	0.204	0.040	0.051	0.064	0.081	0.102	0.125	0.156	0.188	0.204
$\frac{E}{E_x}$ (in.)	.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
	1.00	X 755-T																			
	.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
E_{ex} (in.)	.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
	1.00	X 755-T																			
	.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
$\frac{E}{(in.)}$.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
	1.00	X 755-T																			
	.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
$\frac{E}{(in.)}$.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
	1.00	X 755-T																			
	.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
$\frac{E}{(in.)}$.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
	1.00	X 755-T																			
	.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
$\frac{E}{(in.)}$.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
	1.00	X 755-T																			
	.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
$\frac{E}{(in.)}$.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
	1.00	X 755-T																			
	.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
$\frac{E}{(in.)}$.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
	1.00	X 755-T																			
	.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
$\frac{E}{(in.)}$.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
	1.00	X 755-T																			
	.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
$\frac{E}{(in.)}$.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
	1.00	X 755-T																			
	.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
$\frac{E}{(in.)}$.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
	1.00	X 755-T																			
	.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
$\frac{E}{(in.)}$.40	X 755-T																			
	.51	X 755-T																			
	.65	X 755-T																			
	.79	X 755-T																			
	1.00	X																			

TABLE 8.- COMPARATIVE DESIGNS OF Y- AND Z-STIFFENED PANELS [$F_1 = 5.0$ kips/inch; $c = 1$; minimum-weight designs in parentheses]

t_w t_g	Panel	L = 10 in.										L = 20 in.										
		t_g , in.					t_g , in.					t_g , in.					t_g , in.					
J_F (in.)	0.40 Y 758-1	0.040 (55.3)	0.051 (48.5)	0.064 (41.2)	0.081 (35.3)	0.108 (37.1)	0.125 (32.9)	0.156 (32.9)	0.188 (31.1)	0.208 (26.5)	0.250 (26.5)	0.054 (56.3)	0.065 (46.9)	0.081 (40.9)	0.108 (36.5)	0.125 (32.7)	0.156 (31.1)	0.188 (26.7)	0.208 (26.7)	0.250 (26.7)		
	0.51 Y 758-1					(58.2)	52.1	46.1	39.2	32.9	30.9					(56.3)	50.7 (41.5)	45.1 (38.5)	38.6 (34.5)	32.7 (30.7)	29.1 (25.1)	
	0.63 Y 758-1					(60.6)	55.7	49.9	44.2	37.2	32.9	31.1	29.4	25.8		(58.6)	54.9 (46.6)	49.3 (41.9)	43.9 (37.7)	37.8 (33.8)	32.6 (28.7)	
	0.79 Y 758-1					(60.8)	57.9	52.9	47.4	41.2	37.6	34.9	32.8	28.3	24.7		56.8 (57.6)	58.7 (58.1)	52.1 (49.2)	45.8 (42.8)	39.5 (31.2)	32.4 (28.2)
	1.00 Y 758-1					(60.8)	55.9	54.6	49.6	40.0	36.6	33.1	26.8	26.8		(57.6)	54.2 (50.2)	49.3 (43.7)	43.2 (36.2)	37.8 (31.2)	26.7 (22.2)	
	1.40 Y 758-1					(63.4)	55.7	51.4	46.8	41.9	36.8	31.9	10.8	10.8		(52.7)	51.0 (46.9)	49.2 (43.0)	42.1 (37.0)	37.0 (31.0)	12.1 (11.0)	
σ_{sp} (lb/in.)	.51 Y 758-1					(54.0)	28.0	15.6	9.0	3.3	2.8	2.4	29.1 (18.4)			(56.3)	36.0 (41.5)	16.9 (18.1)	10.6 (18.5)	5.6 (5.1)	32.7 (29.6)	
	.63 Y 758-1					(59.0)	38.0	17.3	9.8	3.3	2.8	2.4	13.0			(57.2)	40.1 (38.2)	18.1 (18.9)	10.6 (17.6)	5.6 (5.1)	32.7 (29.6)	
	.79 Y 758-1					(58.6)	43.7	21.1	10.3	3.6	2.8	2.4	15.2	9.3		(48.4)	21.5 (33.8)	11.1 (17.8)	10.0 (15.1)	5.6 (5.1)	32.7 (29.6)	
	1.00 Y 758-1					(57.5)	42.2	21.6	10.8	3.6	2.8	2.4	10.8			(47.5)	22.2 (40.2)	14.0	17.2	9.9	12.7 (10.6)	
	1.40 Y 758-1					(1.35)	1.78	2.16	2.74	3.14	3.14	3.14	1.78	1.78		(2.69)	1.78 (2.15)	2.16 (2.43)	2.74 (2.74)	3.14 (3.14)	1.78 (1.78)	
	.51 Y 758-1					(1.35)	1.78	2.16	2.74	3.14	3.14	3.14	1.78	1.78		(1.54)	1.78 (2.15)	2.16 (2.43)	2.74 (2.74)	3.14 (3.14)	1.78 (1.78)	
H (in.)	.63 Y 758-1					(1.36)	1.72	2.16	2.74	3.14	3.14	3.14	1.78	1.78		(1.78)	2.16 (2.45)	2.16 (2.45)	2.74 (2.74)	3.14 (3.14)	1.78 (1.78)	
	.79 Y 758-1					(1.49)	1.72	2.16	2.74	3.14	3.14	3.14	1.78	1.78		(1.78)	2.16 (2.45)	2.16 (2.45)	2.74 (2.74)	3.14 (3.14)	1.78 (1.78)	
	1.00 Y 758-1					(1.57)	1.72	2.16	2.74	3.14	3.14	3.14	1.78	1.78		(1.78)	2.16 (2.45)	2.16 (2.45)	2.74 (2.74)	3.14 (3.14)	1.78 (1.78)	
	1.40 Y 758-1					(1.76)	1.78	2.16	2.74	3.14	3.14	3.14	1.78	1.78		(1.78)	2.16 (2.45)	2.16 (2.45)	2.74 (2.74)	3.14 (3.14)	1.78 (1.78)	
	.51 Y 758-1					(1.76)	1.78	2.16	2.74	3.14	3.14	3.14	1.78	1.78		(1.78)	2.16 (2.45)	2.16 (2.45)	2.74 (2.74)	3.14 (3.14)	1.78 (1.78)	
	.63 Y 758-1					(1.76)	1.78	2.16	2.74	3.14	3.14	3.14	1.78	1.78		(1.78)	2.16 (2.45)	2.16 (2.45)	2.74 (2.74)	3.14 (3.14)	1.78 (1.78)	
S (in.)	.51 Y 758-1					(1.52)	2.65	4.20	6.22	8.23	3.45	3.45	3.28			(1.50)	2.47 (2.03)	3.86 (3.84)	6.22	8.23	3.28	
	.63 Y 758-1					(1.21)	2.09	3.58	5.59	8.02	4.03	4.03	3.95			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	.79 Y 758-1					(1.09)	1.74	2.92	4.85	6.22	3.55	3.55	3.45			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	1.00 Y 758-1					(.99)	1.54	2.48	4.08	6.22	3.45	3.45	3.38			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	1.40 Y 758-1					(.77)	4.44	6.68	8.62	10.2	8.62	8.62	8.62			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	.51 Y 758-1					(.258)	1.268	2.91	4.32	5.26	3.22	3.22	3.14			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
H (in.)	.63 Y 758-1					(.328)	345	562	822	1100	552	552	544			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	.79 Y 758-1					(.386)	.414	.444	.464	.484	.401	.439	.428			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	1.00 Y 758-1					(.477)	.483	.535	.562	.593	.509	.535	.521			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	1.40 Y 758-1					(.615)	.626	.707	.749	.782	.682	.742	.785			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	.51 Y 758-1					(.727)	.834	.949	1.06	1.06	.742	.742	.785			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	.63 Y 758-1					(.727)	.834	.949	1.06	1.06	.742	.742	.785			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
P (in.)	.51 Y 758-1					(.483)	.567	.650	.722	.784	.623	.682	.758			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	.63 Y 758-1					(.506)	.605	.714	.803	.862	.671	.739	.806			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	.79 Y 758-1					(.567)	.633	.771	.927	.862	.738	.862	.901			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	1.00 Y 758-1					(.615)	.658	.996	1.06	1.06	.742	.742	.785			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	1.40 Y 758-1					(1.80)	18.0	18.0	18.0	18.0	18.0	18.0	18.0			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	.51 Y 758-1					(18.0)	18.0	18.0	18.0	18.0	18.0	18.0	18.0			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
b_w t_w	.63 Y 758-1					(18.6)	18.0	18.0	18.0	18.0	18.0	18.0	18.0			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	.79 Y 758-1					(18.6)	18.0	18.0	18.0	18.0	18.0	18.0	18.0			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	1.00 Y 758-1					(21.0)	18.0	18.0	18.0	18.0	18.0	18.0	18.0			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	1.40 Y 758-1					(18.2)	18.0	18.0	18.0	18.0	18.0	18.0	18.0			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	.51 Y 758-1					(19.6)	18.0	18.0	18.0	18.0	18.0	18.0	18.0			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
	.63 Y 758-1					(19.6)	18.0	18.0	18.0	18.0	18.0	18.0	18.0			(1.26)	2.02 (2.03)	3.47 (3.47)	6.02	8.02	3.95	
b_g t_g	.51 Y 758-1					(25.9)	40.3	55.2	72.9	94.8	58.8	65.3			(23.0)	35.7 (32.0)	49.8 (30.2)	67.9 (41.3)				
	.63 Y 758-1					(23.0)	36.8	54.5	74.2	93.2	59.7	64.2	58.4			(23.0)	35.1 (32.0)	53.2 (32.8)	72.4 (48.2)			
	.79 Y 758-1					(23.0)	37.1	53.6	74.7	93.2	59.1	65.7	58.4			(33.5)	51.4 (38.0)	57.5 (36.0)	75.3 (53.0)			
	1.00 Y 758-1					(23.0)	37.1	54.1	75.8	90.5	59.1	65.7	58.4			(35.5)	51.9 (37.5)	57.5 (36.0)	75.3 (53.0)			
	1.40 Y 758-1					(26.8)	47.1	59.8	80.8	104.0	64.6	70.9				(35.5)	51.9 (37.5)	57.5 (36.0)	75.3 (53.0)			
	.51 Y 758-1					(26.8)	47.1	59.8	80.8	104.0	64.6	70.9				(35.5)	51.9 (37.5)	57.5 (36.0)	75.3 (53.0)			
b_g t_g	.63 Y 758-1					(26.8)	47.1	59.8	80.8	104.0	64.6	70.9				(35.5)	51.9 (37.5)	57.5 (36.0)	75.3 (53.0)			
	.79 Y 758-1					(23.0)	37.1	54.1	75.8	90.5	59.1	65.7	58.4			(35.5)	51.9 (37.5)	57.5 (36.0)	75.3 (53.0)			
b_g t_g	1.00 Y 758-1					(23.0)	37.1	54.1	75.8	90.5	59.1	65.7	58.4			(35.5)	51.9 (37.5)	57.5 (36.0)	75.3 (53.0)			
	1.40 Y 758-1					(26.8)	47.1	59.8	80.8	104.0	64.6	70.9				(35.5)	51.9 (37.5)	57.5 (36.0)	75.3 (53.0)			

NATIONAL ADVISORY
COMMITTEE FOR AERONAUTICS

TABLE 7. - COMPARATIVE DESIGNS OF Y- AND X-SHIPPED PANELS - Unpublished

TABLE 8.- COMPARATIVE DESIGNS OF Y- AND Z-STIFFENED PANELS [$P_y = 5.0$ kips/inch; $c = 1$; minimum-weight designs in parentheses]

		L = 10 in.										L = 20 in.										
		t _w , in.					t _s , in.					t _w , in.					t _s , in.					
		Panel	t _w	t _s	t _w	t _s	t _w	t _s	t _w	t _s	t _w	t _s	t _w	t _s	t _w	t _s	t _w	t _s	t _w	t _s		
δ_T (in.)	0.40	A-1	.755	.5	0.040	0.051	0.064	0.081	0.108	0.125	0.156	0.188	0.204	0.250	0.051	0.064	0.081	0.108	0.149	0.156	0.188	0.204
	.51	A-2	.755	.5					(55.3)	(41.2)	(39.6)	(35.3)	(32.8)	(26.5)					(52.7)	(46.9)	(40.1)	(34.4)
	.63	A-3	.755	.5					(58.2)	52.1	46.1	39.2	32.9	31.1	(26.7)				(55.5)	(56.3)	(57.5)	(26.8)
	.79	A-4	.755	.5					(60.6)	55.7	49.9	44.2	39.3	34.4	30.7				(50.9)	(36.5)	(35.6)	(32.7)
	1.00	A-5	.755	.5					(60.8)	57.9	52.9	47.3	39.8	32.6	29.8				(31.1)	(31.1)	(35.7)	(35.7)
									(60.4)	58.9	54.6	49.6	40.5	38.2	34.9							
σ_{ex} (lb/in.)	.40	B-1	.755	.5					(43.4)	(33.3)	(28.4)	(24.8)	(18.6)	(10.4)					(40.9)	(37.0)	(32.7)	(29.6)
	.51	B-2	.755	.5					(54.0)	26.0	15.6	9.6	2.0	16.6	13.1				(41.5)	36.0	16.9	10.6
	.63	B-3	.755	.5					(59.0)	36.0	17.3	9.8	4.9	25.1	20.9				(41.5)	27.8	16.8	10.6
	.79	B-4	.755	.5					(58.6)	43.7	23.1	10.3	3.6	36.0	17.5				(57.2)	40.1	18.1	10.6
	1.00	B-5	.755	.5					(40.0)	42.2	21.6	10.8	3.1	29.0	18.8				(40.8)	22.2	11.1	10.6
									(40.9)	42.2	21.6	10.8	3.1	26.6	9.9				(40.8)	22.2	11.1	10.6
M (in.)	.40	C-1	.755	.5					(1.35)	1.78	1.72	1.65	1.56	1.45	1.40				(1.35)	1.78	1.65	1.40
	.51	C-2	.755	.5					(1.35)	1.78	2.16	2.16	2.14	2.14	2.03				(2.69)	1.78	2.16	2.03
	.63	C-3	.755	.5					(1.36)	1.78	2.16	2.74	2.14	2.03	3.45				(2.36)	1.78	2.16	2.03
	.79	C-4	.755	.5					(1.49)	1.72	2.16	2.74	3.45	4.28	3.28				(1.54)	1.72	2.16	2.74
	1.00	C-5	.755	.5					(1.57)	1.78	2.16	2.74	3.45	4.28	3.28				(1.54)	1.78	2.16	2.74
									(1.56)	1.78	2.16	2.63	3.28	3.95	4.28				(1.54)	1.78	2.16	2.74
S (in.)	.40	D-1	.755	.5					(1.96)	1.74	1.74	1.72	1.65	1.56	1.40				(1.68)	1.74	1.74	1.72
	.51	D-2	.755	.5					(1.52)	2.65	4.20	4.20	4.20	4.20	4.20				(2.33)	1.74	1.74	1.72
	.63	D-3	.755	.5					(1.79)	2.74	3.45	3.45	3.45	3.45	3.45				(1.40)	2.62	3.86	3.86
	.79	D-4	.755	.5					(1.49)	1.72	2.16	2.74	3.45	4.28	3.28				(1.40)	2.62	3.86	3.86
	1.00	D-5	.755	.5					(1.57)	1.78	2.16	2.74	3.45	4.28	3.28				(1.40)	2.62	3.86	3.86
									(1.56)	1.78	2.16	2.74	3.45	4.28	3.28				(1.40)	2.62	3.86	3.86
H (in.)	.40	E-1	.755	.5					(1.96)	1.74	1.74	1.72	1.65	1.56	1.40				(1.68)	1.74	1.74	1.72
	.51	E-2	.755	.5					(1.52)	2.65	4.20	4.20	4.20	4.20	4.20				(2.33)	1.74	1.74	1.72
	.63	E-3	.755	.5					(1.21)	2.09	3.54	5.57	5.76	5.66	5.66				(1.40)	2.62	3.86	3.86
	.79	E-4	.755	.5					(1.09)	1.74	2.98	4.86	4.93	7.76	7.76				(1.40)	2.62	3.86	3.86
	1.00	E-5	.755	.5					(2.03)	1.54	1.54	1.54	1.54	1.54	1.54				(1.40)	2.62	3.86	3.86
									(0.99)	1.54	2.48	4.08	4.95	6.04	12.2				(1.40)	2.62	3.86	3.86
P (in.)	.40	F-1	.755	.5					(.298)	.565	.565	.565	.565	.565	.565				(.280)	.565	.565	.565
	.51	F-2	.755	.5					(.328)	.745	.745	.745	.745	.745	.745				(.346)	.745	.745	.745
	.63	F-3	.755	.5					(.486)	.745	.745	.745	.745	.745	.745				(.450)	.745	.745	.745
	.79	F-4	.755	.5					(.477)	.483	.535	.588	.789	.872	.872				(.452)	.535	.535	.535
	1.00	F-5	.755	.5					(.569)	.551	.628	.707	.742	.786	.807				(.557)	.629	.709	.741
									(.615)	.658	.808	.956	1.06	1.06	1.06				(.615)	.808	.956	1.06
b_w t_w	.40	G-1	.755	.5					(1.428)	1.484	1.560	1.540	1.577	1.515	1.40				(1.446)	1.562	1.552	1.500
	.51	G-2	.755	.5					(1.483)	.567	.650	.689	.965	1.02	1.02				(1.498)	.565	.656	.770
	.63	G-3	.755	.5					(.506)	.609	.715	.829	.918	.918	.918				(1.582)	.609	.715	.829
	.79	G-4	.755	.5					(1.567)	.633	.771	.923	.989	1.17	1.40				(1.635)	.774	.927	.989
	1.00	G-5	.755	.5					(1.724)	1.619	.682	1.00	1.40	1.34	1.38				(1.659)	.808	.956	1.14
									(1.615)	.658	.808	.956	1.14	1.34	1.38				(1.615)	.808	.956	1.14
b_g t_s	.40	H-1	.755	.5					(18.0)	18.0	18.0	18.0	18.0	18.0	18.0				(18.0)	18.0	18.0	18.0
	.51	H-2	.755	.5					(15.0)	18.0	18.0	18.0	18.0	18.0	18.0				(15.2)	18.0	18.0	18.0
	.63	H-3	.755	.5					(18.2)	18.0	18.0	18.0	18.0	18.0	18.0				(20.6)	18.0	18.0	18.0
	.79	H-4	.755	.5					(19.6)	18.0	18.0	18.0	18.0	18.0	18.0				(18.0)	18.0	18.0	18.0
	1.00	H-5	.755	.5					(21.0)	18.0	18.0	18.0	18.0	18.0	18.0				(18.0)	18.0	18.0	18.0
									(21.0)	18.0	18.0	18.0	18.0	18.0	18.0				(21.0)	18.0	18.0	18.0
b_g t_s	.40	I-1	.755	.5					(29.2)	154.8	156.5	156.5	156.5	156.5	156.5				(23.0)	35.7	49.5	56.5
	.51	I-2	.755	.5					(25.9)	40.3	55.2	57.2	57.2	57.2	57.2				(23.0)	35.7	49.5	56.5
	.63	I-3	.755	.5					(23.0)	56.8	58.5	74.2	91.9	91.9	91.9				(23.0)	35.7	49.5	56.5
	.79	I-4	.755	.5					(23.0)	55.7	53.6	51.2	45.2	45.2	45.2				(23.0)	35.7	49.5	56.5
	1.00	I-5	.755	.5					(23.0)	53.6	53.6	51.2	41.9	58.5	76.4				(35.5)	53.9	76.9	91.3
									(23.0)	53.6	53.6	51.2	41.9	58.5	80.8				(27.5)	42.3	62.6	83.5

TABLE 8.- COMPARATIVE DESIGNS OF Y- AND Z-STIFFENED PANELS - Concluded

TABLE 9.— X-PANEL PROPERTIES [$\frac{t_w}{t_s} = 0.40$; $\frac{b_a}{t_y} = 9.3$; $\frac{b_y}{t_y} = 1.06$; $\frac{b_u}{t_y} = 0.94$; $\frac{b_p}{t_y} = 2.13$; $\frac{b_r}{t_y} = 0.69$; $E = 1.1$; $\frac{d}{t_y} = 1.5$; $\frac{r}{t_y} = 4.6$]

$\frac{t_w}{t_s}$	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33
23	1.496	1.512	1.527	1.542	1.557	1.571	1.585	1.598	1.612	1.626	1.637	1.649	1.661	1.673	1.685	1.696
24	1.481	1.497	1.512	1.526	1.541	1.555	1.568	1.582	1.595	1.608	1.620	1.632	1.644	1.656	1.667	1.678
25	1.467	1.482	1.497	1.512	1.526	1.541	1.555	1.568	1.582	1.595	1.608	1.616	1.628	1.639	1.650	1.668
26	1.453	1.469	1.483	1.497	1.511	1.525	1.538	1.552	1.564	1.576	1.588	1.600	1.612	1.623	1.635	1.646
27	1.441	1.456	1.470	1.484	1.498	1.511	1.524	1.537	1.550	1.562	1.574	1.586	1.597	1.608	1.619	1.630
28	1.429	1.444	1.458	1.472	1.485	1.498	1.511	1.524	1.536	1.548	1.560	1.572	1.583	1.594	1.605	1.616
29	1.418	1.432	1.446	1.460	1.473	1.486	1.498	1.511	1.523	1.535	1.547	1.558	1.569	1.580	1.591	1.602
30	1.407	1.421	1.435	1.448	1.461	1.474	1.486	1.499	1.511	1.523	1.535	1.547	1.558	1.569	1.580	1.591
31	1.397	1.411	1.424	1.437	1.450	1.463	1.477	1.487	1.499	1.511	1.523	1.535	1.547	1.558	1.569	1.580
32	1.388	1.401	1.414	1.427	1.440	1.453	1.466	1.479	1.491	1.503	1.515	1.527	1.538	1.549	1.560	1.571
33	1.378	1.391	1.404	1.417	1.430	1.442	1.454	1.467	1.479	1.491	1.503	1.515	1.527	1.538	1.549	1.560
34	1.370	1.383	1.395	1.408	1.420	1.433	1.445	1.457	1.469	1.481	1.493	1.505	1.517	1.529	1.541	1.553
35	1.363	1.375	1.387	1.399	1.411	1.423	1.435	1.447	1.459	1.471	1.483	1.495	1.507	1.519	1.531	1.543
36	1.356	1.368	1.380	1.392	1.404	1.416	1.428	1.440	1.452	1.464	1.476	1.488	1.500	1.512	1.524	1.536
37	1.349	1.361	1.373	1.385	1.397	1.409	1.421	1.433	1.445	1.457	1.469	1.481	1.493	1.505	1.517	1.529
38	1.342	1.354	1.366	1.378	1.390	1.402	1.414	1.426	1.438	1.450	1.462	1.474	1.486	1.498	1.510	1.522
39	1.335	1.347	1.359	1.371	1.383	1.395	1.407	1.419	1.431	1.443	1.455	1.467	1.479	1.491	1.503	1.515
40	1.329	1.341	1.353	1.365	1.377	1.389	1.401	1.413	1.425	1.437	1.449	1.461	1.473	1.485	1.497	1.509
41	1.318	1.330	1.342	1.354	1.366	1.378	1.390	1.402	1.414	1.426	1.438	1.450	1.462	1.474	1.486	1.498
42	1.309	1.321	1.333	1.345	1.357	1.369	1.381	1.393	1.405	1.417	1.429	1.441	1.453	1.465	1.477	1.489
43	1.300	1.312	1.324	1.336	1.348	1.360	1.372	1.384	1.396	1.408	1.420	1.432	1.444	1.456	1.468	1.480
44	1.289	1.301	1.313	1.325	1.337	1.349	1.361	1.373	1.385	1.397	1.409	1.421	1.433	1.445	1.457	1.469
45	1.279	1.291	1.303	1.315	1.327	1.339	1.351	1.363	1.375	1.387	1.399	1.411	1.423	1.435	1.447	1.459
46	1.269	1.281	1.293	1.305	1.317	1.329	1.341	1.353	1.365	1.377	1.389	1.401	1.413	1.425	1.437	1.449
47	1.261	1.273	1.285	1.297	1.309	1.321	1.333	1.345	1.357	1.369	1.381	1.393	1.405	1.417	1.429	1.441
48	1.254	1.266	1.278	1.290	1.302	1.314	1.326	1.338	1.350	1.362	1.374	1.386	1.398	1.410	1.422	1.434
49	1.247	1.259	1.271	1.283	1.295	1.307	1.319	1.331	1.343	1.355	1.367	1.379	1.391	1.403	1.415	1.427
50	1.239	1.251	1.263	1.275	1.287	1.299	1.311	1.323	1.335	1.347	1.359	1.371	1.383	1.395	1.407	1.419
51	1.231	1.243	1.255	1.267	1.279	1.291	1.303	1.315	1.327	1.339	1.351	1.363	1.375	1.387	1.399	1.411
52	1.224	1.236	1.248	1.260	1.272	1.284	1.296	1.308	1.320	1.332	1.344	1.356	1.368	1.380	1.392	1.404
53	1.217	1.229	1.241	1.253	1.265	1.277	1.289	1.301	1.313	1.325	1.337	1.349	1.361	1.373	1.385	1.397
54	1.209	1.221	1.233	1.245	1.257	1.269	1.281	1.293	1.305	1.317	1.329	1.341	1.353	1.365	1.377	1.389
55	1.199	1.211	1.223	1.235	1.247	1.259	1.271	1.283	1.295	1.307	1.319	1.331	1.343	1.355	1.367	1.379
56	1.191	1.203	1.215	1.227	1.239	1.251	1.263	1.275	1.287	1.299	1.311	1.323	1.335	1.347	1.359	1.371
57	1.183	1.195	1.207	1.219	1.231	1.243	1.255	1.267	1.279	1.291	1.303	1.315	1.327	1.339	1.351	1.363
58	1.175	1.187	1.199	1.211	1.223	1.235	1.247	1.259	1.271	1.283	1.295	1.307	1.319	1.331	1.343	1.355
59	1.167	1.179	1.191	1.203	1.215	1.227	1.239	1.251	1.263	1.275	1.287	1.299	1.311	1.323	1.335	1.347
60	1.159	1.171	1.183	1.195	1.207	1.219	1.231	1.243	1.255	1.267	1.279	1.291	1.303	1.315	1.327	1.339
61	1.151	1.163	1.175	1.187	1.199	1.211	1.223	1.235	1.247	1.259	1.271	1.283	1.295	1.307	1.319	1.331
62	1.143	1.155	1.167	1.179	1.191	1.203	1.215	1.227	1.239	1.251	1.263	1.275	1.287	1.299	1.311	1.323
63	1.135	1.147	1.159	1.171	1.183	1.195	1.207	1.219	1.231	1.243	1.255	1.267	1.279	1.291	1.303	1.315
64	1.127	1.139	1.151	1.163	1.175	1.187	1.199	1.211	1.223	1.235	1.247	1.259	1.271	1.283	1.295	1.307
65	1.119	1.131	1.143	1.155	1.167	1.179	1.191	1.203	1.215	1.227	1.239	1.251	1.263	1.275	1.287	1.299
66	1.111	1.123	1.135	1.147	1.159	1.171	1.183	1.195	1.207	1.219	1.231	1.243	1.255	1.267	1.279	1.291
67	1.103	1.115	1.127	1.139	1.151	1.163	1.175	1.187	1.199	1.211	1.223	1.235	1.247	1.259	1.271	1.283
68	1.095	1.107	1.119	1.131	1.143	1.155	1.167	1.179	1.191	1.203	1.215	1.227	1.239	1.251	1.263	1.275
69	1.087	1.099	1.111	1.123	1.135	1.147	1.159	1.171	1.183	1.195	1.207	1.219	1.231	1.243	1.255	1.267
70	1.079	1.091	1.103	1.115	1.127	1.139	1.151	1.163	1.175	1.187	1.199	1.211	1.223	1.235	1.247	1.259
71	1.071	1.083	1.095	1.107	1.119	1.131	1.143	1.155	1.167	1.179	1.191	1.203	1.215	1.227	1.239	1.251
72	1.063	1.075	1.087	1.099	1.111	1.123	1.135	1.147	1.159	1.171	1.183	1.195	1.207	1.219	1.231	1.243
73	1.055	1.067	1.079	1.091	1.103	1.115	1.127	1.139	1.151	1.163	1.175	1.187	1.199	1.211	1.223	1.235
74	1.047	1.059	1.071	1.083	1.095	1.107	1.119	1.131	1.143	1.155	1.167	1.179	1.191	1.203	1.215	1.227
75	1.039	1.051	1.063	1.075	1.087	1.099	1.111	1.123	1.135	1.147	1.159	1.171	1.183	1.195	1.207	1.219
76	1.031	1.043	1.055	1.067	1.079	1.091	1.103	1.115	1.127	1.139	1.151	1.163	1.175	1.187	1.199	1.211
77	1.023	1.035	1.047	1.059	1.071	1.083	1.095	1.107	1.119	1.131	1.143	1.155	1.167	1.179	1.191	1.203
78	1.015	1.027	1.039	1.051	1.063	1.075	1.087	1.109	1.121	1.133	1.145	1.157	1.169	1.181	1.193	1.205
79	1.007	1.019	1.031	1.043	1.055	1.067	1.079	1.091	1.103	1.115	1.127	1.139	1.151	1.163	1.175	1.187
80	999	1.011	1.023	1.035	1.047	1.059	1.071	1.083	1.095	1.107	1.119	1.131	1.143	1.155	1.167	1.179
81	991	1.003	1.015	1.027	1.039	1.051	1.063	1.075	1.087	1.099	1.111	1.123	1.135	1.147	1.159	1.171
82	983	995	1.007	1.019	1.031	1.043	1.055	1.067	1.079	1.091	1.103	1.115	1.127	1.139	1.151	1.163
83	975	987	999	1.011	1.023	1.035	1.047	1.059	1.071	1.083	1.095	1.107	1.119	1.131	1.143	1.155
84	967	979	991	1.003	1.015	1.027	1.039	1.051	1.063	1.075	1.087	1.099	1.111	1.123	1.135	1.147
85	959	971	983	995	1.007	1.019	1.031	1.043	1.055	1.067	1.079	1.091	1.103	1.115	1.127	1.139
86	951	963	975	987	999	1.011	1.023	1.035	1.047	1.059	1.071	1.083	1.095	1.107	1.119	1.131
87	943	955	967	979	991	1.003	1.015	1.027	1.039	1.051	1.063	1.075				

TABLE 10 - Y-PANEL PROPERTIES $\frac{t_y}{t_s} = 0.751$; $\frac{b_y}{t_y} = 9.3$; $\frac{b_x}{t_y} = 1.04$; $\frac{t_x}{t_y} = 1.06$; $\frac{b_z}{t_y} = 0.94$; $\frac{t_z}{t_y} = 2.13$; $\frac{b_p}{t_y} = 0.69$; $\frac{t_p}{t_y} = 1$; $\frac{t_e}{t_y} = 2.0$; $\frac{t_s}{t_y} = 6.0$

$\frac{t_y}{t_s}$	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33
23	1.759	1.770	1.792	1.812	1.832	1.852	1.871	1.889	1.907	1.924	1.941	1.958	1.974	1.989	2.004	2.019
24	1.728	1.749	1.770	1.790	1.810	1.829	1.848	1.866	1.884	1.901	1.918	1.935	1.952	1.968	1.981	1.996
25	1.705	1.729	1.750	1.770	1.789	1.808	1.827	1.845	1.862	1.880	1.896	1.912	1.928	1.944	1.959	1.974
26	1.689	1.710	1.730	1.750	1.768	1.787	1.807	1.824	1.842	1.859	1.875	1.891	1.907	1.923	1.938	1.952
27	1.674	1.692	1.712	1.731	1.750	1.769	1.787	1.805	1.822	1.839	1.857	1.873	1.890	1.907	1.922	1.937
28	1.654	1.674	1.694	1.714	1.732	1.750	1.769	1.786	1.803	1.820	1.836	1.853	1.869	1.883	1.898	1.912
29	1.638	1.658	1.678	1.697	1.715	1.734	1.751	1.769	1.787	1.804	1.821	1.838	1.854	1.870	1.885	1.899
30	1.623	1.642	1.662	1.681	1.699	1.717	1.734	1.752	1.770	1.787	1.804	1.820	1.837	1.853	1.868	1.882
31	1.608	1.628	1.647	1.665	1.683	1.701	1.719	1.735	1.753	1.770	1.788	1.795	1.812	1.829	1.845	1.861
32	1.594	1.613	1.632	1.650	1.668	1.686	1.704	1.722	1.740	1.758	1.775	1.782	1.799	1.816	1.832	1.848
33	1.581	1.600	1.618	1.636	1.654	1.671	1.689	1.707	1.724	1.741	1.758	1.775	1.792	1.809	1.825	1.841
34	1.565	1.587	1.605	1.623	1.641	1.659	1.677	1.695	1.712	1.730	1.747	1.764	1.781	1.798	1.815	1.831
35	1.551	1.569	1.587	1.605	1.623	1.641	1.659	1.677	1.695	1.712	1.730	1.747	1.764	1.781	1.798	1.815
36	1.535	1.552	1.570	1.588	1.606	1.624	1.642	1.660	1.678	1.695	1.712	1.730	1.747	1.764	1.781	1.798
37	1.520	1.538	1.556	1.574	1.592	1.610	1.628	1.646	1.664	1.681	1.698	1.715	1.732	1.749	1.766	1.783
38	1.505	1.523	1.541	1.559	1.577	1.595	1.613	1.631	1.649	1.666	1.683	1.700	1.717	1.734	1.751	1.768
39	1.490	1.508	1.526	1.544	1.562	1.580	1.598	1.616	1.634	1.651	1.668	1.685	1.702	1.719	1.736	1.753
40	1.475	1.493	1.511	1.529	1.547	1.565	1.583	1.601	1.619	1.636	1.653	1.670	1.687	1.704	1.721	1.738
41	1.460	1.478	1.486	1.514	1.532	1.550	1.568	1.586	1.604	1.621	1.638	1.655	1.672	1.689	1.706	1.723
42	1.445	1.463	1.471	1.497	1.518	1.536	1.554	1.572	1.590	1.607	1.624	1.641	1.658	1.675	1.692	1.709
43	1.430	1.448	1.456	1.482	1.499	1.516	1.534	1.552	1.570	1.587	1.604	1.621	1.638	1.655	1.672	1.689
44	1.415	1.433	1.441	1.467	1.485	1.502	1.520	1.538	1.556	1.573	1.590	1.607	1.624	1.641	1.658	1.675
45	1.400	1.418	1.426	1.451	1.469	1.487	1.505	1.523	1.540	1.557	1.574	1.591	1.608	1.625	1.642	1.659
46	1.385	1.393	1.401	1.426	1.444	1.462	1.480	1.498	1.515	1.532	1.549	1.566	1.583	1.599	1.616	1.633
47	1.370	1.378	1.386	1.404	1.422	1.440	1.458	1.476	1.493	1.510	1.527	1.544	1.561	1.578	1.595	1.612
48	1.355	1.363	1.371	1.389	1.407	1.425	1.443	1.461	1.478	1.495	1.512	1.529	1.546	1.563	1.580	1.597
49	1.340	1.348	1.356	1.374	1.392	1.410	1.428	1.446	1.463	1.480	1.497	1.514	1.531	1.548	1.565	1.582
50	1.325	1.333	1.341	1.359	1.377	1.395	1.413	1.431	1.448	1.465	1.482	1.499	1.516	1.533	1.550	1.567
51	1.310	1.318	1.326	1.344	1.362	1.380	1.398	1.416	1.433	1.450	1.467	1.484	1.501	1.518	1.535	1.552
52	1.295	1.303	1.311	1.329	1.347	1.365	1.383	1.401	1.418	1.435	1.452	1.469	1.486	1.503	1.520	1.537
53	1.280	1.288	1.296	1.314	1.332	1.350	1.368	1.386	1.403	1.420	1.437	1.454	1.471	1.488	1.505	1.522
54	1.265	1.273	1.281	1.299	1.317	1.335	1.353	1.371	1.388	1.405	1.422	1.439	1.456	1.473	1.490	1.507
55	1.250	1.258	1.266	1.284	1.302	1.320	1.338	1.356	1.373	1.390	1.407	1.424	1.441	1.458	1.475	1.492
56	1.235	1.243	1.251	1.269	1.287	1.305	1.323	1.341	1.358	1.375	1.392	1.409	1.426	1.443	1.460	1.477
57	1.220	1.228	1.236	1.254	1.272	1.290	1.308	1.326	1.343	1.360	1.377	1.394	1.411	1.428	1.445	1.462
58	1.205	1.213	1.221	1.239	1.257	1.275	1.293	1.311	1.328	1.345	1.362	1.379	1.396	1.413	1.430	1.447
59	1.190	1.198	1.206	1.224	1.242	1.260	1.278	1.296	1.313	1.330	1.347	1.364	1.381	1.398	1.415	1.432
60	1.175	1.183	1.191	1.209	1.227	1.245	1.263	1.281	1.298	1.315	1.332	1.349	1.366	1.383	1.400	1.417
61	1.160	1.168	1.176	1.194	1.212	1.230	1.248	1.266	1.283	1.300	1.317	1.334	1.351	1.368	1.385	1.402
62	1.145	1.153	1.161	1.179	1.197	1.215	1.233	1.251	1.268	1.285	1.302	1.319	1.336	1.353	1.370	1.387
63	1.130	1.138	1.146	1.164	1.182	1.200	1.218	1.236	1.253	1.270	1.287	1.304	1.321	1.338	1.355	1.372
64	1.115	1.123	1.131	1.149	1.167	1.185	1.203	1.221	1.238	1.255	1.272	1.289	1.306	1.323	1.340	1.357
65	1.100	1.108	1.116	1.134	1.152	1.170	1.188	1.206	1.223	1.240	1.257	1.274	1.291	1.308	1.325	1.342
66	1.085	1.093	1.101	1.119	1.137	1.155	1.173	1.191	1.208	1.225	1.242	1.259	1.276	1.293	1.310	1.327
67	1.070	1.078	1.086	1.094	1.112	1.130	1.148	1.166	1.183	1.200	1.217	1.234	1.251	1.268	1.285	1.302
68	1.055	1.063	1.071	1.079	1.097	1.115	1.133	1.151	1.168	1.185	1.202	1.219	1.236	1.253	1.270	1.287
69	1.040	1.048	1.056	1.064	1.082	1.100	1.118	1.136	1.153	1.170	1.187	1.204	1.221	1.238	1.255	1.272
70	1.025	1.033	1.041	1.049	1.067	1.085	1.103	1.121	1.138	1.155	1.172	1.189	1.206	1.223	1.240	1.257
71	1.010	1.018	1.026	1.034	1.052	1.070	1.088	1.106	1.123	1.140	1.157	1.174	1.191	1.208	1.225	1.242
72	995	1.003	1.011	1.019	1.037	1.055	1.073	1.091	1.108	1.125	1.142	1.159	1.176	1.193	1.210	1.227
73	980	988	996	1.004	1.022	1.040	1.058	1.076	1.093	1.110	1.127	1.144	1.161	1.178	1.195	1.212
74	965	973	981	989	1.007	1.025	1.043	1.061	1.078	1.095	1.112	1.129	1.146	1.163	1.180	1.197
75	950	958	966	974	982	990	1.008	1.026	1.043	1.060	1.077	1.094	1.111	1.128	1.145	1.162
76	935	943	951	959	967	975	983	991	1.008	1.025	1.042	1.059	1.076	1.093	1.110	1.127
77	920	928	936	944	952	960	968	976	984	991	1.008	1.025	1.042	1.059	1.076	1.093
78	905	913	921	929	937	945	953	961	969	977	984	991	1.008	1.025	1.042	1.059
79	890	898	906	914	922	930	938	946	954	962	970	977	984	991	1.008	1.025
80	875	883	891	899	907	915	923	931	939	947	955	962	970	977	984	991
81	860	868	876	884	892	899	907	915	923	931	939	947	955	962	970	977
82	845	853	861	869	877	885	893	901	909	917	925	933	941	949	957	965
83	830	838	846	854	862	870	878	886	894	902	910	918	926	934	942	950
84	815	823	831	839	847	855	863	871	879	887	895	903	911	919	927	935
85	800	808	816	824	832	840	848	856	864	872	880	888	896	904	912	920
86	785	793	801	809	817	825	833	841	849	857	865	873	881	889	897	905
87	770	778	786	794	802	810	818	826	834	842	850	858	866	874	882	890
88	755	763	771	779	787	795	803	811	819	827	835	843	851	859	867	875
89	740	748	756	764	772	780	788	796	804	812	820	828	836	844	852	860
90	725	733	741	749	757	765	773	781	789	797	805					

TABLE II - X-PANEL PROPERTIES $t_w = 0.631$; $\frac{b_A}{t_w} = 9.3$; $\frac{b_Y}{t_w} = 1.04$; $\frac{b_X}{t_w} = 1.06$; $\frac{b_T}{t_w} = 0.94$; $\frac{b_F}{t_w} = 2.13$; $\frac{b_R}{t_w} = 0.69$; $\frac{r_w}{t_w} = 1$; $\frac{a}{t_w} = 1.6$; $\frac{r}{t_w} = 6.1$

$\frac{b}{t_w}$	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33
$\frac{b}{t_w} = 1$	2.059	2.080	2.115	2.148	2.167	2.192	2.216	2.240	2.263	2.285	2.306	2.327	2.347	2.366	2.385	2.404
	2.051	2.059	2.067	2.113	2.139	2.164	2.185	2.211	2.234	2.256	2.277	2.297	2.316	2.337	2.357	2.375
	2.005	2.053	2.060	2.088	2.112	2.136	2.160	2.184	2.206	2.228	2.247	2.270	2.290	2.309	2.328	2.347
	1.980	2.008	2.035	2.061	2.086	2.110	2.134	2.157	2.180	2.203	2.228	2.253	2.273	2.293	2.310	2.330
	1.956	1.984	2.010	2.034	2.061	2.085	2.109	2.132	2.154	2.176	2.197	2.217	2.238	2.257	2.276	2.294
	1.931	1.961	1.987	2.017	2.034	2.062	2.085	2.108	2.130	2.151	2.172	2.193	2.213	2.233	2.253	2.273
	1.912	1.939	1.965	1.990	2.015	2.039	2.062	2.085	2.107	2.129	2.149	2.169	2.189	2.208	2.227	2.245
	1.893	1.918	1.941	1.964	1.993	2.017	2.040	2.063	2.084	2.106	2.126	2.146	2.166	2.186	2.206	2.226
	1.872	1.898	1.924	1.948	1.973	1.998	2.029	2.041	2.063	2.084	2.105	2.125	2.145	2.165	2.185	2.204
	1.853	1.879	1.904	1.929	1.952	1.976	1.998	2.021	2.042	2.063	2.084	2.104	2.125	2.145	2.165	2.185
$\frac{b}{t_w} = \frac{1}{2}$	1.835	1.861	1.886	1.910	1.927	1.957	1.979	2.001	2.023	2.044	2.064	2.084	2.104	2.124	2.144	2.164
	1.818	1.842	1.868	1.892	1.908	1.928	1.948	1.969	1.990	2.005	2.025	2.045	2.065	2.085	2.105	2.125
	1.792	1.810	1.834	1.858	1.868	1.890	1.910	1.930	1.950	1.970	1.990	2.009	2.029	2.047	2.065	2.082
	1.771	1.791	1.811	1.835	1.848	1.867	1.890	1.911	1.931	1.951	1.971	1.991	2.011	2.031	2.051	2.071
	1.751	1.765	1.784	1.804	1.815	1.835	1.857	1.877	1.897	1.917	1.937	1.957	1.977	1.997	2.017	2.037
	1.731	1.751	1.771	1.791	1.810	1.830	1.851	1.871	1.891	1.911	1.931	1.951	1.971	1.991	2.011	2.031
	1.702	1.725	1.747	1.769	1.791	1.812	1.833	1.853	1.873	1.893	1.913	1.933	1.953	1.973	1.993	2.013
	1.682	1.700	1.722	1.744	1.765	1.785	1.806	1.826	1.846	1.866	1.886	1.906	1.926	1.946	1.966	1.986
	1.662	1.687	1.707	1.727	1.747	1.767	1.787	1.807	1.827	1.847	1.867	1.887	1.907	1.927	1.947	1.967
	1.641	1.666	1.687	1.708	1.728	1.748	1.768	1.788	1.808	1.828	1.848	1.868	1.888	1.908	1.928	1.948
$\frac{b}{t_w} = \frac{1}{3}$	1.621	1.642	1.662	1.682	1.702	1.722	1.742	1.762	1.782	1.802	1.822	1.842	1.862	1.882	1.902	1.922
	1.596	1.617	1.637	1.657	1.677	1.697	1.717	1.737	1.757	1.777	1.797	1.817	1.837	1.857	1.877	1.897
	1.578	1.599	1.619	1.638	1.657	1.676	1.695	1.715	1.735	1.755	1.775	1.795	1.815	1.835	1.855	1.875
	1.562	1.582	1.601	1.620	1.639	1.657	1.676	1.694	1.713	1.732	1.752	1.772	1.792	1.812	1.832	1.852
	1.546	1.566	1.585	1.604	1.622	1.640	1.658	1.676	1.695	1.714	1.733	1.753	1.773	1.793	1.813	1.833
	1.532	1.551	1.569	1.588	1.606	1.624	1.641	1.659	1.678	1.697	1.716	1.735	1.755	1.775	1.795	1.815
	1.511	1.529	1.548	1.566	1.585	1.603	1.622	1.641	1.660	1.679	1.698	1.717	1.737	1.757	1.777	1.797
	1.492	1.510	1.528	1.547	1.566	1.585	1.604	1.623	1.642	1.661	1.680	1.699	1.719	1.739	1.759	1.779
	1.474	1.491	1.509	1.528	1.547	1.566	1.585	1.604	1.623	1.642	1.661	1.680	1.699	1.719	1.739	1.759
	1.453	1.472	1.491	1.509	1.528	1.547	1.566	1.585	1.604	1.623	1.642	1.661	1.680	1.699	1.719	1.739
$\frac{b}{t_w} = \frac{1}{4}$	1.432	1.451	1.470	1.489	1.507	1.526	1.545	1.564	1.583	1.602	1.621	1.640	1.659	1.678	1.698	1.718
	1.412	1.431	1.450	1.469	1.487	1.506	1.525	1.544	1.563	1.582	1.601	1.620	1.639	1.658	1.678	1.698
	1.391	1.410	1.429	1.448	1.467	1.486	1.505	1.524	1.543	1.562	1.581	1.600	1.619	1.638	1.658	1.678
	1.371	1.390	1.409	1.428	1.447	1.466	1.485	1.504	1.523	1.542	1.561	1.580	1.599	1.618	1.638	1.658
	1.350	1.369	1.388	1.407	1.426	1.445	1.464	1.483	1.502	1.521	1.540	1.559	1.578	1.597	1.617	1.637
	1.329	1.348	1.367	1.386	1.405	1.424	1.443	1.462	1.481	1.500	1.519	1.538	1.557	1.576	1.596	1.616
	1.309	1.328	1.347	1.366	1.385	1.404	1.423	1.442	1.461	1.480	1.499	1.518	1.537	1.556	1.576	1.596
	1.289	1.308	1.327	1.346	1.365	1.384	1.403	1.422	1.441	1.460	1.479	1.498	1.517	1.536	1.555	1.575
	1.269	1.288	1.307	1.326	1.345	1.364	1.383	1.402	1.421	1.440	1.459	1.478	1.497	1.516	1.535	1.555
	1.249	1.268	1.287	1.306	1.325	1.344	1.363	1.382	1.401	1.420	1.439	1.458	1.477	1.496	1.515	1.535
$\frac{b}{t_w} = \frac{1}{5}$	1.229	1.248	1.267	1.286	1.305	1.324	1.343	1.362	1.381	1.400	1.419	1.438	1.457	1.476	1.495	1.515
	1.209	1.228	1.247	1.266	1.285	1.304	1.323	1.342	1.361	1.380	1.399	1.418	1.437	1.456	1.475	1.495
	1.189	1.208	1.227	1.246	1.265	1.284	1.303	1.322	1.341	1.360	1.379	1.398	1.417	1.436	1.455	1.475
	1.169	1.188	1.207	1.226	1.245	1.264	1.283	1.302	1.321	1.340	1.359	1.378	1.397	1.416	1.435	1.455
	1.149	1.168	1.187	1.206	1.225	1.244	1.263	1.282	1.301	1.320	1.339	1.358	1.377	1.396	1.415	1.435
	1.129	1.148	1.167	1.186	1.205	1.224	1.243	1.262	1.281	1.300	1.319	1.338	1.357	1.376	1.395	1.415
	1.109	1.128	1.147	1.166	1.185	1.204	1.223	1.242	1.261	1.280	1.299	1.318	1.337	1.356	1.375	1.395
	1.089	1.108	1.127	1.146	1.165	1.184	1.203	1.222	1.241	1.260	1.279	1.298	1.317	1.336	1.355	1.375
	1.069	1.088	1.107	1.126	1.145	1.164	1.183	1.202	1.221	1.240	1.259	1.278	1.297	1.316	1.335	1.355
	1.049	1.068	1.087	1.106	1.125	1.144	1.163	1.182	1.201	1.220	1.239	1.258	1.277	1.296	1.315	1.335
$\frac{b}{t_w} = \frac{1}{6}$	1.029	1.048	1.067	1.086	1.105	1.124	1.143	1.162	1.181	1.200	1.219	1.238	1.257	1.276	1.295	1.315
	1.009	1.028	1.047	1.066	1.085	1.104	1.123	1.142	1.161	1.180	1.199	1.218	1.237	1.256	1.275	1.295
	989	1.008	1.027	1.046	1.065	1.084	1.103	1.122	1.141	1.160	1.179	1.198	1.217	1.236	1.255	1.275
	969	988	1.007	1.026	1.045	1.064	1.083	1.102	1.121	1.140	1.159	1.178	1.197	1.216	1.235	1.255
	949	968	987	1.006	1.025	1.044	1.063	1.082	1.101	1.120	1.139	1.158	1.177	1.196	1.215	1.235
	929	948	967	986	1.005	1.024	1.043	1.062	1.081	1.100	1.119	1.138	1.157	1.176	1.195	1.215
	909	928	947	966	985	1.004	1.023	1.042	1.061	1.080	1.099	1.118	1.137	1.156	1.175	1.195
	889	908	927	946	965	984	1.003	1.022	1.041	1.060	1.079	1.098	1.117	1.136	1.155	1.175
	869	888	907	926	945	964	983	1.002	1.021	1.040	1.059	1.078	1.097	1.116	1.135	1.155
	850	869	888	907	926	945	964	983	1.002	1.021	1.040	1.059	1.078	1.097	1.116	1.135
$\frac{b}{t_w} = \frac{1}{7}$	830	849	868	887	906	925	944	963	982	1.001	1.020	1.039	1.058	1.077	1.096	1.115
	810	829	848	867	886	905	924	943	962	981	1.000	1.019	1.038	1.057	1.076	1.095
	790	809	828	847	866	885	904	923	942	961	980	1.000	1.019	1.038	1.057	1.076
	770	789	808	827	846	865	884	903	922	941	960	979	1.000	1.019</td		

TABLE II.-Y-PANEL PROPERTIES $\frac{t_w}{t_0} = 0.79$; $\frac{b_A}{t_y} = 9.3$; $\frac{b_y}{t_y} = 1.04$; $\frac{t_L}{t_w} = 1.06$; $\frac{b_L}{b_y} = 0.94$; $\frac{t_p}{t_y} = 2.13$; $\frac{b_p}{t_y} = 0.69$; $\frac{r}{r_k} = 1$; $\frac{d}{t_b} = 2.3$; $\frac{p}{t_b} = 7.7$

$\frac{t_w}{t_0}$	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33
23	2.517	2.520	2.522	2.524	2.525	2.527	2.528	2.529	2.530	2.531	2.532	2.533	2.533	2.533	2.533	2.533
24	2.495	2.498	2.500	2.502	2.503	2.505	2.506	2.507	2.508	2.509	2.510	2.510	2.510	2.510	2.510	2.510
25	2.473	2.475	2.477	2.479	2.480	2.482	2.483	2.484	2.485	2.486	2.487	2.487	2.487	2.487	2.487	2.487
26	2.451	2.453	2.455	2.457	2.458	2.459	2.460	2.461	2.462	2.463	2.464	2.464	2.464	2.464	2.464	2.464
27	2.429	2.431	2.433	2.435	2.436	2.437	2.438	2.439	2.440	2.441	2.442	2.442	2.442	2.442	2.442	2.442
28	2.407	2.409	2.411	2.413	2.414	2.415	2.416	2.417	2.418	2.419	2.420	2.420	2.420	2.420	2.420	2.420
29	2.385	2.387	2.389	2.391	2.392	2.393	2.394	2.395	2.396	2.397	2.398	2.398	2.398	2.398	2.398	2.398
30	2.363	2.365	2.367	2.369	2.370	2.371	2.372	2.373	2.374	2.375	2.376	2.376	2.376	2.376	2.376	2.376
31	2.341	2.343	2.345	2.347	2.348	2.349	2.350	2.351	2.352	2.353	2.354	2.354	2.354	2.354	2.354	2.354
32	2.319	2.321	2.323	2.325	2.326	2.327	2.328	2.329	2.330	2.331	2.332	2.332	2.332	2.332	2.332	2.332
33	2.296	2.298	2.300	2.302	2.303	2.304	2.305	2.306	2.307	2.308	2.309	2.309	2.309	2.309	2.309	2.309
34	2.274	2.276	2.278	2.280	2.281	2.282	2.283	2.284	2.285	2.286	2.287	2.287	2.287	2.287	2.287	2.287
35	2.252	2.254	2.256	2.258	2.259	2.260	2.261	2.262	2.263	2.264	2.265	2.265	2.265	2.265	2.265	2.265
36	2.230	2.232	2.234	2.236	2.237	2.238	2.239	2.240	2.241	2.242	2.243	2.243	2.243	2.243	2.243	2.243
37	2.208	2.210	2.212	2.214	2.215	2.216	2.217	2.218	2.219	2.220	2.221	2.221	2.221	2.221	2.221	2.221
38	2.186	2.188	2.190	2.192	2.193	2.194	2.195	2.196	2.197	2.198	2.199	2.199	2.199	2.199	2.199	2.199
39	2.164	2.166	2.168	2.170	2.171	2.172	2.173	2.174	2.175	2.176	2.177	2.177	2.177	2.177	2.177	2.177
40	2.142	2.143	2.144	2.145	2.146	2.147	2.148	2.149	2.150	2.151	2.152	2.152	2.152	2.152	2.152	2.152
41	2.120	2.121	2.122	2.123	2.124	2.125	2.126	2.127	2.128	2.129	2.130	2.130	2.130	2.130	2.130	2.130
42	2.098	2.099	2.100	2.101	2.102	2.103	2.104	2.105	2.106	2.107	2.108	2.108	2.108	2.108	2.108	2.108
43	2.076	2.077	2.078	2.079	2.080	2.081	2.082	2.083	2.084	2.085	2.086	2.086	2.086	2.086	2.086	2.086
44	2.054	2.055	2.056	2.057	2.058	2.059	2.060	2.061	2.062	2.063	2.064	2.064	2.064	2.064	2.064	2.064
45	2.032	2.033	2.034	2.035	2.036	2.037	2.038	2.039	2.040	2.041	2.042	2.042	2.042	2.042	2.042	2.042
46	2.010	2.011	2.012	2.013	2.014	2.015	2.016	2.017	2.018	2.019	2.020	2.020	2.020	2.020	2.020	2.020
47	1.988	1.989	1.990	1.991	1.992	1.993	1.994	1.995	1.996	1.997	1.998	1.998	1.998	1.998	1.998	1.998
48	1.966	1.967	1.968	1.969	1.970	1.971	1.972	1.973	1.974	1.975	1.976	1.976	1.976	1.976	1.976	1.976
49	1.944	1.945	1.946	1.947	1.948	1.949	1.950	1.951	1.952	1.953	1.954	1.954	1.954	1.954	1.954	1.954
50	1.922	1.923	1.924	1.925	1.926	1.927	1.928	1.929	1.930	1.931	1.932	1.932	1.932	1.932	1.932	1.932
51	1.900	1.901	1.902	1.903	1.904	1.905	1.906	1.907	1.908	1.909	1.910	1.910	1.910	1.910	1.910	1.910
52	1.878	1.879	1.880	1.881	1.882	1.883	1.884	1.885	1.886	1.887	1.888	1.888	1.888	1.888	1.888	1.888
53	1.856	1.857	1.858	1.859	1.860	1.861	1.862	1.863	1.864	1.865	1.866	1.866	1.866	1.866	1.866	1.866
54	1.834	1.835	1.836	1.837	1.838	1.839	1.840	1.841	1.842	1.843	1.844	1.844	1.844	1.844	1.844	1.844
55	1.812	1.813	1.814	1.815	1.816	1.817	1.818	1.819	1.820	1.821	1.822	1.822	1.822	1.822	1.822	1.822
56	1.790	1.791	1.792	1.793	1.794	1.795	1.796	1.797	1.798	1.799	1.800	1.800	1.800	1.800	1.800	1.800
57	1.768	1.769	1.770	1.771	1.772	1.773	1.774	1.775	1.776	1.777	1.778	1.778	1.778	1.778	1.778	1.778
58	1.746	1.747	1.748	1.749	1.750	1.751	1.752	1.753	1.754	1.755	1.756	1.756	1.756	1.756	1.756	1.756
59	1.724	1.725	1.726	1.727	1.728	1.729	1.730	1.731	1.732	1.733	1.734	1.734	1.734	1.734	1.734	1.734
60	1.702	1.703	1.704	1.705	1.706	1.707	1.708	1.709	1.710	1.711	1.712	1.712	1.712	1.712	1.712	1.712
61	1.680	1.681	1.682	1.683	1.684	1.685	1.686	1.687	1.688	1.689	1.690	1.690	1.690	1.690	1.690	1.690
62	1.658	1.659	1.660	1.661	1.662	1.663	1.664	1.665	1.666	1.667	1.668	1.668	1.668	1.668	1.668	1.668
63	1.636	1.637	1.638	1.639	1.640	1.641	1.642	1.643	1.644	1.645	1.646	1.646	1.646	1.646	1.646	1.646
64	1.614	1.615	1.616	1.617	1.618	1.619	1.620	1.621	1.622	1.623	1.624	1.624	1.624	1.624	1.624	1.624
65	1.592	1.593	1.594	1.595	1.596	1.597	1.598	1.599	1.600	1.601	1.602	1.602	1.602	1.602	1.602	1.602
66	1.570	1.571	1.572	1.573	1.574	1.575	1.576	1.577	1.578	1.579	1.580	1.580	1.580	1.580	1.580	1.580
67	1.548	1.549	1.550	1.551	1.552	1.553	1.554	1.555	1.556	1.557	1.558	1.558	1.558	1.558	1.558	1.558
68	1.526	1.527	1.528	1.529	1.530	1.531	1.532	1.533	1.534	1.535	1.536	1.536	1.536	1.536	1.536	1.536
69	1.504	1.505	1.506	1.507	1.508	1.509	1.510	1.511	1.512	1.513	1.514	1.514	1.514	1.514	1.514	1.514
70	1.482	1.483	1.484	1.485	1.486	1.487	1.488	1.489	1.490	1.491	1.492	1.492	1.492	1.492	1.492	1.492
71	1.460	1.461	1.462	1.463	1.464	1.465	1.466	1.467	1.468	1.469	1.470	1.470	1.470	1.470	1.470	1.470
72	1.438	1.439	1.440	1.441	1.442	1.443	1.444	1.445	1.446	1.447	1.448	1.448	1.448	1.448	1.448	1.448
73	1.416	1.417	1.418	1.419	1.420	1.421	1.422	1.423	1.424	1.425	1.426	1.426	1.426	1.426	1.426	1.426
74	1.394	1.395	1.396	1.397	1.398	1.399	1.400	1.401	1.402	1.403	1.404	1.404	1.404	1.404	1.404	1.404
75	1.372	1.373	1.374	1.375	1.376	1.377	1.378	1.379	1.380	1.381	1.382	1.382	1.382	1.382	1.382	1.382
76	1.350	1.351	1.352	1.353	1.354	1.355	1.356	1.357	1.358	1.359	1.360	1.360	1.360	1.360	1.360	1.360
77	1.328	1.329	1.330	1.331	1.332	1.333	1.334	1.335	1.336	1.337	1.338	1.338	1.338	1.338	1.338	1.338
78	1.306	1.307	1.308	1.309	1.310	1.311	1.312	1.313	1.314	1.315	1.316	1.316	1.316	1.316	1.316	1.316
79	1.284	1.285	1.286	1.287	1.288	1.289	1.290	1.291	1.292	1.293	1.294	1.294	1.294	1.294	1.294	1.294
80	1.262	1.263	1.264	1.265	1.266	1.267	1.268	1.269	1.270	1.271	1.272	1.272	1.272	1.272	1.272	1.272
81	1.240	1.241	1.242	1.243	1.244	1.245	1.246	1.247	1.248	1.249	1.250	1.250	1.250	1.250	1.250	1.250
82	1.218	1.219	1.220	1.221	1.222	1.223	1.224	1.225	1.226	1.227	1.228	1.228	1.228	1.228	1.228	1.228
83	1.196	1.197	1.198	1.199	1.200	1.201	1.202	1.203	1.204	1.205	1.206	1.206	1.206	1.206	1.206	1.206
84	1.174	1.175	1.176	1.177	1.178	1.179	1.180	1.181	1.182	1.183	1.184	1.184	1.184	1.184	1.184	1.184
85	1.152	1.153	1.154	1.155	1.156	1.157	1.158	1.159	1.160	1.161	1.162	1.162	1.162	1.162	1.162	1.162
86	1.130	1.131	1.132	1.133	1.134	1.135	1.136	1.137	1.138	1.139	1.140	1.140	1.140	1.140	1.140	1.140
87	1.108	1.109	1.110	1.111	1.112	1.113	1.114	1.115	1.116	1.117	1.118	1.118	1.118	1.118	1.118	1.118

TABLE 15.- X-PANEL PROPERTIES [$t_w = 1.00$; $b_x = 9.3$; $b_y = 1.04$; $\frac{t_w}{b_x} = 1.06$; $\frac{t_w}{b_y} = 0.94$; $\frac{t_w}{t_v} = 2.13$; $\frac{b_x}{b_y} = 0.69$; $\frac{t_w}{t_b} = 1$; $\frac{b_x}{t_b} = 2.4$; $\frac{t_w}{t_s} = 7.6$]

$\frac{t_w}{b_x}$	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33
22	3.177	2.223	2.270	2.317	2.355	2.395	2.433	2.470	2.505	2.539	2.572	2.604	2.634	2.663	2.692	2.719
24	3.062	2.115	2.223	2.262	2.304	2.342	2.382	2.420	2.458	2.495	2.527	2.559	2.590	2.624	2.656	2.676
26	2.943	2.069	2.169	2.219	2.260	2.300	2.337	2.380	2.416	2.450	2.483	2.517	2.555	2.586	2.614	2.633
27	2.908	2.049	2.159	2.209	2.250	2.291	2.321	2.362	2.393	2.427	2.461	2.495	2.525	2.554	2.582	2.602
28	2.863	2.013	2.126	2.176	2.216	2.256	2.287	2.327	2.357	2.391	2.421	2.451	2.481	2.511	2.541	2.561
29	2.825	1.973	2.076	2.126	2.166	2.206	2.236	2.267	2.297	2.326	2.356	2.386	2.416	2.446	2.477	2.507
30	2.804	1.953	2.050	2.093	2.133	2.173	2.203	2.233	2.263	2.292	2.321	2.351	2.380	2.411	2.441	2.471
31	2.785	1.930	2.030	2.070	2.108	2.148	2.182	2.212	2.242	2.271	2.301	2.331	2.361	2.391	2.421	2.451
32	2.765	1.895	2.010	2.049	2.089	2.128	2.162	2.191	2.221	2.251	2.281	2.311	2.341	2.371	2.401	2.431
33	2.756	1.875	1.987	2.027	2.062	2.092	2.122	2.152	2.182	2.212	2.242	2.272	2.302	2.332	2.362	2.392
34	2.755	1.865	1.975	2.015	2.055	2.095	2.125	2.155	2.185	2.215	2.245	2.275	2.305	2.335	2.365	2.395
35	2.755	1.855	1.965	2.005	2.045	2.085	2.115	2.145	2.175	2.205	2.235	2.265	2.295	2.325	2.355	2.385
36	2.695	2.741	2.785	2.827	2.868	2.908	2.946	2.983	3.019	3.054	3.084	3.121	3.152	3.184	3.214	3.244
37	2.667	2.712	2.756	2.799	2.839	2.878	2.917	2.956	2.990	3.029	3.059	3.097	3.127	3.154	3.184	3.214
38	2.639	2.684	2.728	2.770	2.810	2.850	2.888	2.927	2.961	2.999	3.030	3.062	3.092	3.122	3.152	3.182
39	2.613	2.597	2.704	2.742	2.782	2.822	2.860	2.898	2.932	2.969	3.001	3.031	3.061	3.091	3.121	3.151
40	2.587	2.604	2.674	2.712	2.752	2.792	2.830	2.868	2.906	2.944	2.974	3.004	3.034	3.064	3.094	3.124
41	2.566	2.582	2.642	2.680	2.718	2.756	2.794	2.832	2.870	2.908	2.938	2.968	3.008	3.038	3.068	3.098
42	2.546	2.562	2.622	2.660	2.698	2.736	2.774	2.812	2.850	2.888	2.918	2.948	2.978	3.008	3.038	3.068
43	2.526	2.542	2.602	2.640	2.678	2.716	2.754	2.792	2.830	2.868	2.908	2.938	2.968	3.008	3.038	3.068
44	2.506	2.522	2.582	2.620	2.658	2.696	2.734	2.772	2.810	2.848	2.878	2.908	2.938	2.968	3.008	3.038
45	2.486	2.496	2.552	2.590	2.628	2.666	2.704	2.742	2.780	2.818	2.848	2.878	2.908	2.938	2.968	3.008
46	2.466	2.476	2.532	2.570	2.608	2.646	2.684	2.722	2.760	2.798	2.828	2.858	2.888	2.918	2.948	2.978
47	2.446	2.456	2.512	2.550	2.588	2.626	2.664	2.702	2.740	2.778	2.808	2.838	2.868	2.898	2.928	2.958
48	2.426	2.436	2.496	2.534	2.572	2.610	2.648	2.686	2.724	2.762	2.792	2.822	2.852	2.882	2.912	2.942
49	2.406	2.416	2.476	2.514	2.552	2.590	2.628	2.666	2.704	2.742	2.772	2.802	2.832	2.862	2.892	2.922
50	2.386	2.396	2.456	2.494	2.532	2.570	2.608	2.646	2.684	2.722	2.752	2.782	2.812	2.842	2.872	2.902
51	2.366	2.376	2.436	2.474	2.512	2.550	2.588	2.626	2.664	2.702	2.732	2.762	2.792	2.822	2.852	2.882
52	2.346	2.356	2.416	2.454	2.492	2.530	2.568	2.606	2.644	2.682	2.712	2.742	2.772	2.802	2.832	2.862
53	2.326	2.336	2.396	2.434	2.472	2.510	2.548	2.586	2.624	2.662	2.692	2.722	2.752	2.782	2.812	2.842
54	2.306	2.316	2.376	2.414	2.452	2.490	2.528	2.566	2.604	2.642	2.672	2.702	2.732	2.762	2.792	2.822
55	2.286	2.296	2.356	2.394	2.432	2.470	2.508	2.546	2.584	2.622	2.652	2.682	2.712	2.742	2.772	2.802
56	2.266	2.276	2.336	2.374	2.412	2.450	2.488	2.526	2.564	2.602	2.632	2.662	2.692	2.722	2.752	2.782
57	2.246	2.256	2.316	2.354	2.392	2.430	2.468	2.506	2.544	2.582	2.612	2.642	2.672	2.702	2.732	2.762
58	2.226	2.236	2.296	2.334	2.372	2.410	2.448	2.486	2.524	2.562	2.592	2.622	2.652	2.682	2.712	2.742
59	2.206	2.216	2.276	2.314	2.352	2.390	2.428	2.466	2.504	2.542	2.572	2.602	2.632	2.662	2.692	2.722
60	2.186	2.196	2.256	2.294	2.332	2.370	2.408	2.446	2.484	2.522	2.552	2.582	2.612	2.642	2.672	2.702
61	2.166	2.176	2.236	2.274	2.312	2.350	2.388	2.426	2.464	2.502	2.532	2.562	2.592	2.622	2.652	2.682
62	2.146	2.156	2.216	2.254	2.292	2.330	2.368	2.406	2.444	2.482	2.512	2.542	2.572	2.602	2.632	2.662
63	2.126	2.136	2.196	2.234	2.272	2.310	2.348	2.386	2.424	2.462	2.492	2.522	2.552	2.582	2.612	2.642
64	2.106	2.116	2.176	2.214	2.252	2.290	2.328	2.366	2.404	2.442	2.472	2.502	2.532	2.562	2.592	2.622
65	2.086	2.096	2.156	2.194	2.232	2.270	2.308	2.346	2.384	2.422	2.452	2.482	2.512	2.542	2.572	2.602
66	2.066	2.076	2.136	2.174	2.212	2.250	2.288	2.326	2.364	2.402	2.432	2.462	2.492	2.522	2.552	2.582
67	2.046	2.056	2.116	2.154	2.192	2.230	2.268	2.306	2.344	2.382	2.412	2.442	2.472	2.502	2.532	2.562
68	2.026	2.036	2.096	2.134	2.172	2.210	2.248	2.286	2.324	2.362	2.392	2.422	2.452	2.482	2.512	2.542
69	2.006	2.016	2.076	2.114	2.152	2.190	2.228	2.266	2.304	2.342	2.372	2.402	2.432	2.462	2.492	2.522
70	1.986	1.996	2.056	2.094	2.132	2.170	2.208	2.246	2.284	2.322	2.352	2.382	2.412	2.442	2.472	2.502
71	1.966	1.976	2.036	2.074	2.112	2.150	2.188	2.226	2.264	2.302	2.332	2.362	2.392	2.422	2.452	2.482
72	1.946	1.956	2.016	2.054	2.092	2.130	2.168	2.206	2.244	2.282	2.312	2.342	2.372	2.402	2.432	2.462
73	1.926	1.936	1.996	2.034	2.072	2.110	2.148	2.186	2.224	2.262	2.292	2.322	2.352	2.382	2.412	2.442
74	1.906	1.916	1.976	2.014	2.052	2.090	2.128	2.166	2.204	2.242	2.272	2.302	2.332	2.362	2.392	2.422
75	1.886	1.896	1.956	1.994	2.032	2.070	2.108	2.146	2.184	2.222	2.252	2.282	2.312	2.342	2.372	2.402
76	1.866	1.876	1.936	1.974	2.012	2.050	2.088	2.126	2.164	2.202	2.232	2.262	2.292	2.322	2.352	2.382
77	1.846	1.856	1.916	1.954	1.992	2.030	2.068	2.106	2.144	2.182	2.212	2.242	2.272	2.302	2.332	2.362
78	1.826	1.836	1.896	1.934	1.972	2.010	2.048	2.086	2.124	2.162	2.192	2.222	2.252	2.282	2.312	2.342
79	1.806	1.816	1.876	1.914	1.952	1.990	2.028	2.066	2.104	2.142	2.172	2.202	2.232	2.262	2.292	2.322
80	1.786	1.796	1.856	1.894	1.932	1.970	2.008	2.046	2.084	2.122	2.152	2.182	2.212	2.242	2.272	2.302
81	1.766	1.776	1.836	1.874	1.912	1.950	1.988	2.026	2.064	2.102	2.132	2.162	2.192	2.222	2.252	2.282
82	1.746	1.756	1.816	1.854	1.892	1.930	1.968	2.006	2.044	2.082	2.112	2.142	2.172	2.202	2.232	2.262
83	1.726	1.736	1.796	1.834	1.872	1.910	1.948	1.986	2.024	2.062	2.092	2.122	2.152	2.182	2.212	2.242
84	1.706	1.716	1.776	1.814	1.852	1.890	1.928	1.966	2.004	2.042	2.072	2.102	2.132	2.162	2.192	2.222
85	1.686	1.696	1.756	1.794	1.832	1.870	1.908	1.946	1.984	2.022	2.052	2.082	2.112	2.142	2.172	2.202
86	1.666	1.676	1.736	1.774	1.812	1.850	1.888	1.926	1.964	2.002	2.032	2.062	2.092	2.122	2.152	2.182
87	1.646	1.656	1.716	1.754	1.792	1.830	1.868	1.906	1.944	1.982	2.012	2.042	2.072	2.102	2.132	2.162
88	1.626	1.636	1.696	1.734	1.772	1.810										

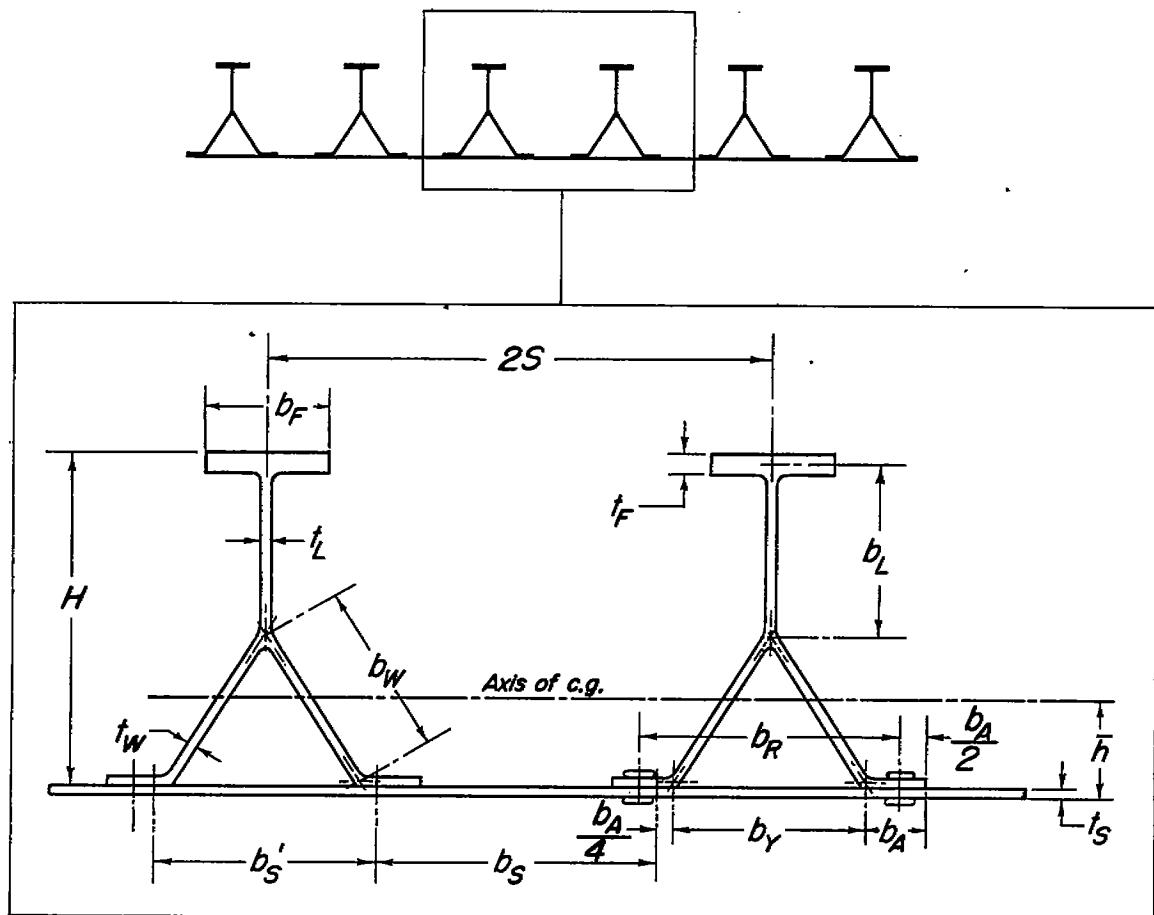
TABLE 14

VALUES AND COMPUTATIONS FOR OBTAINING DESIGN FOR MAXIMUM STRUCTURAL EFFICIENCY
 $[P_1 = 5.0 \text{ kips/in.}; L = 20 \text{ in.}; c = 1; t_S = 0.102 \text{ in.}; \frac{t_W}{t_S} = 0.40]$

Step (1)		Step (3)		Step (4)		Step (5)		Step (8)			Step (9)		Step (10)			Step (12)
$\frac{P_1}{L/\sqrt{c}}$ (ksi)	$\frac{P_1}{t_S}$ (ksi)	b_W t_W	b_S t_S	$\overline{\epsilon}_f$ (ksi)	-	A_1 t_S	$\frac{P_1}{A_1}$ $t_F t_S$	b_S t_S	b_W t_W	$\overline{\epsilon}_f$ (ksi)	A_1 t_S	P_1 (kips/in.)	t_W (in.)	b_S (in.)	b_W (in.)	σ_{cr} (ksi)
0.25	49	18	23 26 30 35 42 50 60 72 84	31.2 31.5 31.6 31.2 29.8 27.8 25.7 22.5 19.2	1.496 1.494 1.407 1.361 1.312 1.270 1.231 1.196 1.171	0.1071 .1090 .1123 .1177 .1278 .1115 .1580 .1860 .2222	36.6 For $t_S = 0.102 \text{ in.}$	25.7	34.2	1.439	5.02	0.040	3.732	1.028	31.5	
		21	23 26 30 35 42 50 60 72 84	33.3 33.6 33.5 32.9 31.5 29.4 26.7 23.4 19.7	1.512 1.497 1.468 1.399 1.345 1.300 1.257 1.220 1.192	.0973 .0995 .1029 .1086 .1179 .1318 .1489 .1750 .2127										
		24	23 26 30 35 42 50 60 72 84	35.1 35.4 35.0 34.1 32.3 30.1 27.2 24.0 20.0	1.585 1.538 1.486 1.434 1.370 1.328 1.283 1.242 1.212	.0899 .0919 .0962 .1022 .1123 .1250 .1332 .1677 .2063										
		27	23 26 30 35 42 50 60 72 84	36.5 36.2 35.8 34.8 32.8 30.2 27.4 24.0 20.2	1.624 1.576 1.522 1.466 1.408 1.356 1.307 1.256 1.231	.0843 .0870 .0915 .0978 .1082 .1221 .1397 .1647 .2010										
		30	23 26 30 35 42 50 60 72 84	37.7 37.2 36.4 35.0 32.6 30.2 27.1 23.7 20.2	1.661 1.612 1.556 1.500 1.437 1.383 1.331 1.285 1.250	.0798 .0834 .0881 .0952 .1067 .1195 .1385 .1642 .1980										
		33	23 26 30 35 42 50 60 72 84	39.0 37.5 36.2 34.7 32.2 29.8 26.7 23.4 20.3	1.696 1.646 1.589 1.520 1.468 1.408 1.354 1.306 1.269	.0756 .0814 .0870 .0911 .1060 .1191 .1381 .1636 .1942										

See appendix A for discussion of steps.

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r = All fillet radii

d = Rivet diameter

p = Rivet pitch

L = Length of panel

Figure 1. - Symbols for panel dimensions.

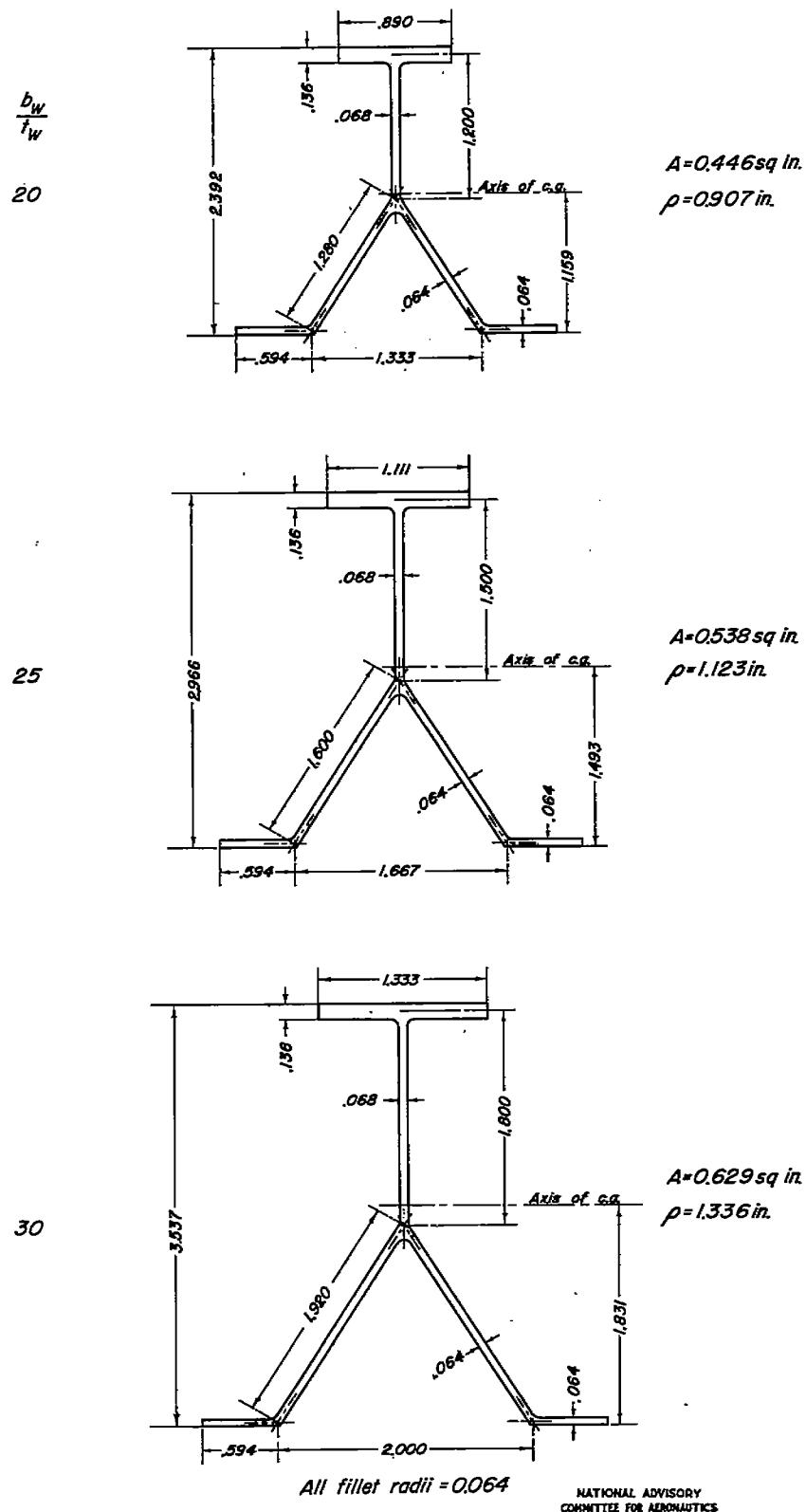
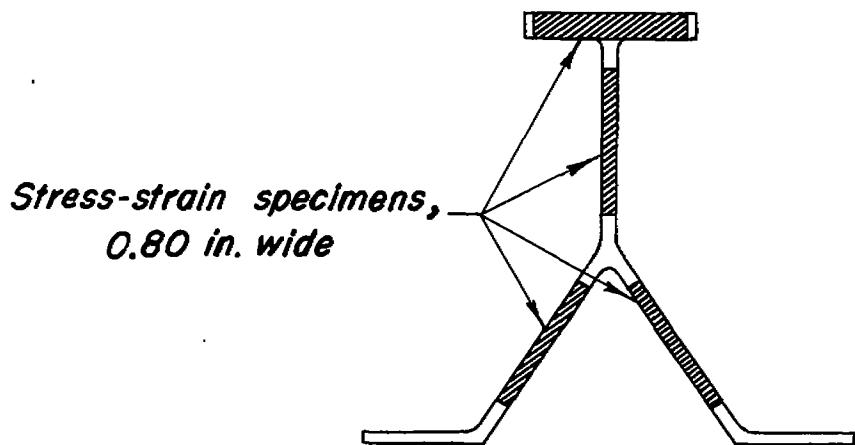


Figure 2.—Dimensions of extrusions used for test specimens.



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Figure 3.- Locations from which stress-strain specimens were cut from Y-section extrusions.

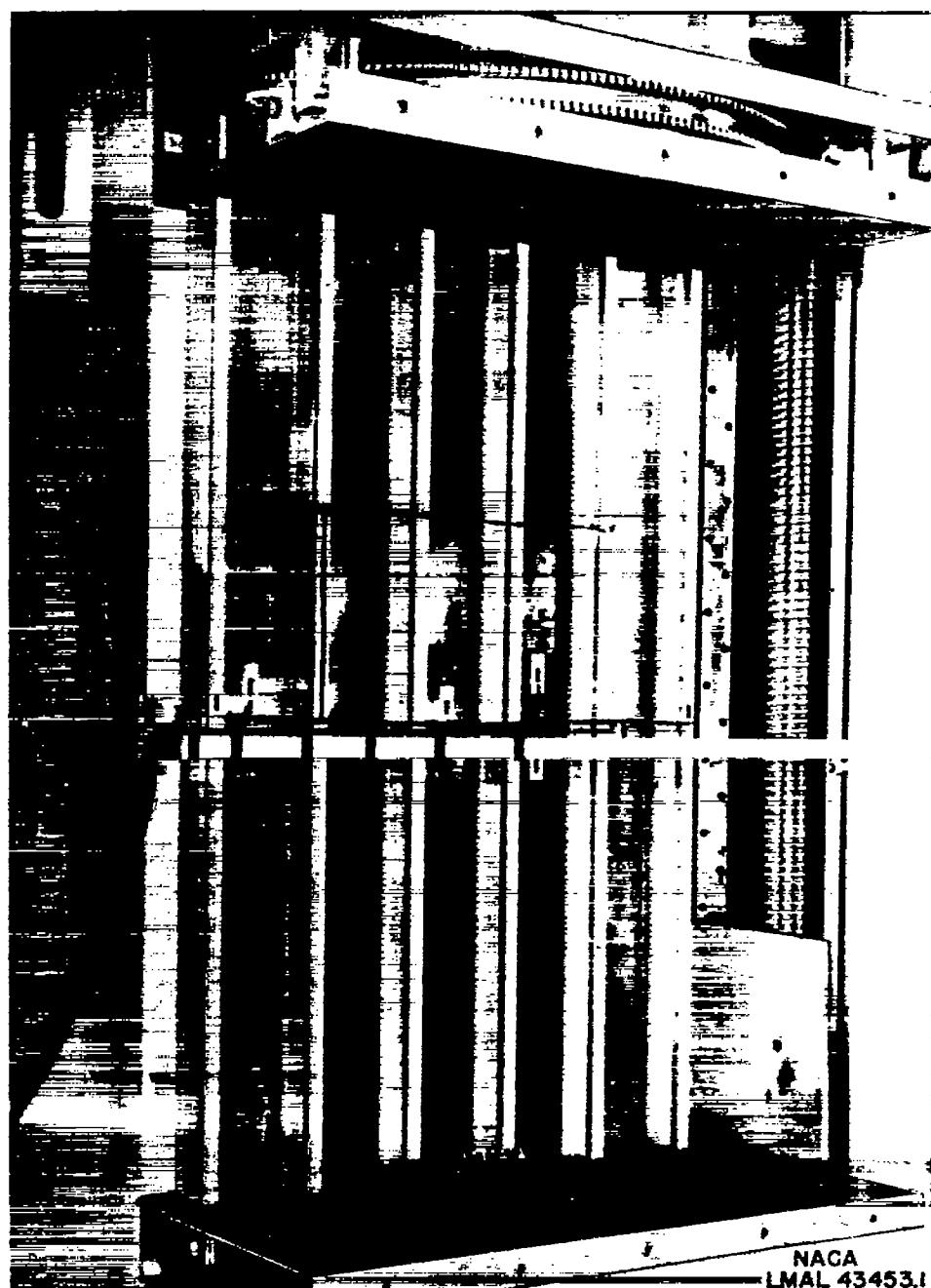


Figure 4.- Test specimen in testing machine.

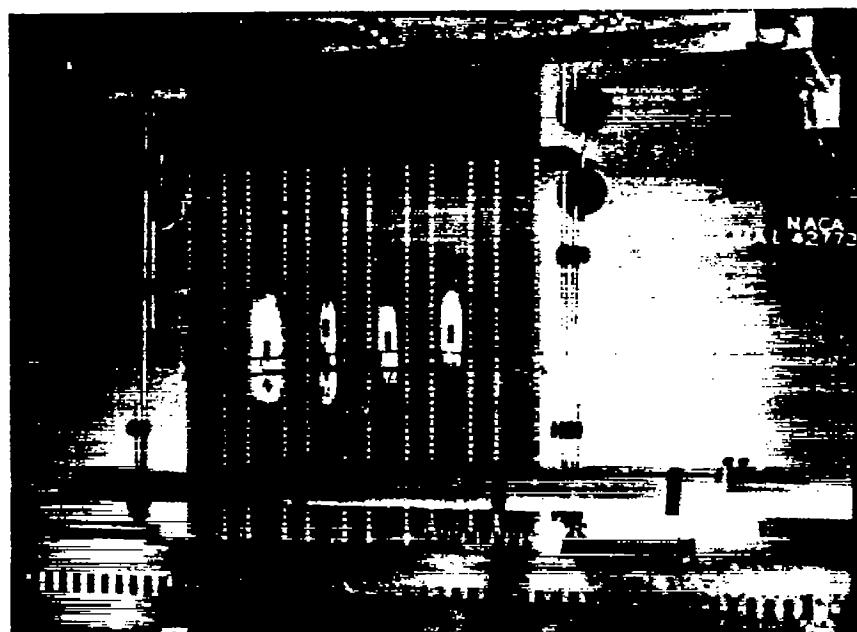
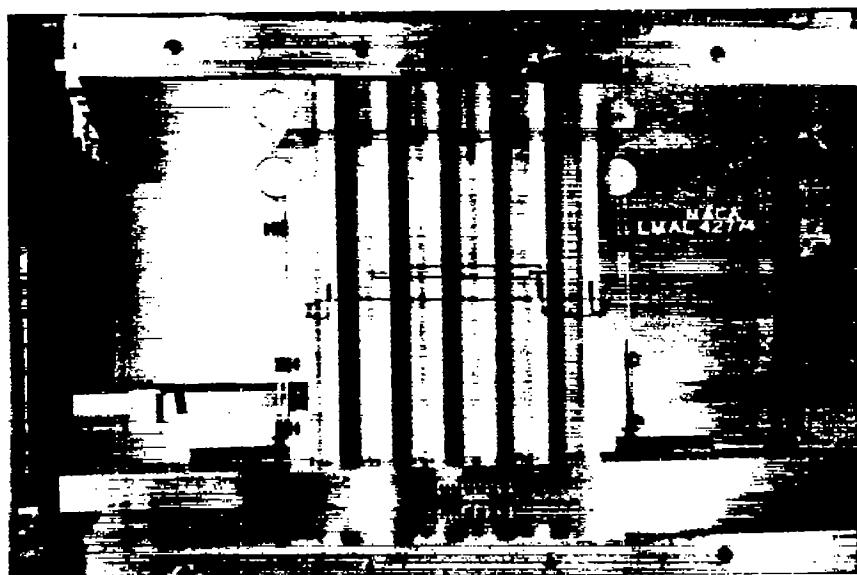


Figure 5.- Test specimen in testing machine with
special bearing block to permit gage wires to be
introduced inside stiffeners.

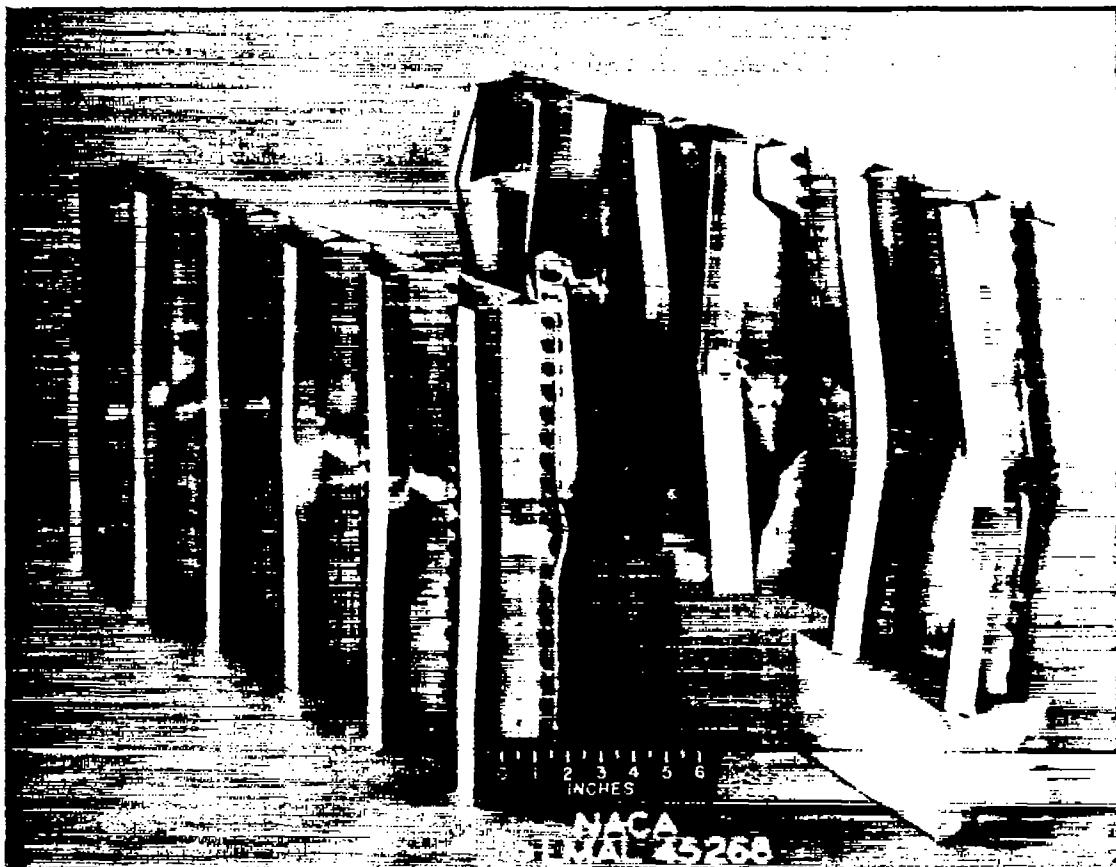


Figure 6.- A 24S-T aluminum-alloy Y-stiffened panel (on the left) and its 75S-T counterpart after failure.

Fig. 7

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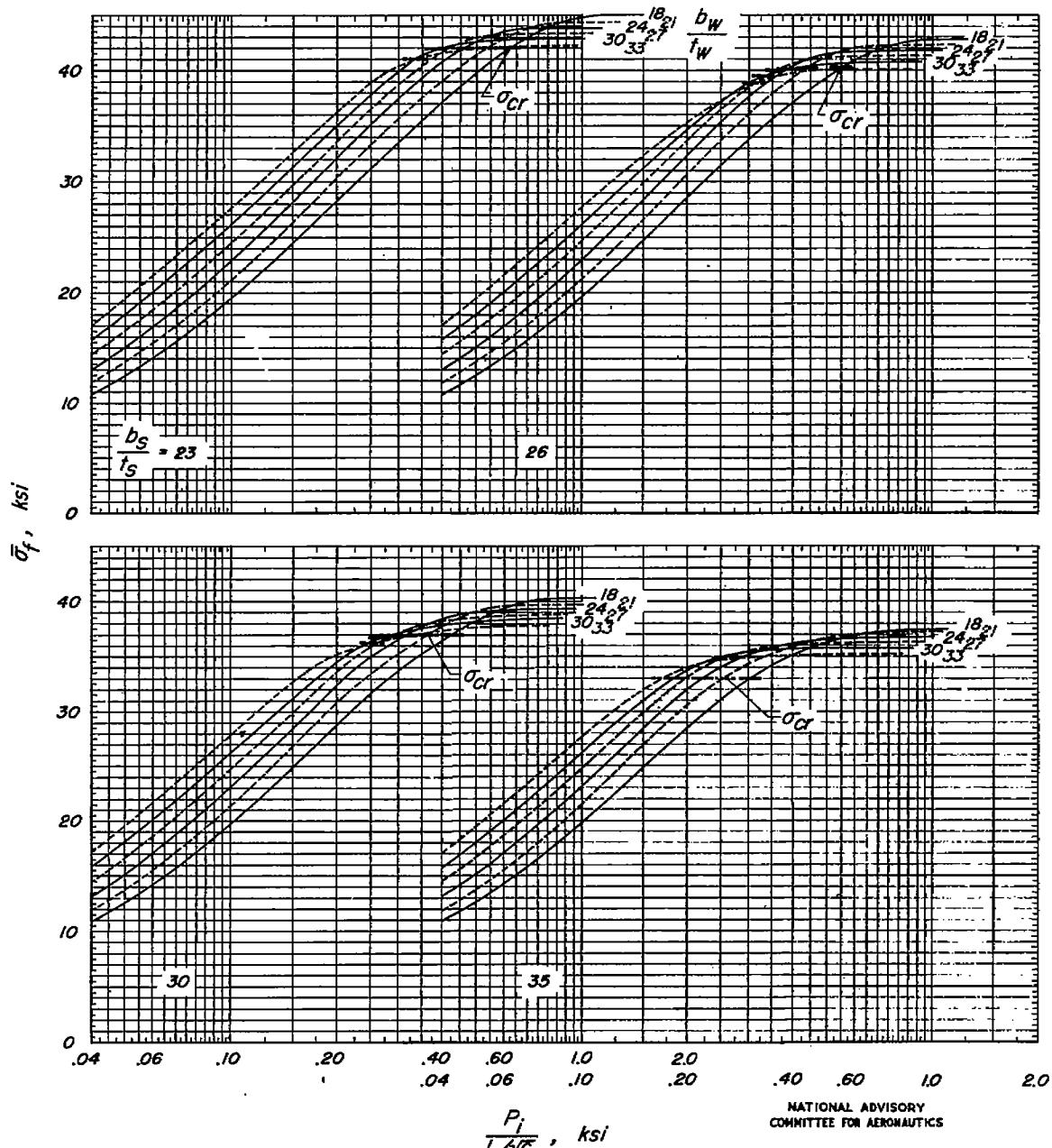


Figure 7.- Design chart for 24S-T Y-panels of the proportions tested. $\frac{t_w}{t_s} = 0.40$.

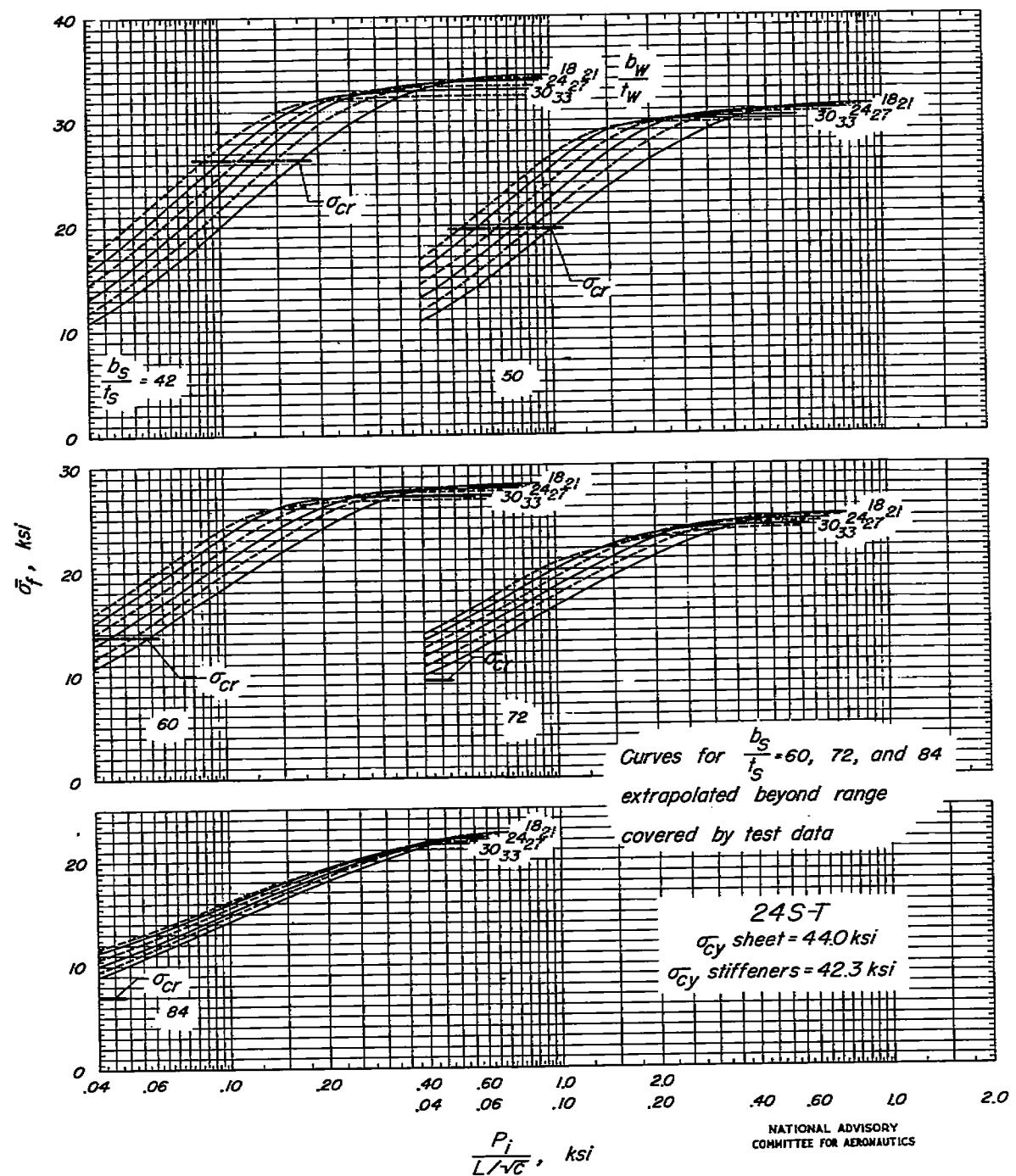


Figure 7. - Concluded.

Fig. 8

NACA TN No. 1389

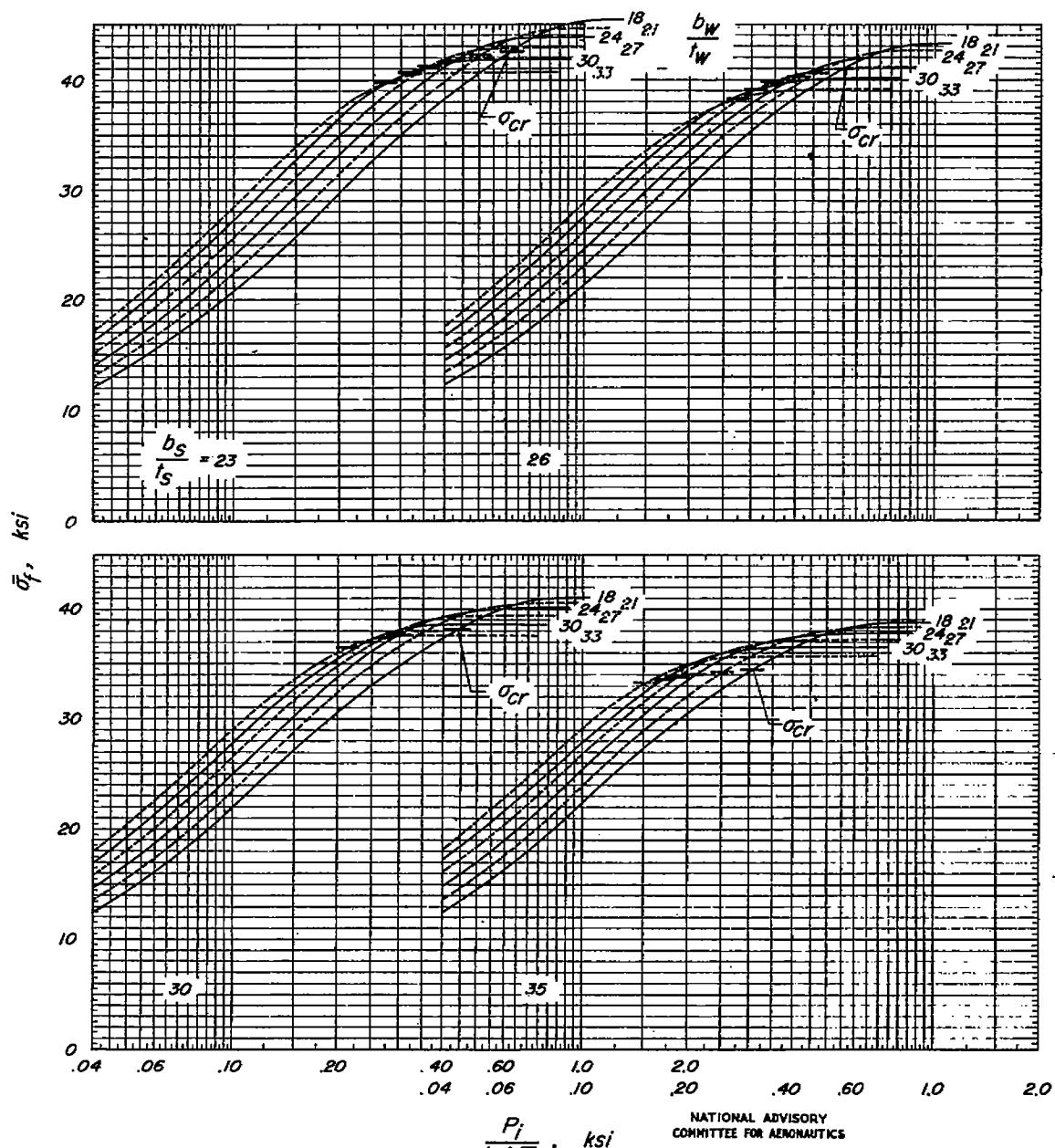


Figure 8. - Design chart for 24S-T Y-panels of the proportions tested. $\frac{t_w}{t_s} = 0.51$.

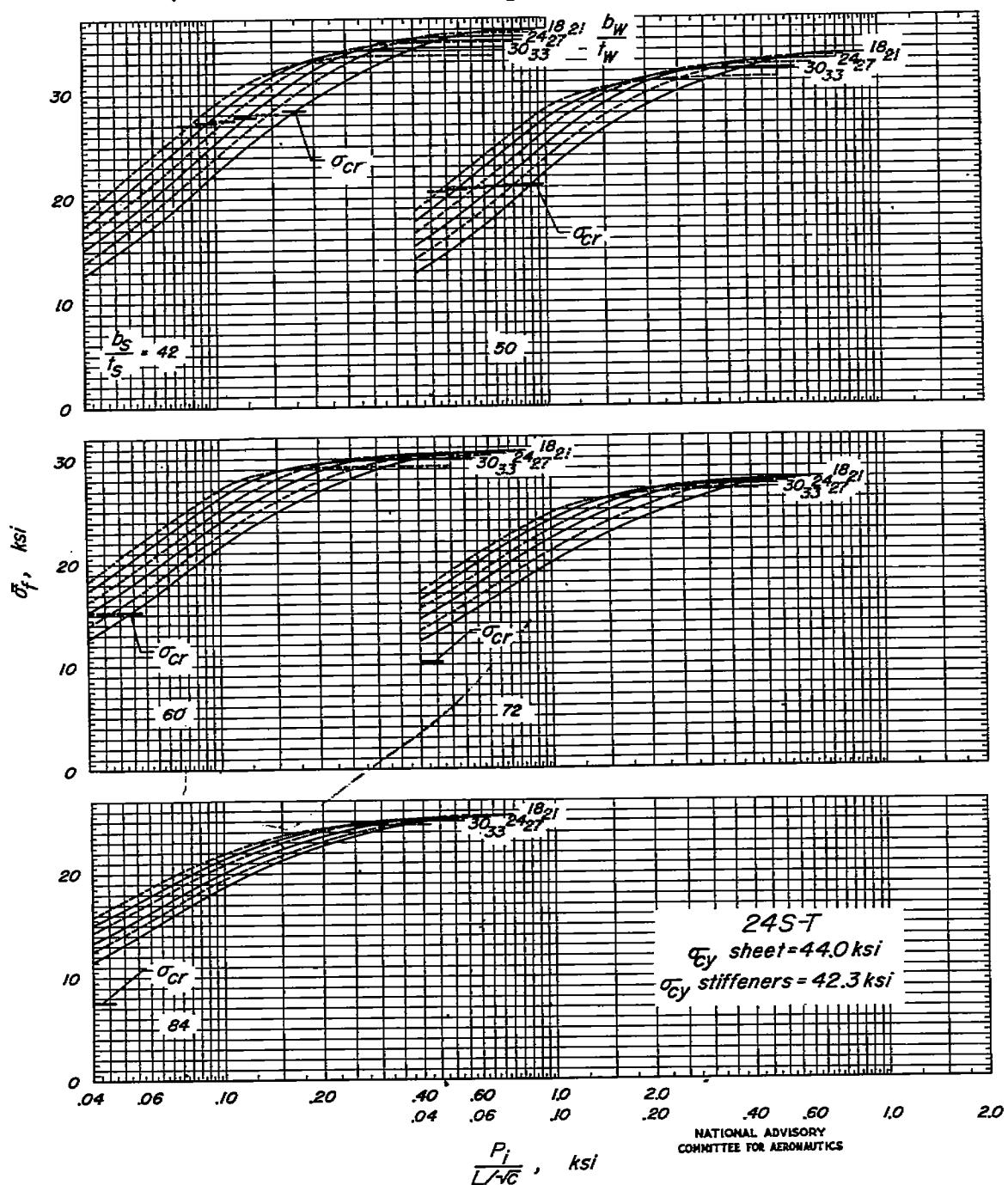


Figure 8. - Concluded.

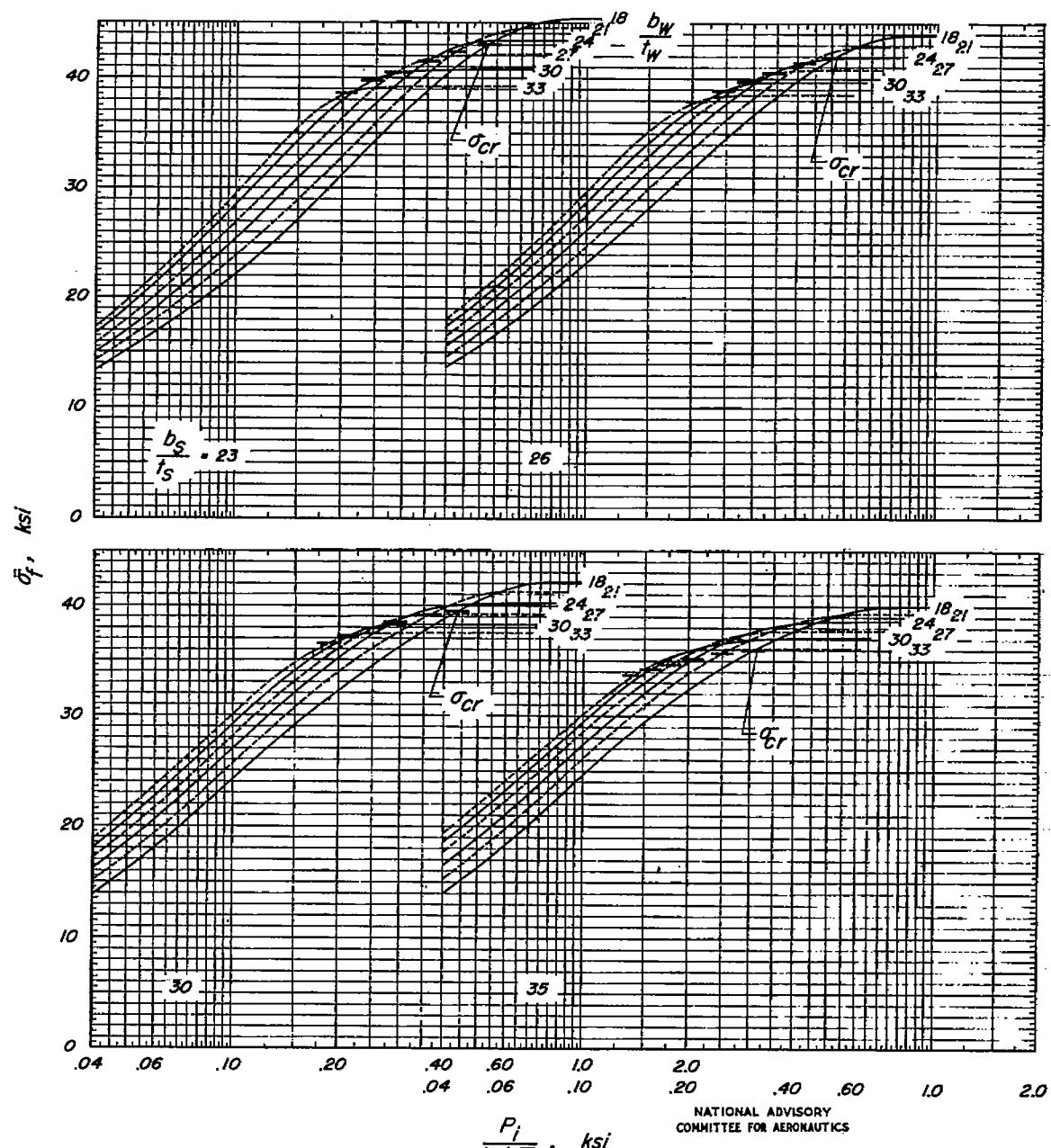


Figure 9.— Design chart for 24S-T Y-panels of the proportions tested. $\frac{t_w}{t_s} = 0.63$.

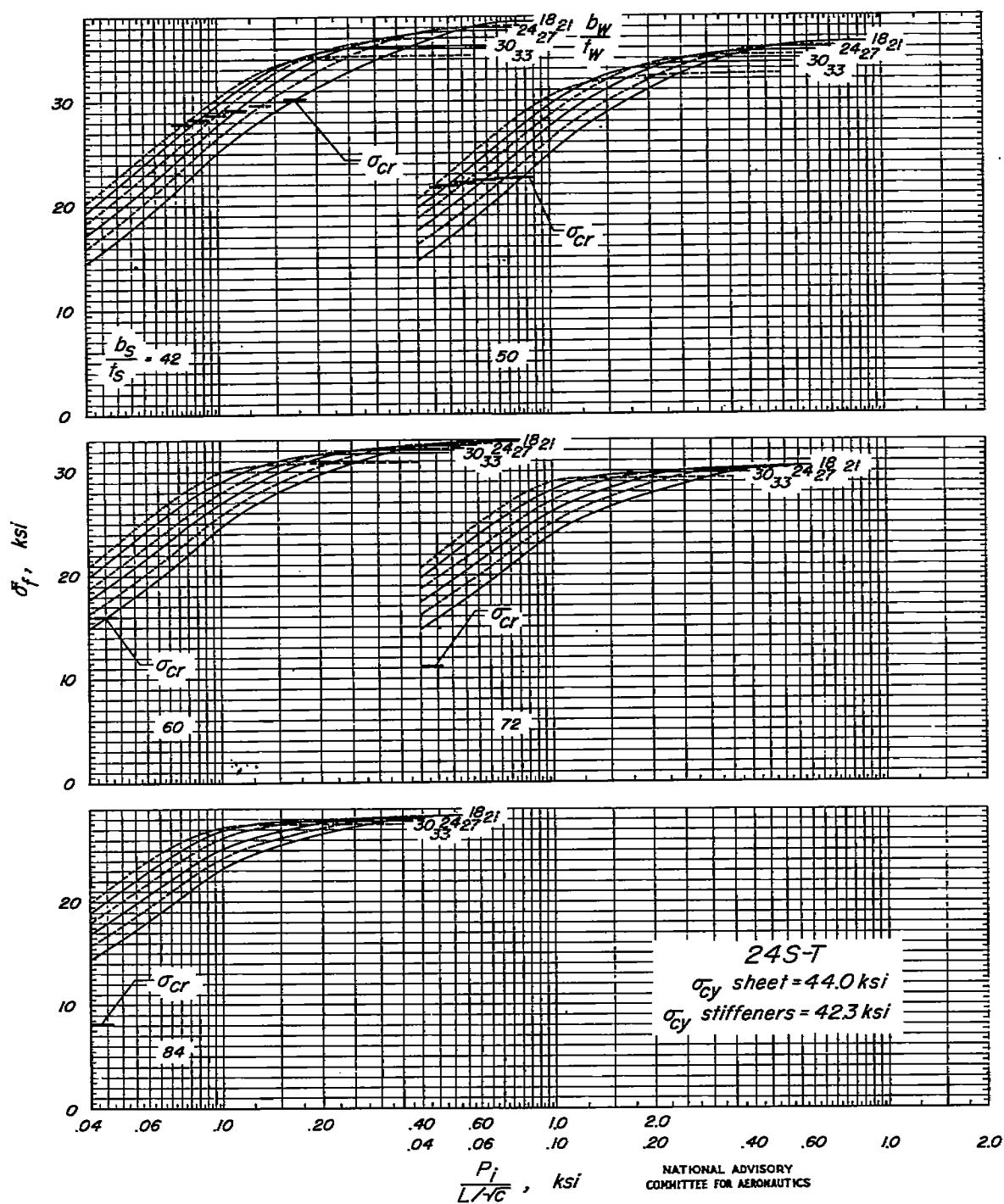


Figure 9.- Concluded.

Fig. 10

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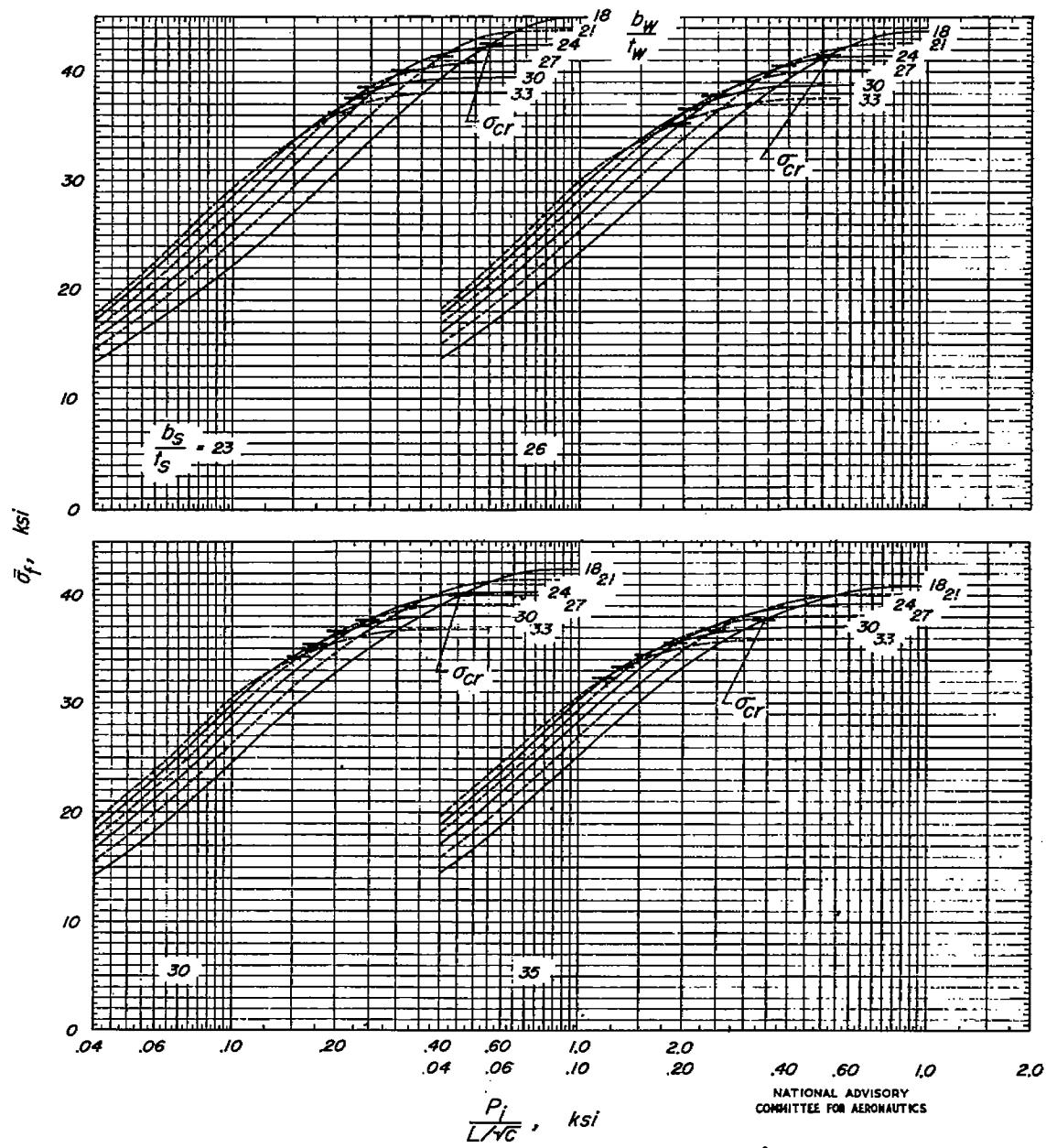


Figure 10.- Design chart for 24S-T Y-panels of the proportions tested, $\frac{t_w}{t_s} = 0.79$.

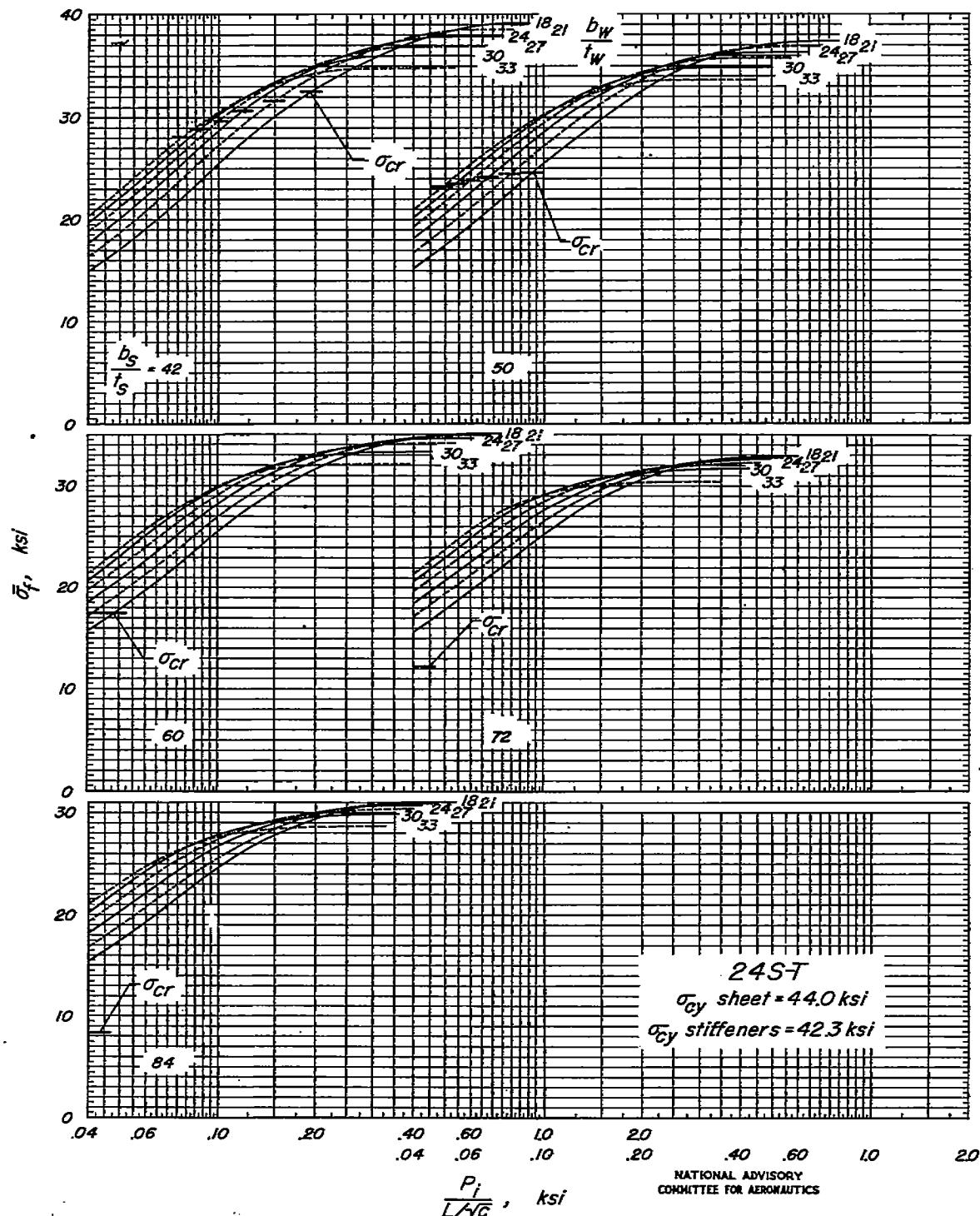


Figure 10 - Concluded.

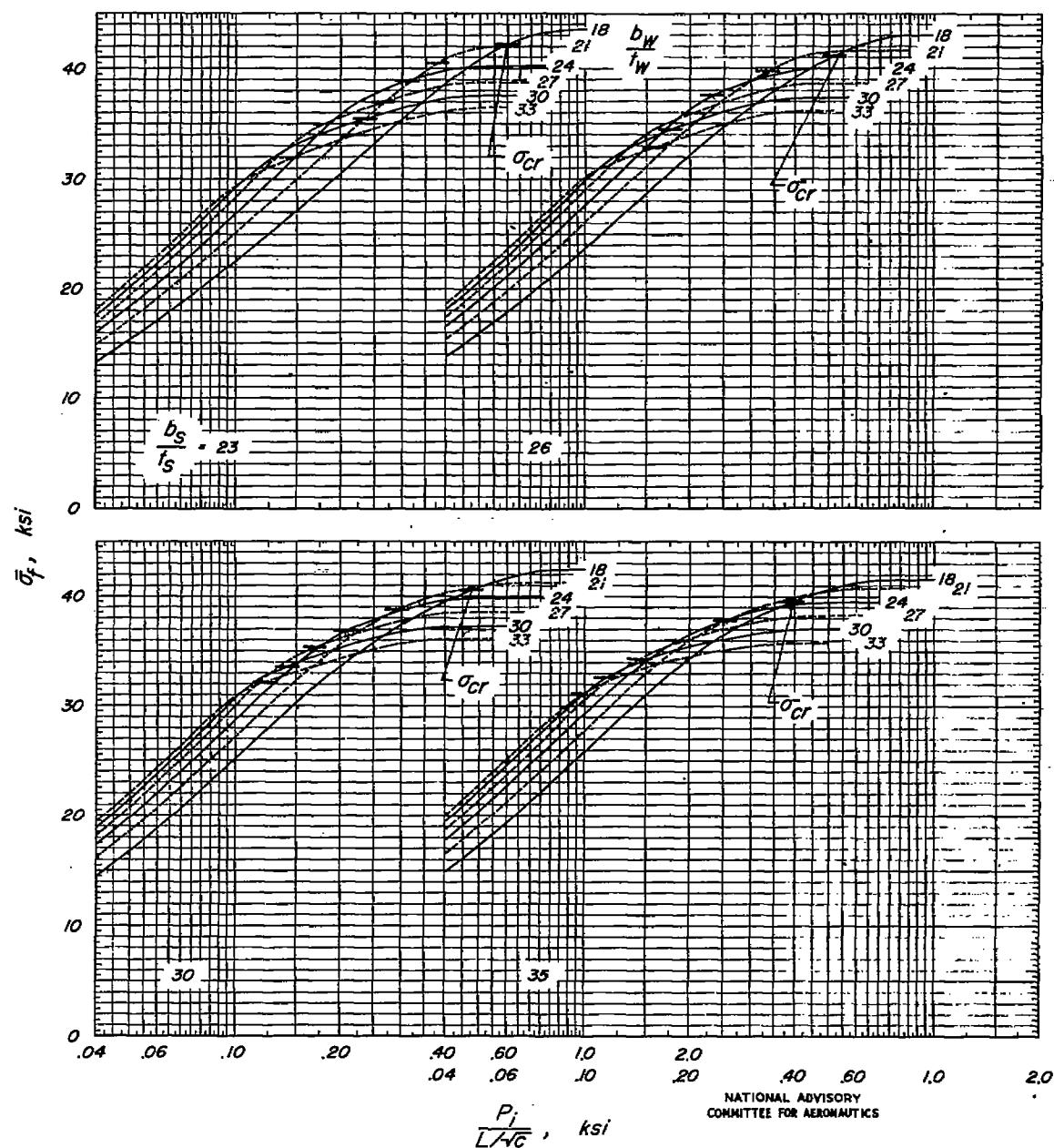


Figure 11. - Design chart for 24S-T Y-panels of the proportions tested. $\frac{t_w}{t_s} = 1.00$.

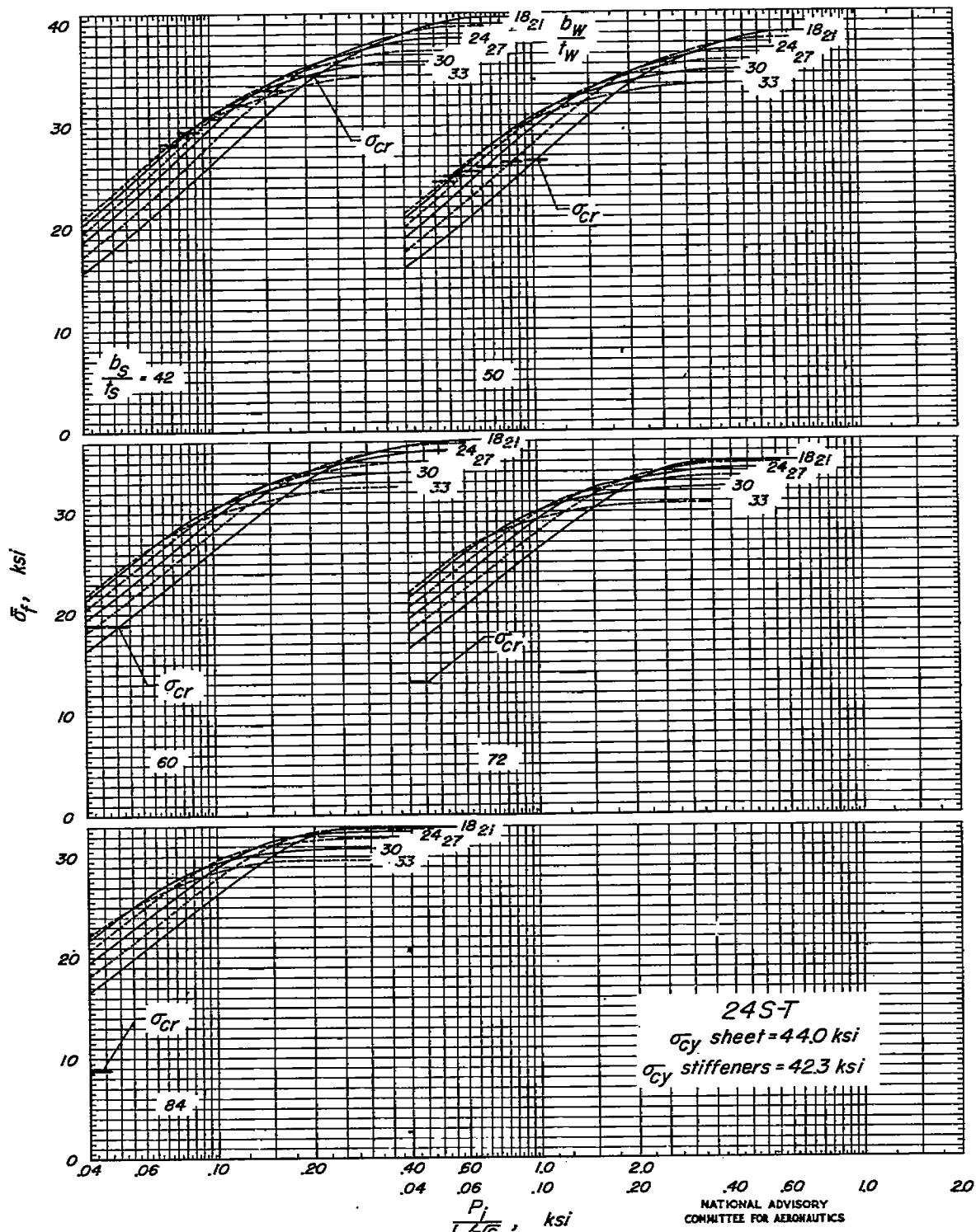
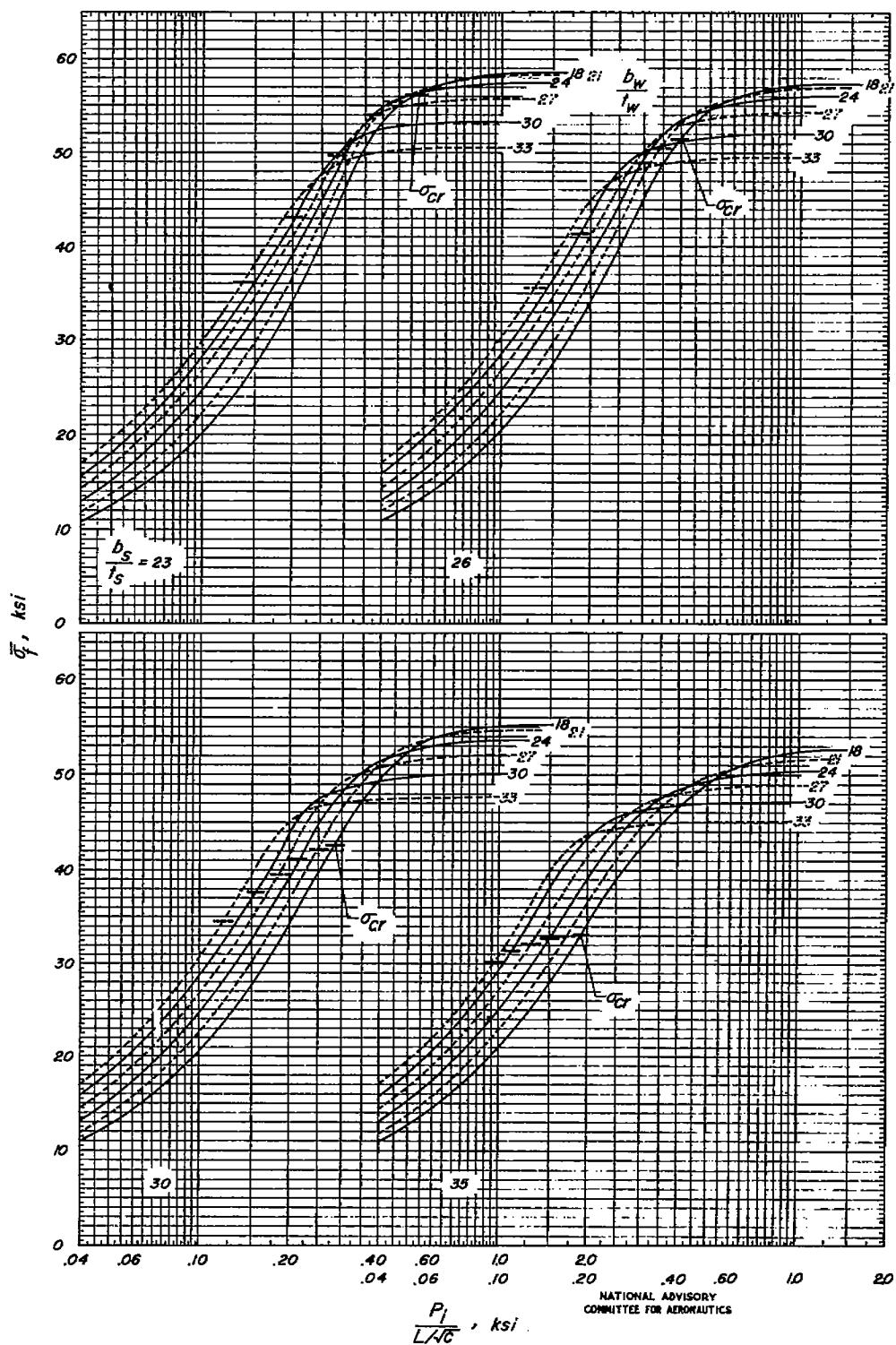


Figure 11.- Concluded.

Fig. 12

NACA TN No. 1389

Figure 12.- Design chart for 75S-T Y-panels of the proportions tested. $\frac{t_w}{t_s} = 0.40$.

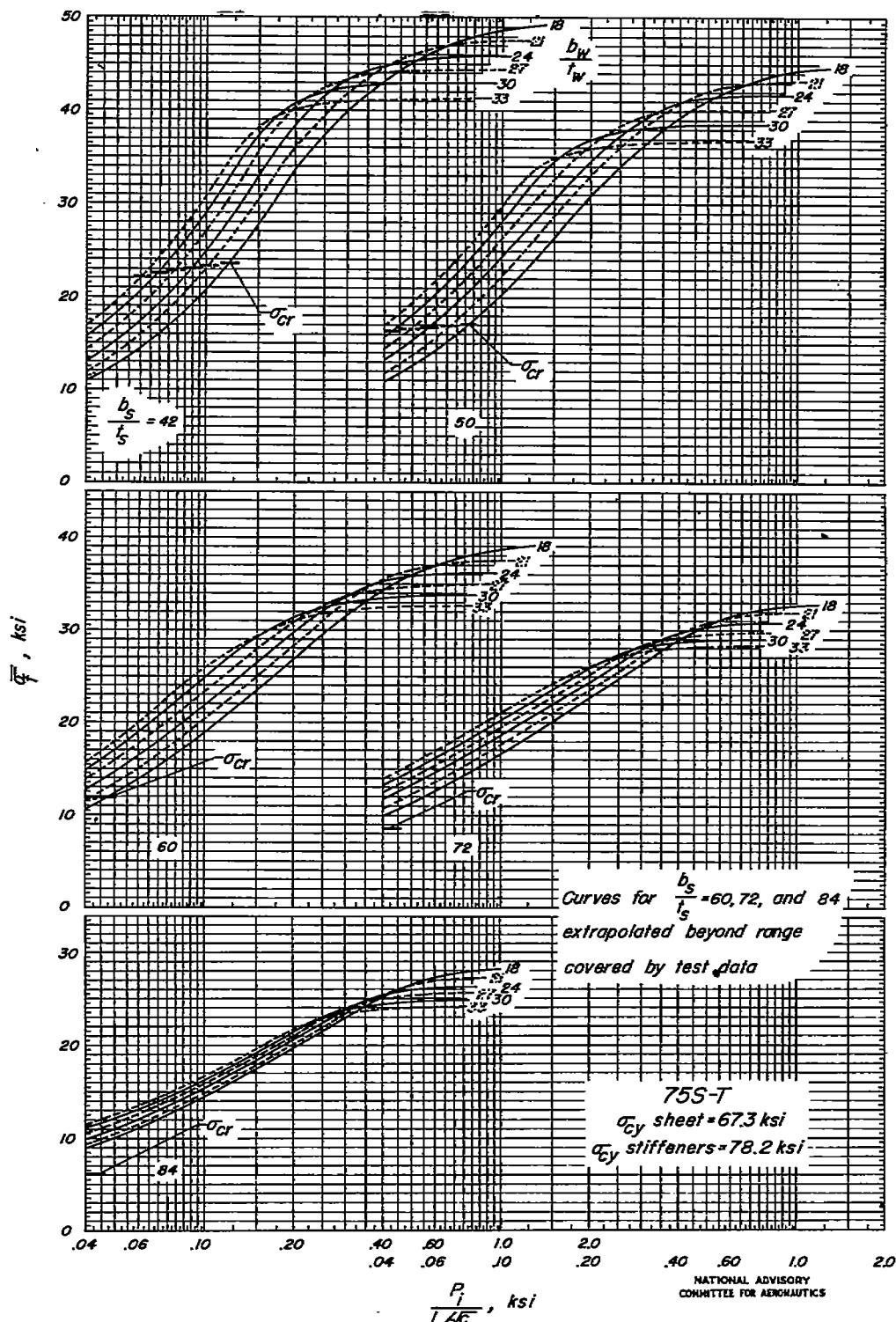
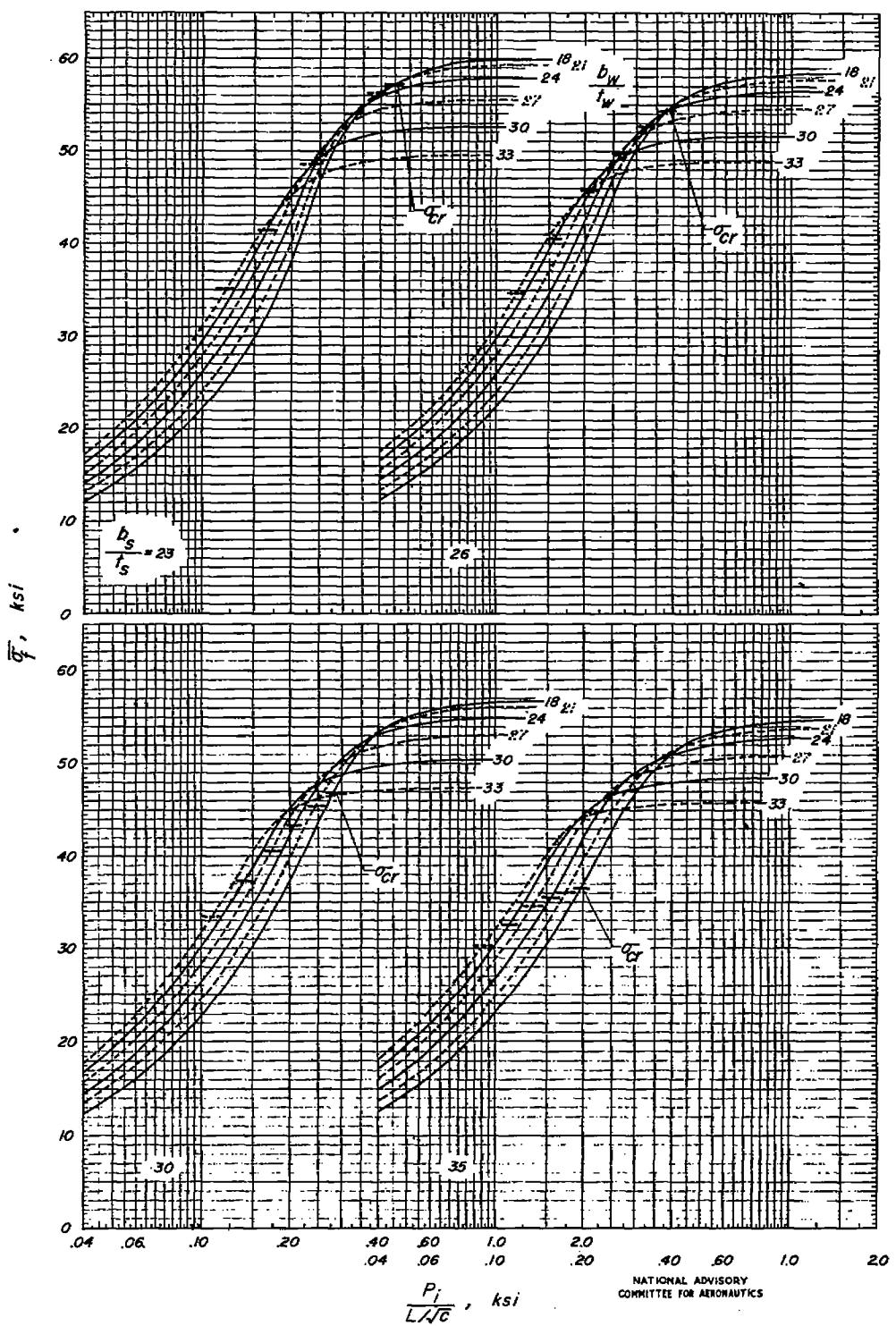


Figure 12-Concluded.

Fig. 13

NACA TN No. 1389

Figure 13—Design chart for 75S-T Y-panels of the proportions tested. $\frac{t_w}{t_s} = 0.51$.NATIONAL ADVISORY
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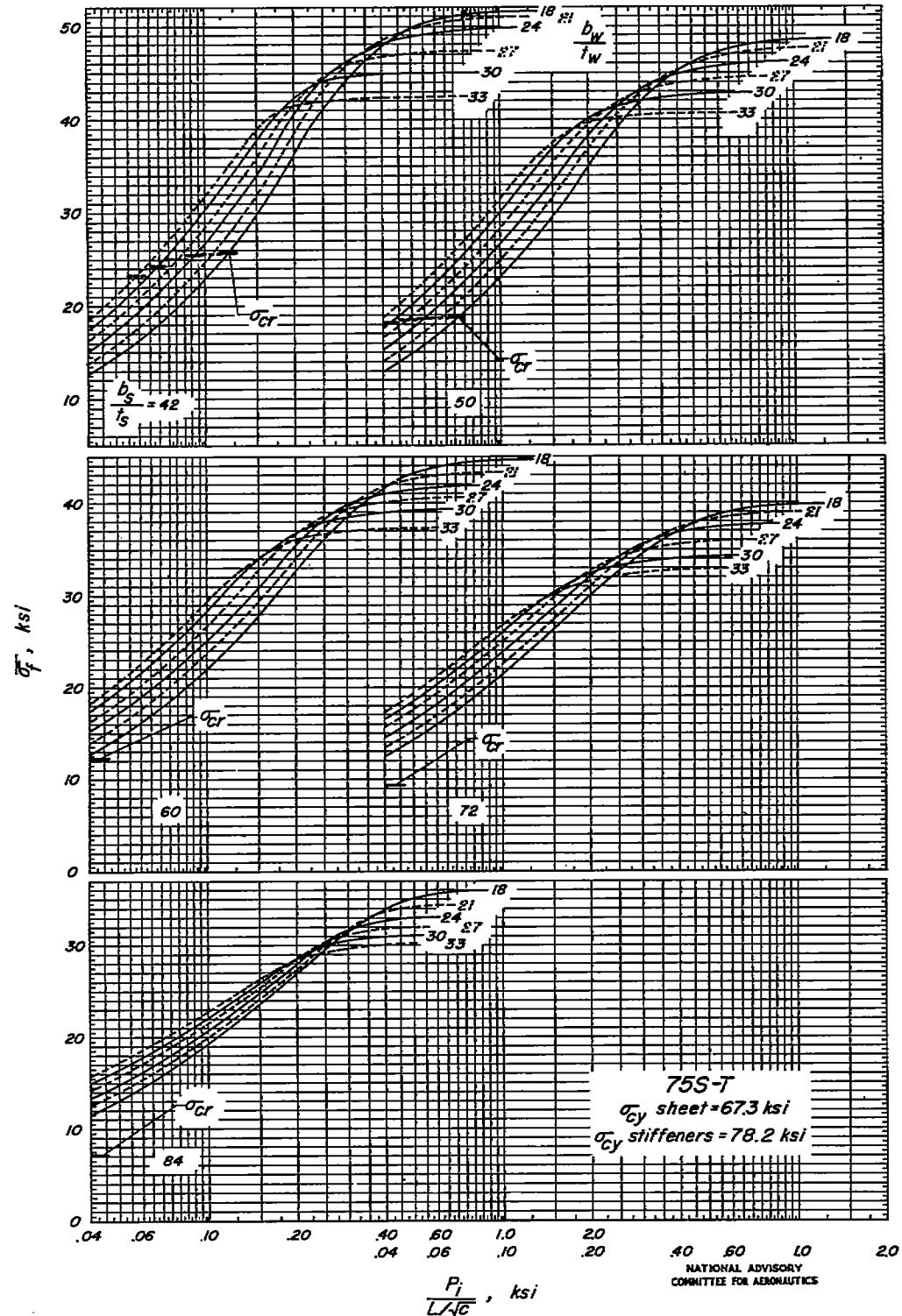
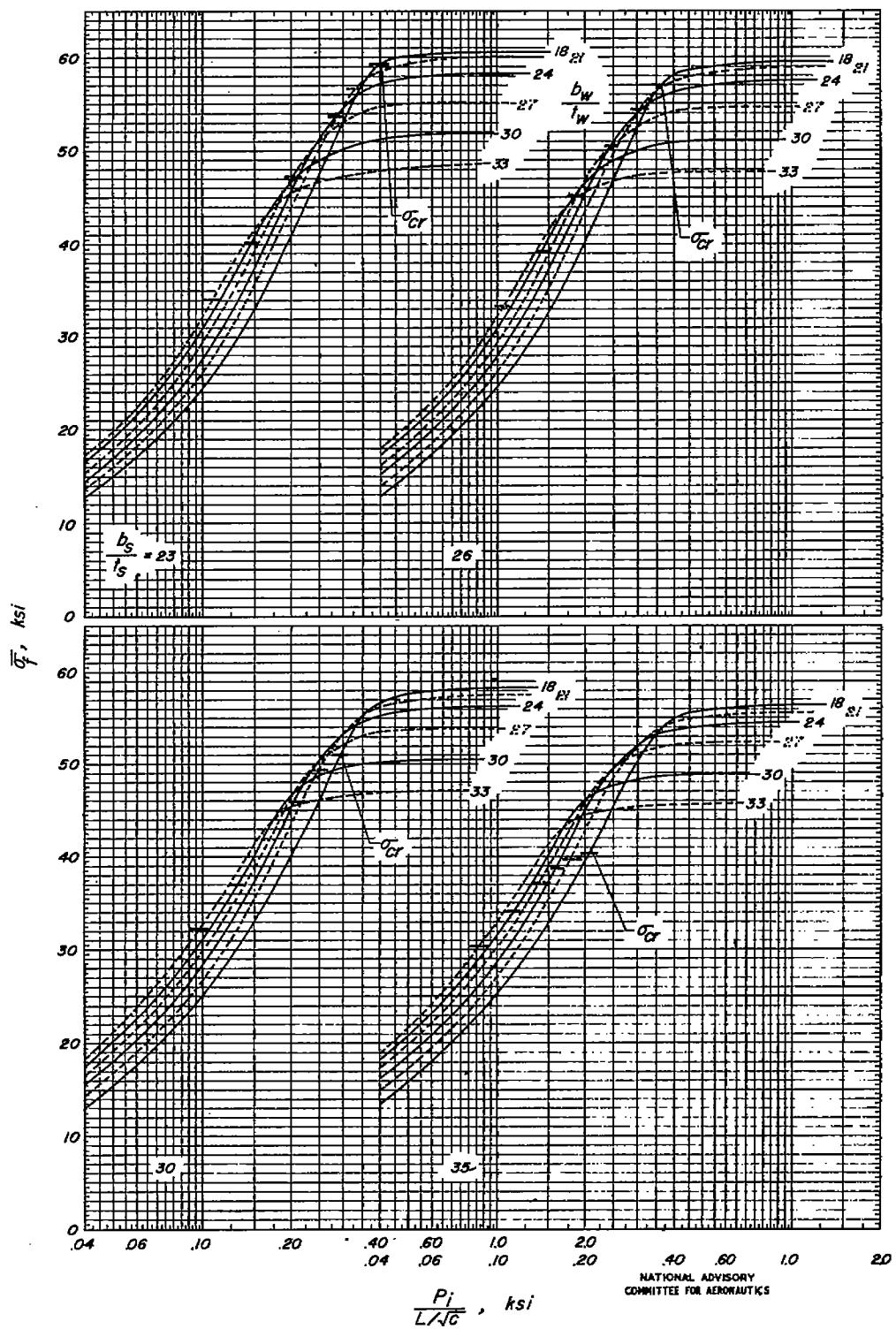


Figure 13.—Concluded.

Fig. 14

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Figure 14.—Design chart for 75S-T Y-panels of the proportions tested. $\frac{t_w}{t_s} = 0.63$.

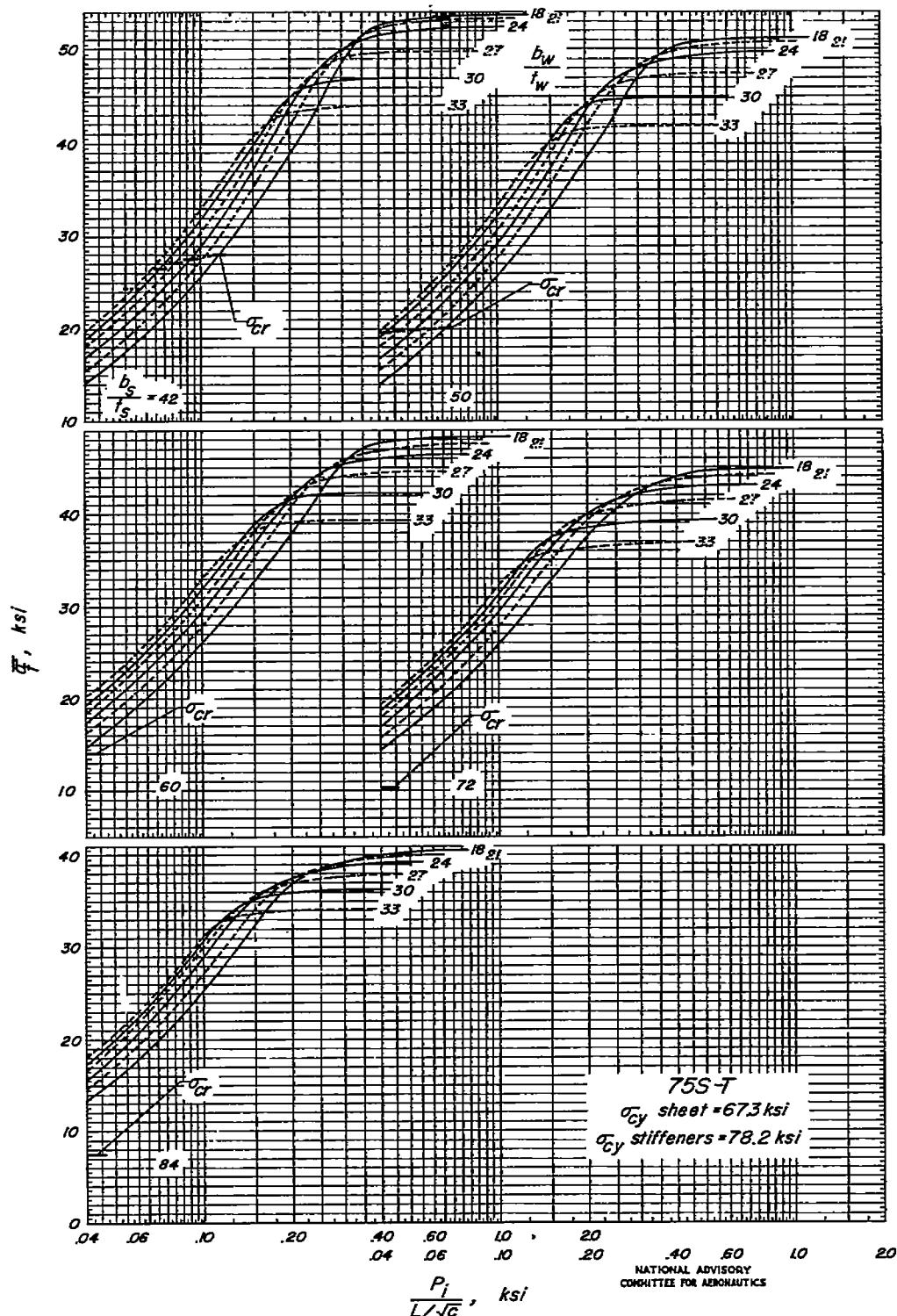


Figure 14.-Concluded.

Fig. 15

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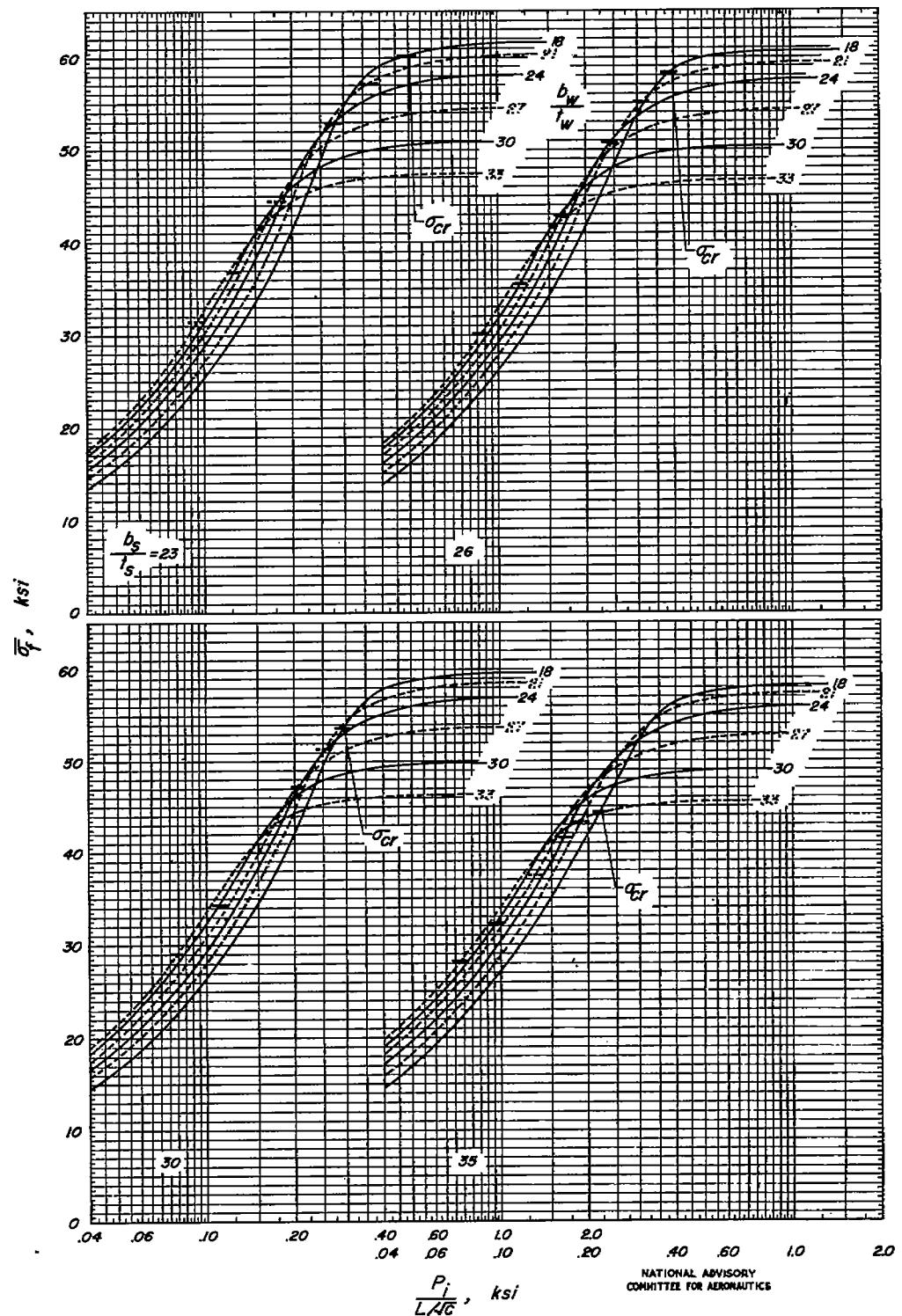


Figure 15.—Design chart for 75S-T Y-panels of the proportions tested. $\frac{t_w}{t_s} = 0.79$.

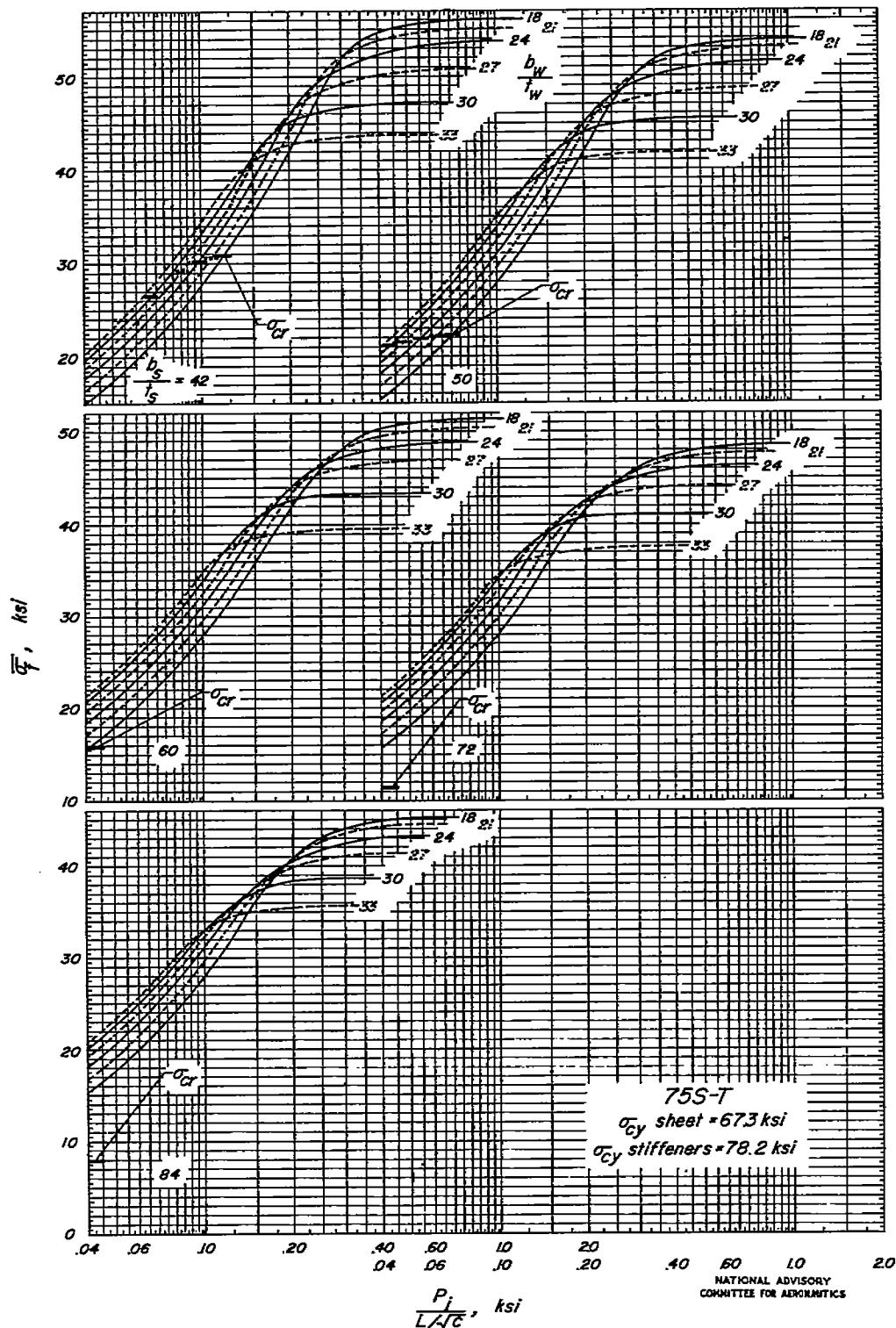


Figure 15.—Concluded.

Fig. 16

NACA TN No. 1389

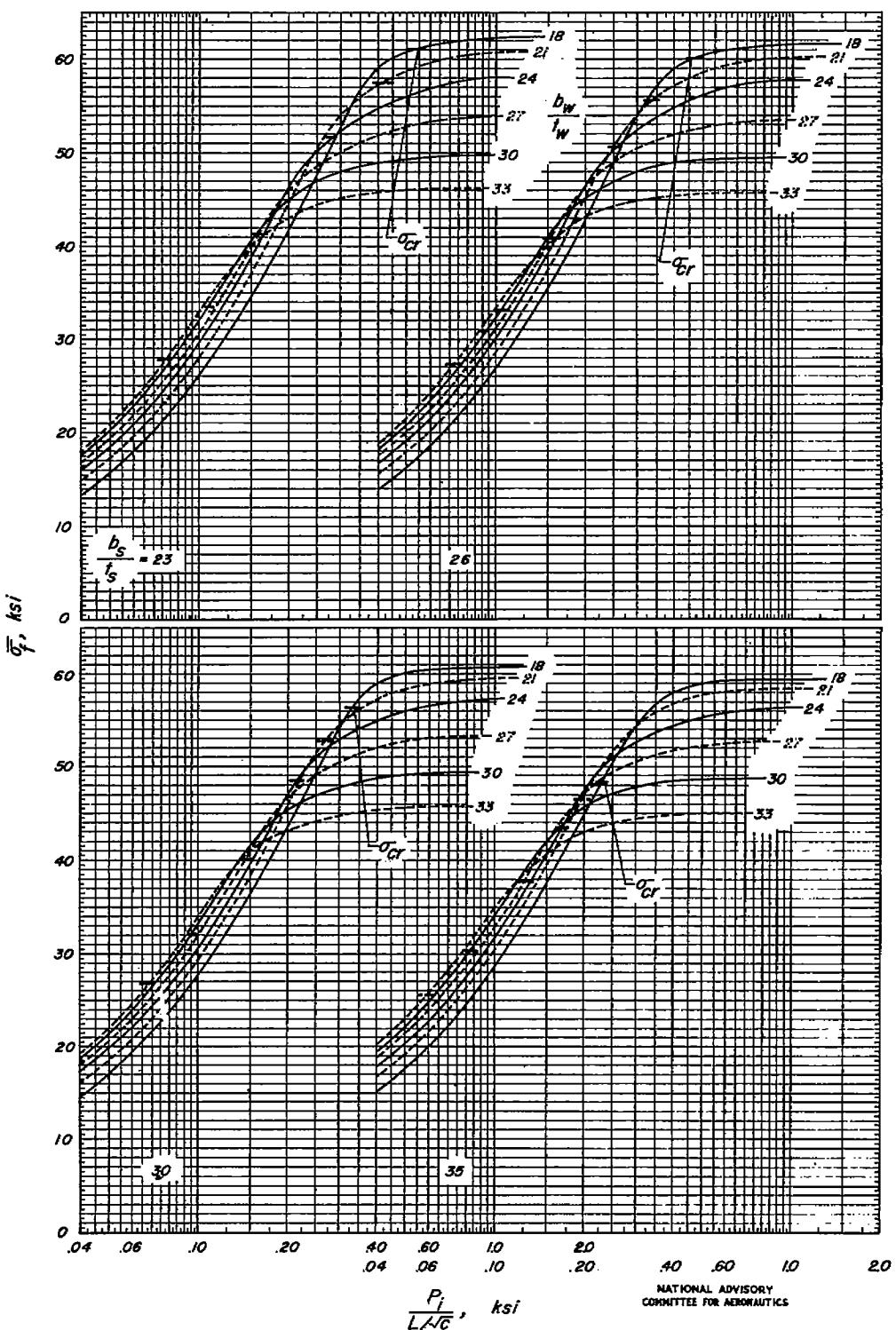


Figure 16.—Design chart for 75S-T Y-panels of the proportions tested. $\frac{t_w}{t_s} = 1.00$.

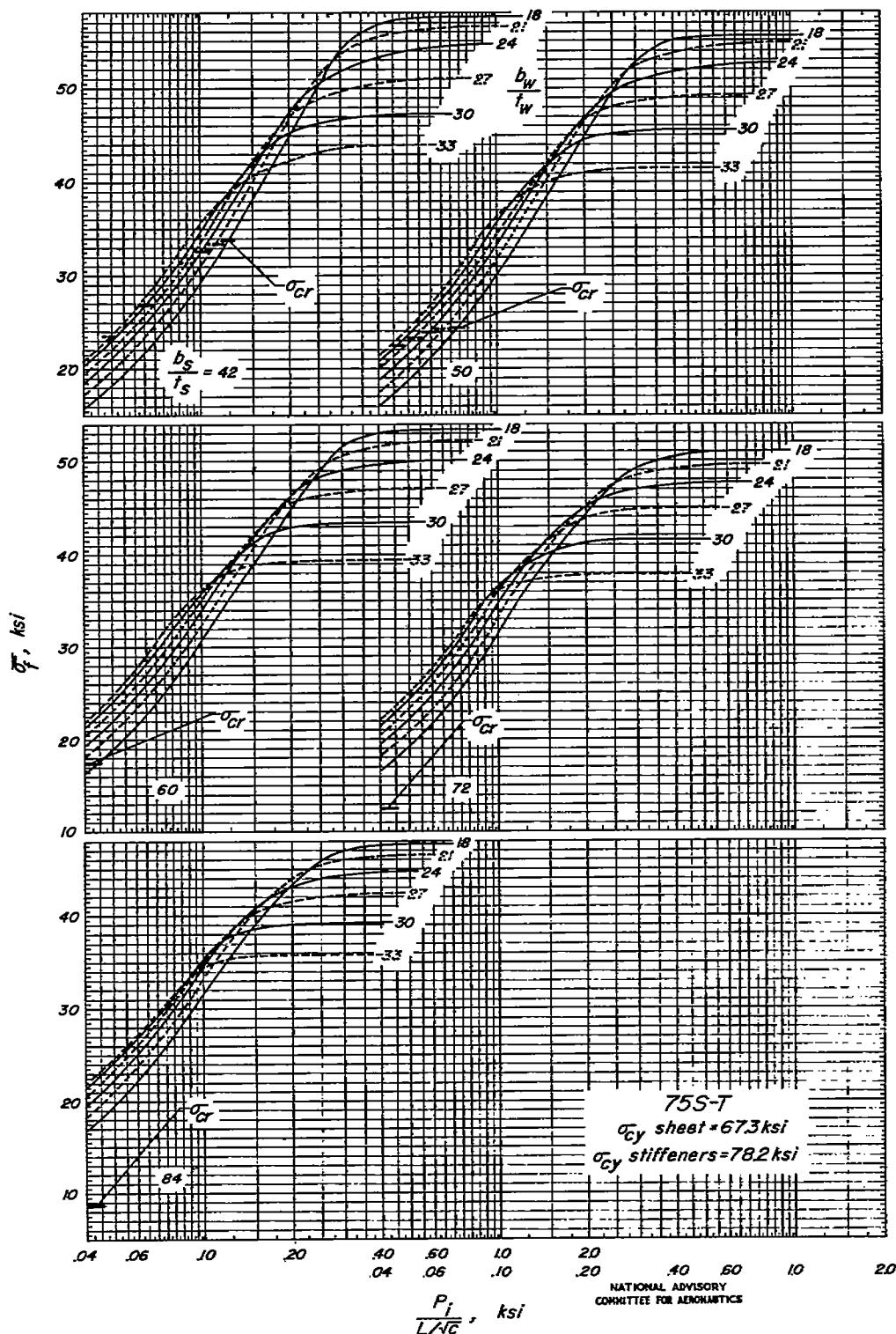


Figure 16—Concluded.

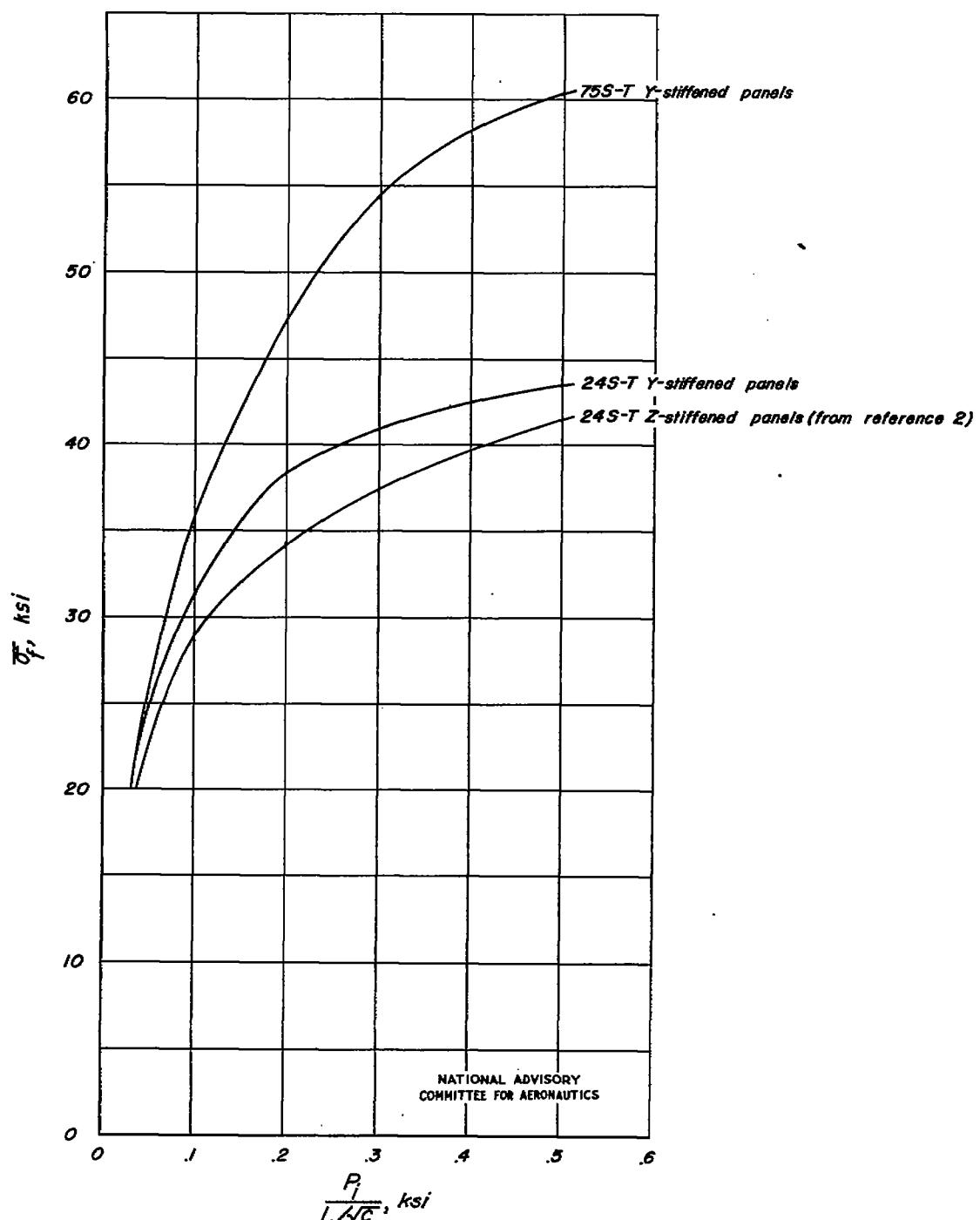


Figure 17-Comparison of envelope curves for 24S-T
Z-stiffened panels (from reference 2) and
for 24S-T and 75S-T Y-stiffened panels.

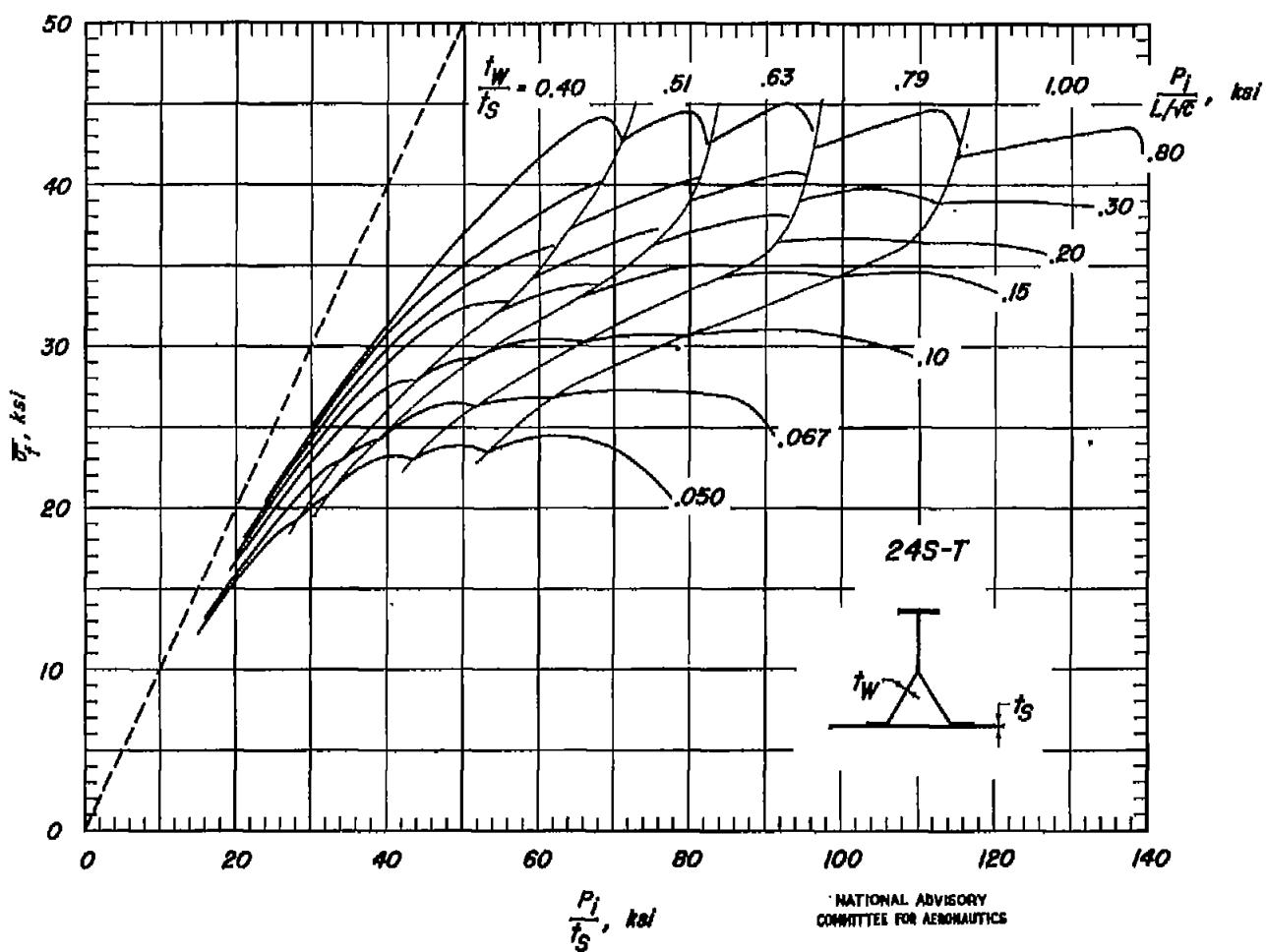


Figure 18. - Design chart for the determination of the average stress at failure that can be carried by minimum-weight designs of 24S-T aluminum-alloy flat compression panels having extruded Y-section stiffeners.

Fig. 19

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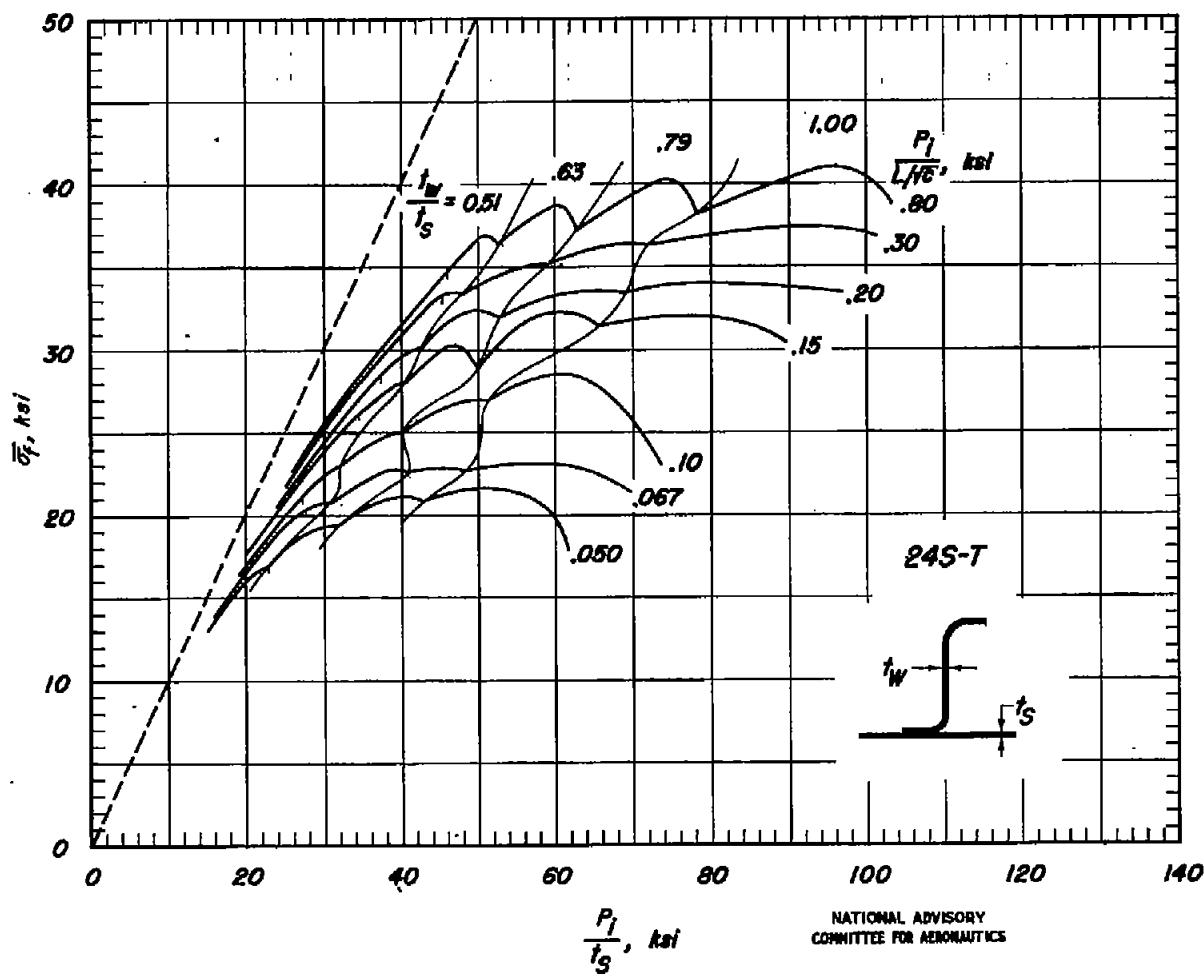


Figure 19.—Design chart for the determination of the average stress at failure that can be carried by minimum-weight designs of 24S-T aluminum-alloy flat compression panels having formed Z-section stiffeners.

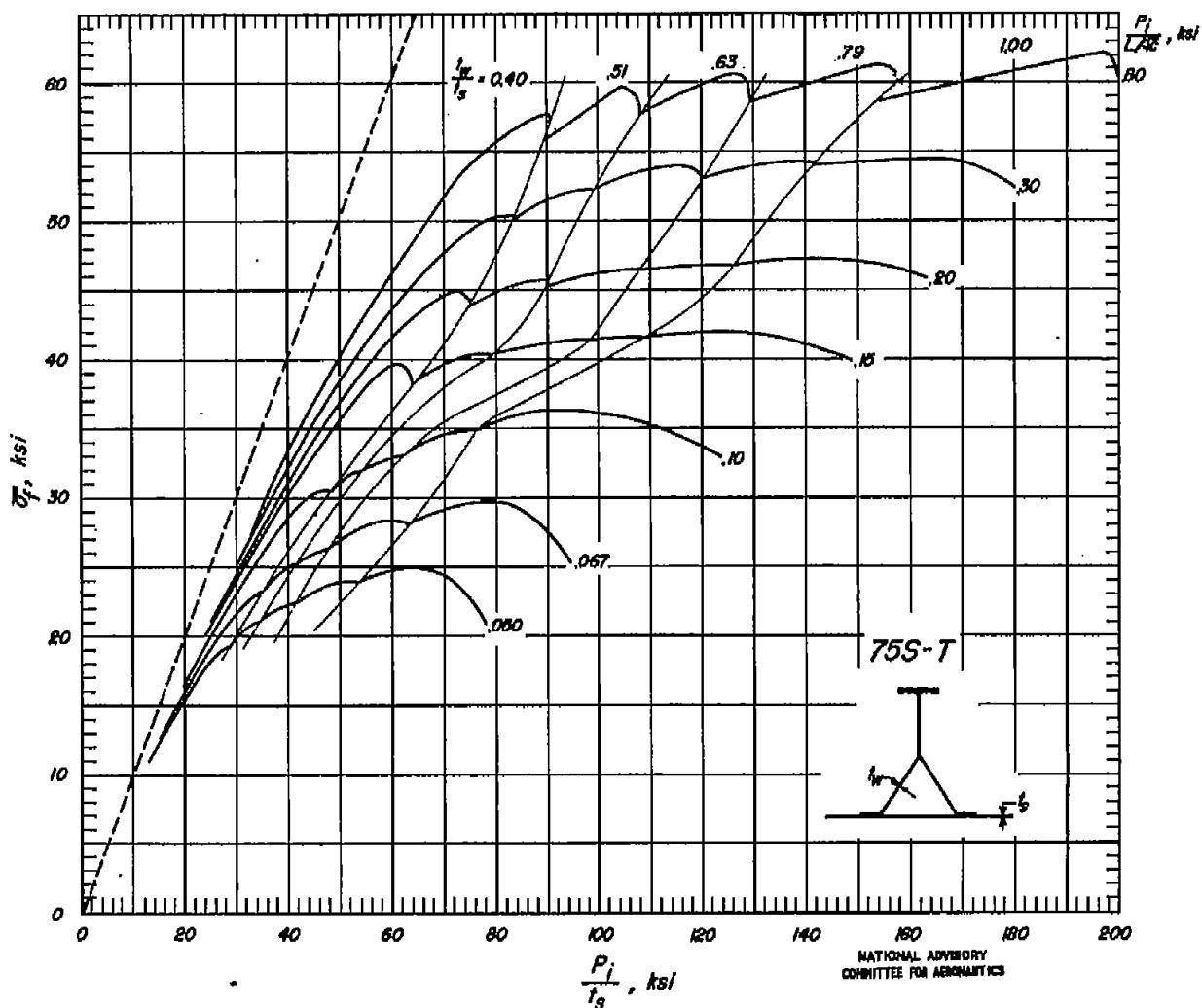
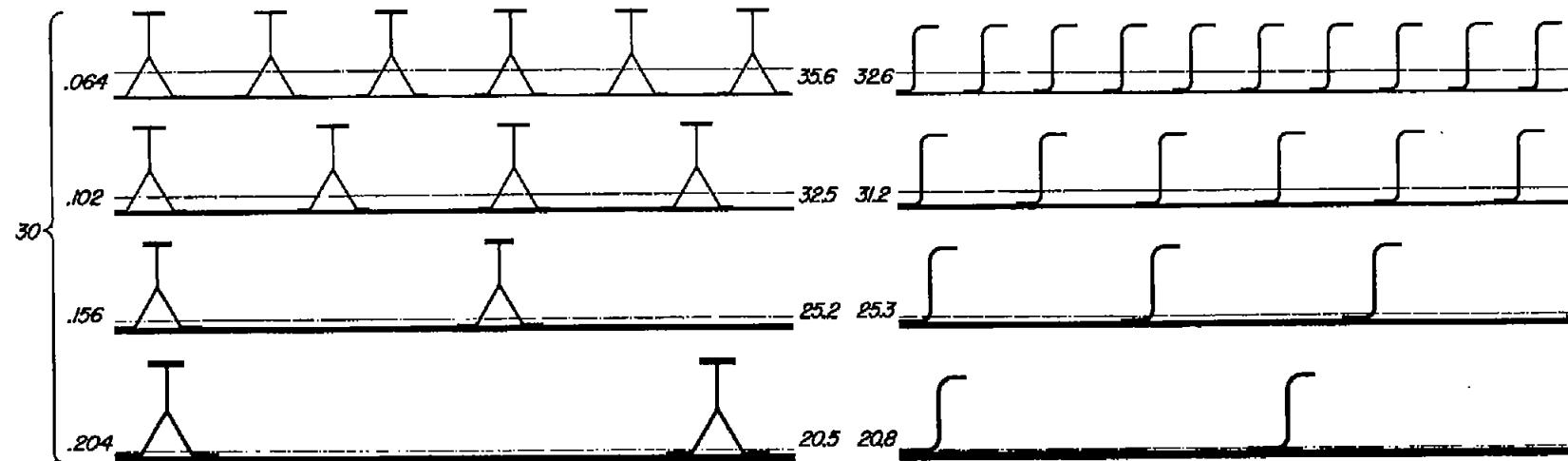
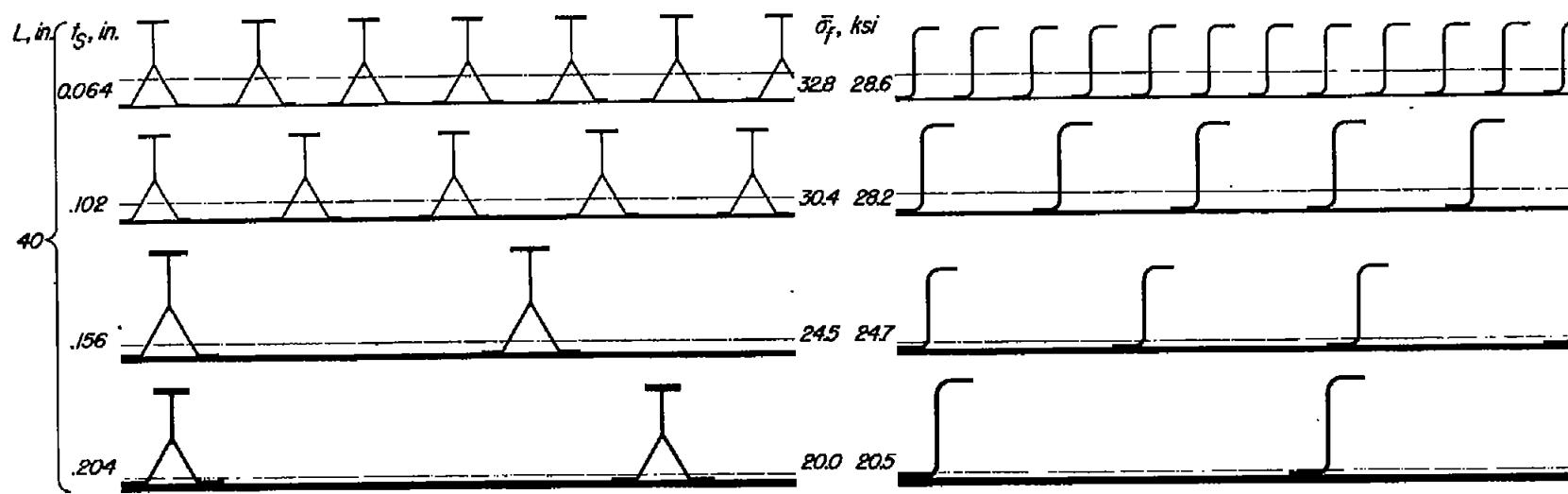


Figure 20: Design chart for the determination of the average stress at failure that can be carried by minimum-weight designs of flat compression panels having Alclad 75S-T aluminum-alloy sheet and extruded 75S-T Y-section stiffeners.

Fig. 21



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NACA TN No. 1389

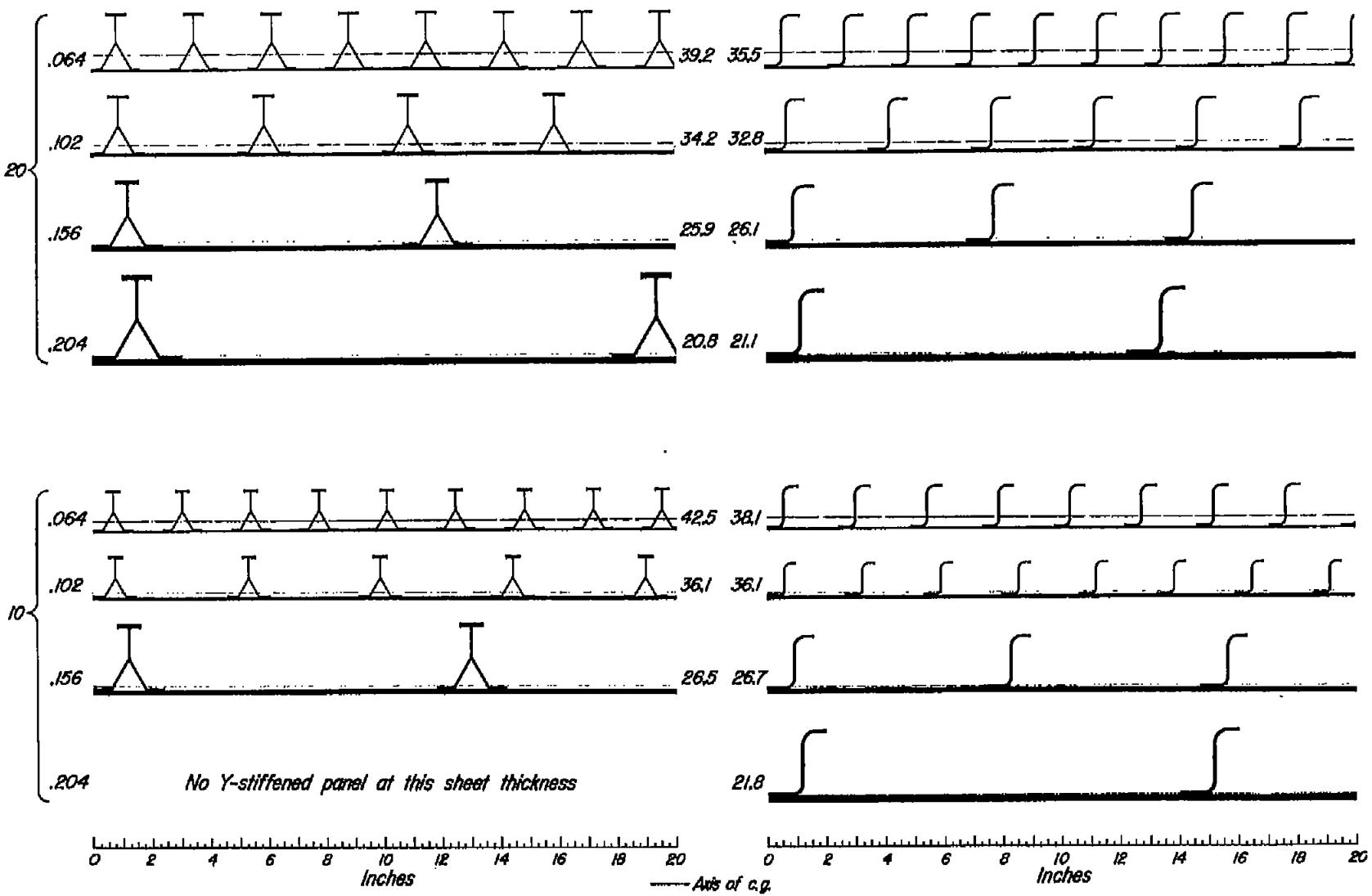
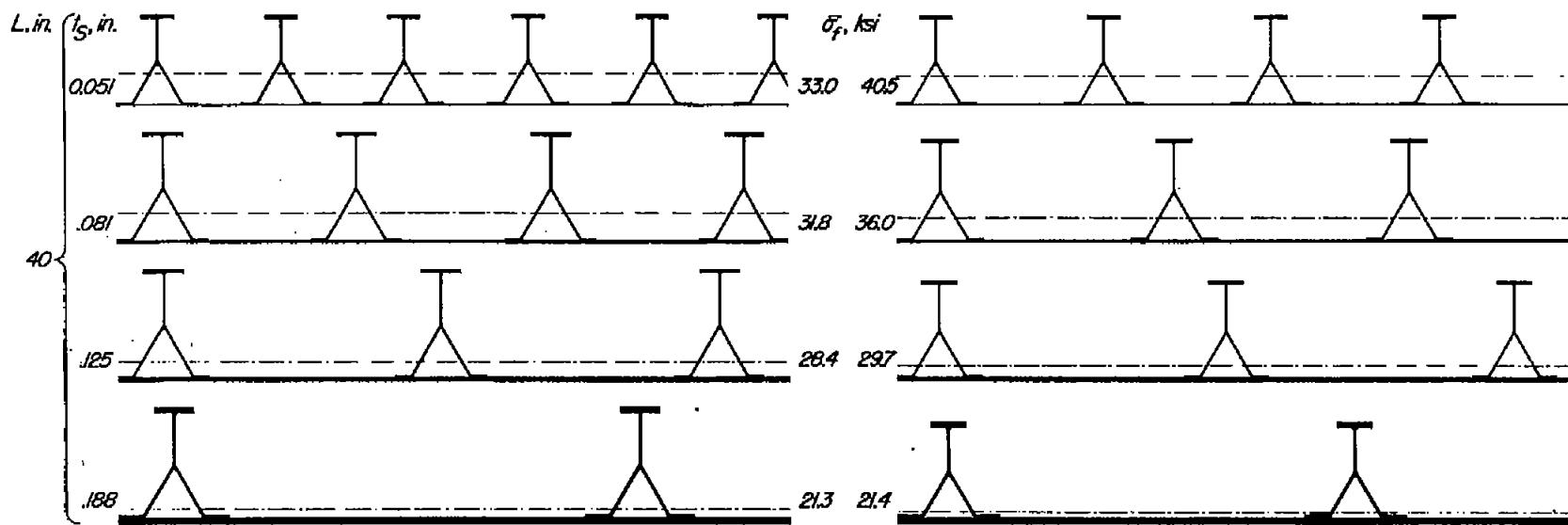
Figure 21.- Comparative minimum-weight designs of Y- and Z-stiffened panels. $P_f = 5.0$ kips per inch; $c=1$.NATIONAL ADVISORY
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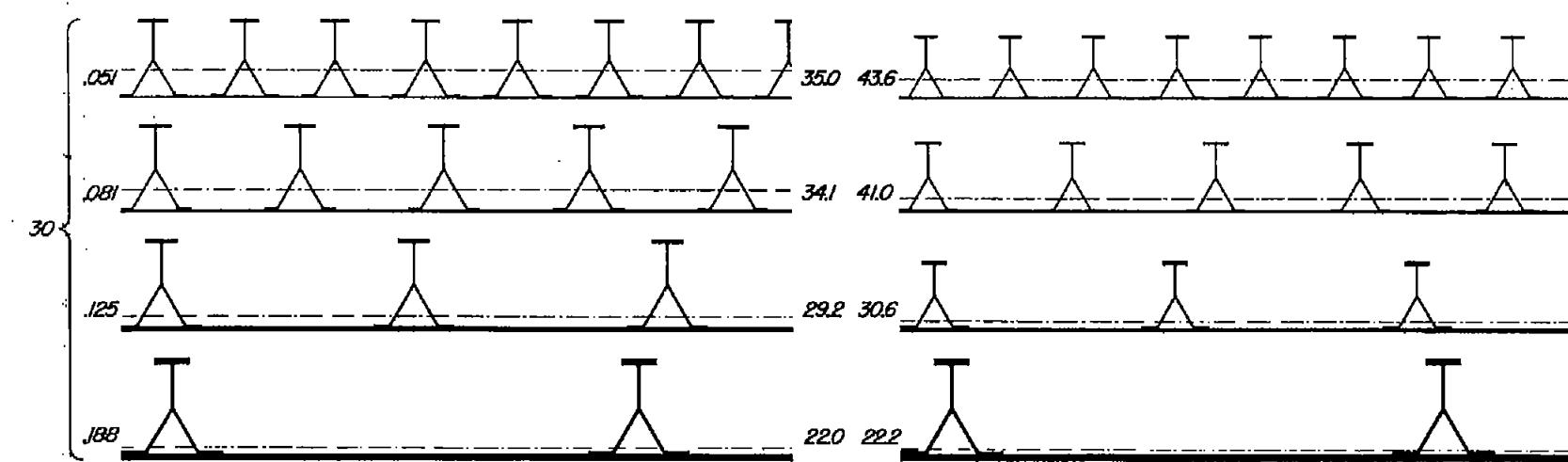
Fig. 22

NACA TN No. 1389



24S-T

75S-T



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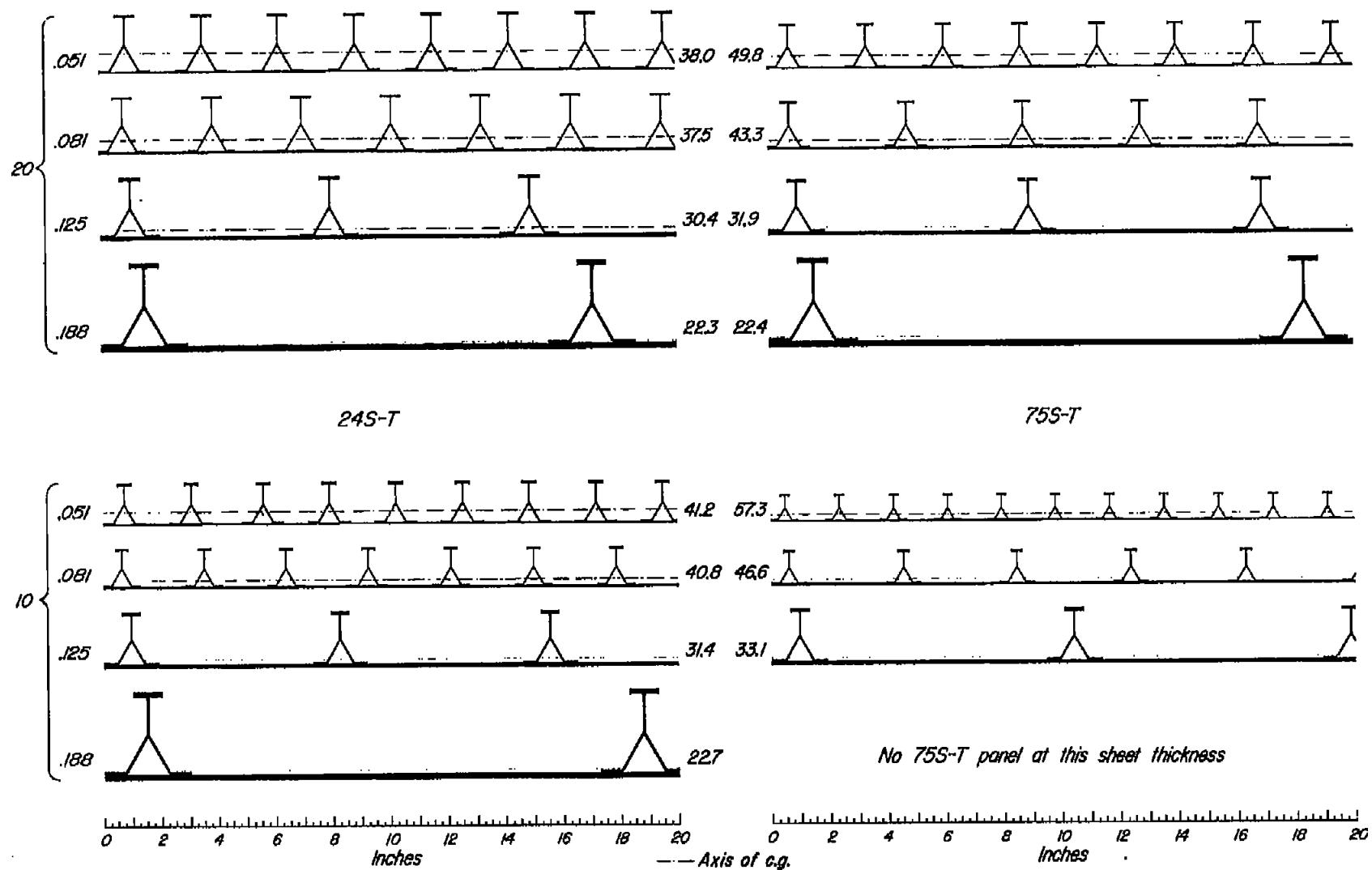


Figure 22.—Comparative minimum-weight designs of 24S-T and 75S-T Y-stiffened panels. $P_i = 5.0$ kips per inch; $c = l$.

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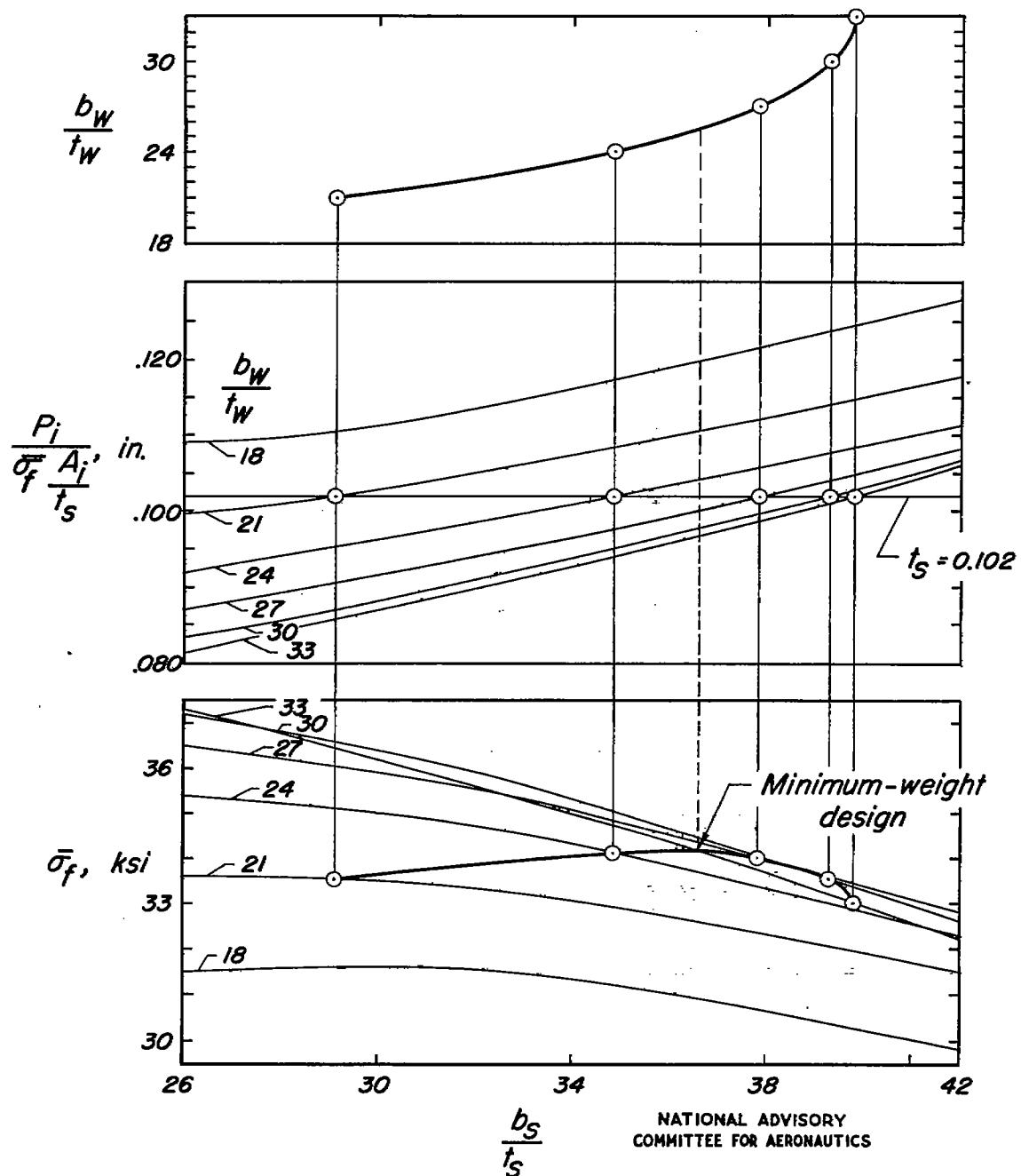
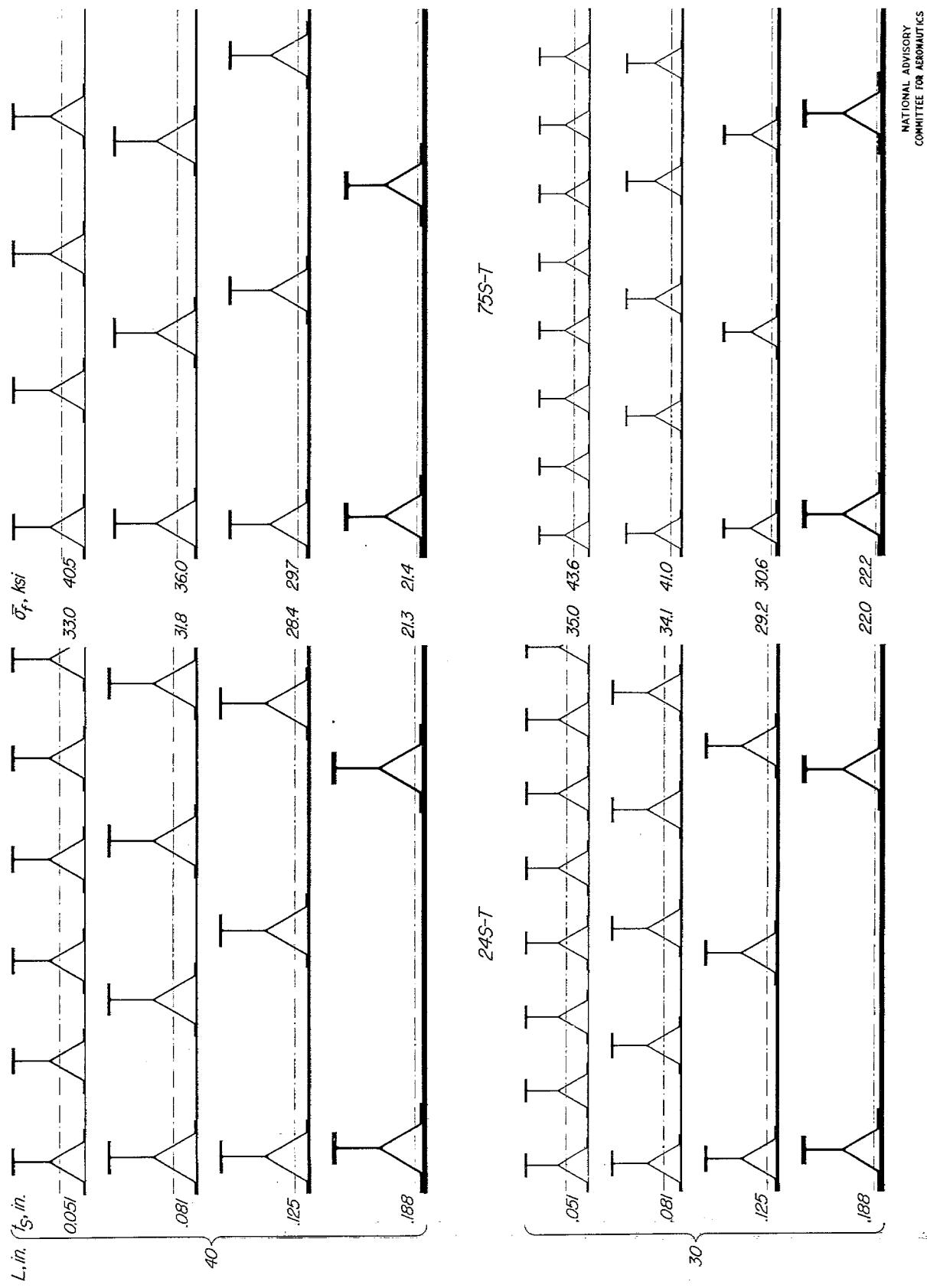
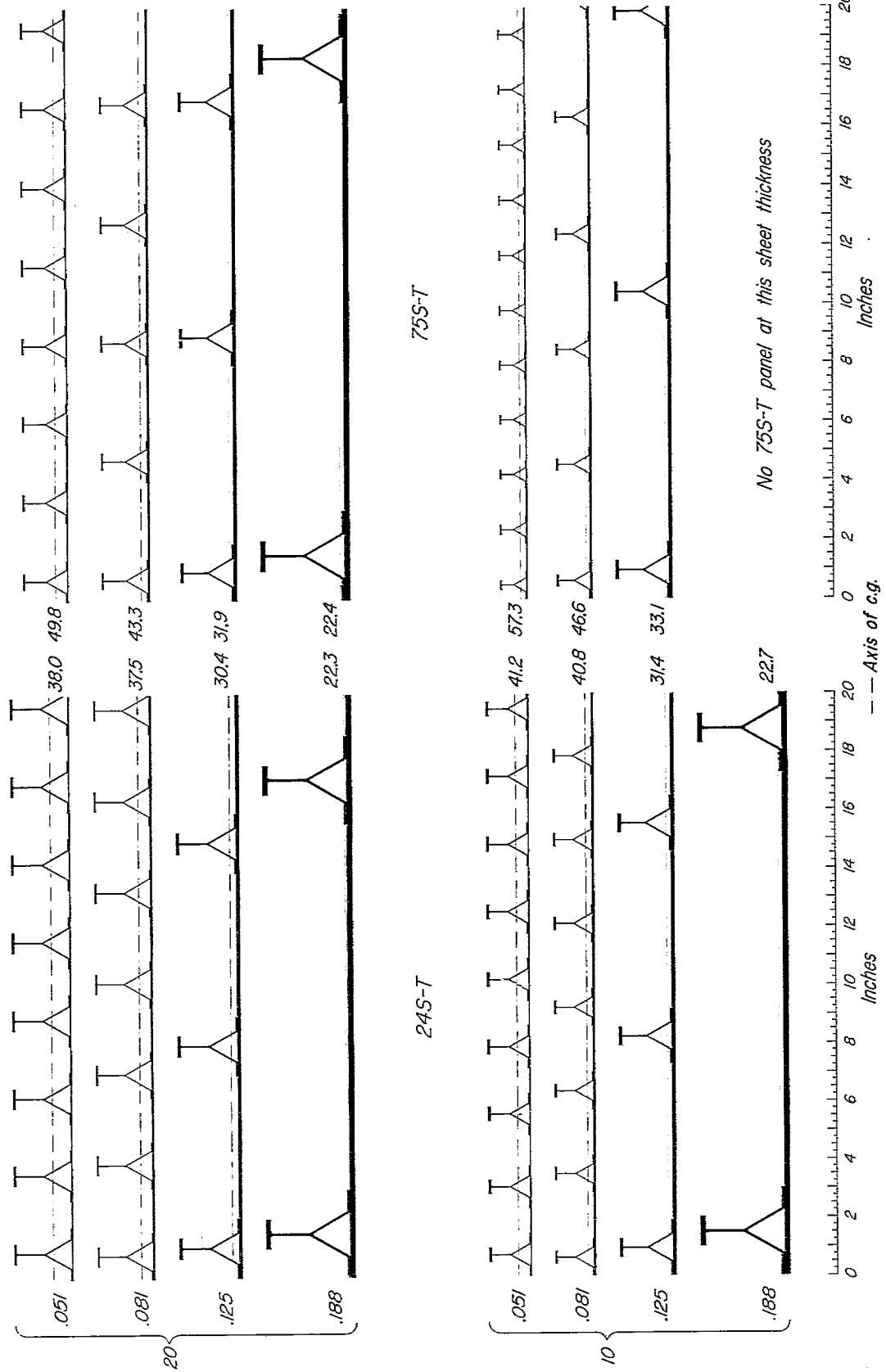


Figure 23 - Plot for obtaining design for maximum structural efficiency (minimum weight). $P_i = 5.0$ kips per inch; $L = 20$ inches; $c = l$; $t_S = 0.102$ inch; $\frac{t_W}{t_S} = 0.40$.

Fig. 22

NACA TN No. 1389

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Figure 22.—Comparative minimum-weight designs of 24S-T and 75S-T Y-stiffened panels. $P_f = 5.0$ kips per inch; $C = 1$.

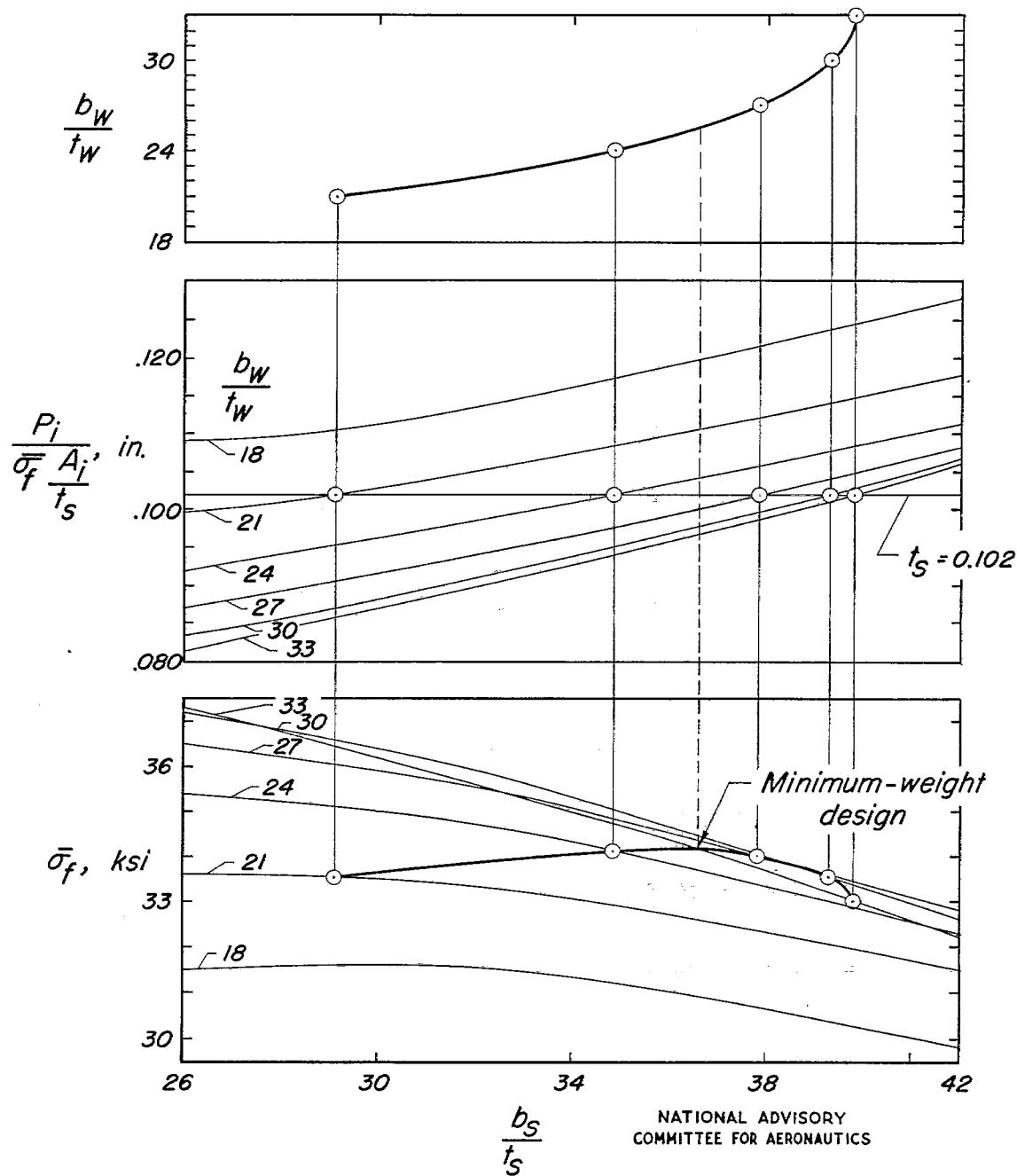


Figure 23.- Plot for obtaining design for maximum structural efficiency (minimum weight). $P_i = 5.0$ kips per inch; $L = 20$ inches; $c = 1$; $t_S = 0.102$ inch; $\frac{t_W}{t_S} = 0.40$.